FATIGUE LIFE ESTIMATION OF TURBINE BLADES DUE TO TRANSIENT THERMAL, VIBRATORY, AND CENTRIFUGAL STRESSES

CENTRE FOR NEWFOUNDLAND STUDIES

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DEEPAK DHAR
FATIGUE LIFE ESTIMATION OF TURBINE BLADES DUE TO TRANSIENT THERMAL, VIBRATORY, AND CENTRIFUGAL STRESSES

By
Deepak Dhar (B.Eng)

A Thesis submitted to the School of Graduate Studies in partial fulfilment of the requirements for the degree of Master of Engineering

Faculty of Engineering and Applied Science
Memorial University of Newfoundland
Oct. 1994

St. John's Newfoundland Canada
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ABSTRACT

The present work involves the study of the three dimensional transient heat transfer and vibratory analysis of turbine blades with mixed and non-linear boundary conditions. The equations derived for the heat transfer analysis are general (three dimensional) in nature and are new contributions in this field. The mathematical model for both the heat transfer and the vibration analysis is formulated using curved, solid, C° continuity, quadratic, serendipity, twenty noded isoparametric finite elements.

The equations which are non-linear in nature for the transient temperature distribution within the turbine blade are derived using the finite element analysis. The non-linearities arise due to the radiative heat transfer and also due to the change in the material properties of the turbine blade with temperature. Using a finite difference scheme, the non-linear differential equations are transformed to non-linear algebraic equations in the time domain. The transient temperatures obtained from the heat transfer analysis are used in the calculation of the temperature gradients and transient thermal stresses. The vibratory analysis is done at first for finding the undamped natural frequencies of the turbine blade. These free vibration studies include the effect of the non-linearity in the stiffness matrix caused by the rotation of the turbine blade. The frequencies arising from the free vibration analysis also change because of the change in the material properties of the turbine blade as the temperature of the turbine blade changes with time. Thereafter, the stresses due to (a) the nozzle excitation and (b) the centrifugal stresses at different rotor speeds are calculated. The nozzle excitation
forces are modelled as a sinusoidal pulse. Finally, the total effect of all the three different types of stresses (transient thermal, centrifugal, and vibratory due to the nozzle excitation) on the fatigue life of the turbine blade is studied.
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This work is dedicated to my parents.
# TABLE OF CONTENTS

<table>
<thead>
<tr>
<th>Section</th>
<th>Page</th>
</tr>
</thead>
<tbody>
<tr>
<td>ABSTRACT</td>
<td>i</td>
</tr>
<tr>
<td>ACKNOWLEDGEMENTS</td>
<td>iii</td>
</tr>
<tr>
<td>TABLE OF CONTENTS</td>
<td>v</td>
</tr>
<tr>
<td>LIST OF FIGURES</td>
<td>ix</td>
</tr>
<tr>
<td>NOMENCLATURE</td>
<td>xiv</td>
</tr>
</tbody>
</table>

## CHAPTER 1

### INTRODUCTION AND LITERATURE SURVEY

1.1 INTRODUCTION .................................................. 1

1.2 THE LITERATURE SURVEY ........................................ 5

1.2.1 HEAT TRANSFER ANALYSIS .................................. 5

1.2.2 VIBRATORY ANALYSIS ..................................... 6

1.2.3 FATIGUE LIFE ANALYSIS .................................. 7

1.3 OBJECTIVES .................................................. 8

## CHAPTER 2

### TRANSIENT HEAT TRANSFER ANALYSIS

2.1 INTRODUCTION ................................................ 11

2.2 TRANSIENT TEMPERATURE DETERMINATION - THE MATHEMATICAL MODEL 13
2.3 DERIVATION OF THE ELEMENTAL EQUATIONS FOR \([CP^*], [KC^*], \{F_c^*\}, \{F_r^*\}\)

2.4 ILLUSTRATION OF THEORY

2.5 TEMPERATURE GRADIENT DISTRIBUTION ACROSS THE AIRFOIL CROSS-SECTION AT DIFFERENT HEIGHTS OF THE BLADE

2.6 TRANSIENT THERMAL STRESS DETERMINATION - THE MATHEMATICAL MODEL

2.7 CONCLUSION

<table>
<thead>
<tr>
<th>CHAPTER 3</th>
</tr>
</thead>
<tbody>
<tr>
<td>FREE VIBRATION ANALYSIS</td>
</tr>
<tr>
<td>3.1 INTRODUCTION</td>
</tr>
<tr>
<td>3.2 MATHEMATICAL FORMULATION</td>
</tr>
<tr>
<td>3.2.1 ([K_s^*]) ELEMENTAL CONVENTIONAL STIFFNESS MATRIX - THE MATHEMATICAL FORMULATION</td>
</tr>
<tr>
<td>3.2.2 ([K_c^*]) ELEMENTAL STRESS STIFFNESS MATRIX - THE MATHEMATICAL FORMULATION</td>
</tr>
<tr>
<td>3.2.3 ([M^*]) ELEMENTAL MASS MATRIX - THE MATHEMATICAL FORMULATION</td>
</tr>
<tr>
<td>3.3 FREE VIBRATION ANALYSIS</td>
</tr>
</tbody>
</table>
3.4 EFFECT OF TEMPERATURE VARIATION ON THE NATURAL FREQUENCIES OF THE BLADE

3.5 CONCLUSION

CHAPTER 4
TRANSIENT STRESS ANALYSIS & FATIGUE LIFE ESTIMATION

4.1 INTRODUCTION

4.2 TRANSIENT RESPONSE OF A ROTOR BLADE DUE TO NOZZLE EXCITATION FORCES - THE MATHEMATICAL FORMULATION

4.3 TRANSIENT RESPONSE DUE TO CENTRIFUGAL FORCE - THE MATHEMATICAL FORMULATION

4.4 VIBRATORY STRESSES

4.5 CENTRIFUGAL STRESSES

4.6 THERMAL STRESSES

4.7 STRESS ANALYSIS OF BLADE DUE TO COMBINED EFFECTS

4.8 FATIGUE LIFE ESTIMATION

4.9 CUMULATIVE FATIGUE DAMAGE

4.10 CONCLUSION

CHAPTER 5
CONCLUSIONS AND RECOMMENDATIONS

5.1 A BRIEF DISCUSSION ABOUT THE INVESTIGATION
AND THE CONCLUSIONS

5.2 LIMITATIONS OF THE PRESENT INVESTIGATION
AND RECOMMENDATIONS FOR FUTURE WORK

REFERENCES

APPENDIX A EXPRESSIONS FOR SHAPE FUNCTIONS
OF CORNER NODES FOR THE TWENTY-NODED
ELEMENT

APPENDIX B EXPRESSIONS FOR SHAPE FUNCTIONS
OF MID-SIDE NODES FOR THE TWENTY-NODED
ELEMENT

APPENDIX C EXPRESSIONS FOR ELEMENTAL MATRICES
AND VECTORS FOR THE TWENTY-NODED
ELEMENT

APPENDIX D VARIATION OF MATERIAL PROPERTIES
OF THE TURBINE BLADE WITH TEMPERATURE

APPENDIX E PHYSICAL INTERPRETATION OF
STRESS STIFFNESS MATRIX
# LIST OF FIGURES

<table>
<thead>
<tr>
<th>NO.</th>
<th>DESCRIPTION</th>
<th>PAGE</th>
</tr>
</thead>
<tbody>
<tr>
<td>1.1</td>
<td>CUTAWAY DIAGRAM OF LYCOMING T53 - L - 13</td>
<td>2</td>
</tr>
<tr>
<td>1.2</td>
<td>BURNT FIRST STAGE TURBINE BLADES</td>
<td>3</td>
</tr>
<tr>
<td>1.3</td>
<td>CROSS SECTION OF FATIGUED TURBINE BLADE</td>
<td>4</td>
</tr>
<tr>
<td>2.1</td>
<td>THREE DIMENSIONAL FINITE ELEMENT MODEL for turbine blade</td>
<td>12</td>
</tr>
<tr>
<td>2.2</td>
<td>SOLID ISOPARAMETRIC, SERENDIPITY, C⁰ CONTINUITY, 20-NODED ELEMENT (LOCAL COORDINATE SYSTEM)</td>
<td>17</td>
</tr>
<tr>
<td>2.3</td>
<td>SOLID ISOPARAMETRIC, SERENDIPITY, C⁰ CONTINUITY, 20-NODED ELEMENT (GLOBAL COORDINATE SYSTEM)</td>
<td>18</td>
</tr>
<tr>
<td>2.4</td>
<td>NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS at leading edge</td>
<td>34</td>
</tr>
<tr>
<td>2.5</td>
<td>NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS at 15 % CHORD LENGTH</td>
<td>35</td>
</tr>
<tr>
<td>2.6</td>
<td>NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS at 30 % CHORD LENGTH</td>
<td>36</td>
</tr>
<tr>
<td>2.7</td>
<td>NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS at 45 % CHORD LENGTH</td>
<td>37</td>
</tr>
<tr>
<td>2.8</td>
<td>NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS at 60 % CHORD LENGTH</td>
<td>38</td>
</tr>
</tbody>
</table>
2.9  NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS

AT 75 % CHORD LENGTH

2.10  NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS

AT TRAILING EDGE

2.11  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION

AT t = 150 SEC (HEIGHT OF THE BLADE = 2.5 cm.)

2.12  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION

AT t = 150 SEC (HEIGHT OF THE BLADE = 5.0 cm.)

2.13  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION

AT t = 150 SEC (HEIGHT OF THE BLADE = 7.5 cm.)

2.14  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION

AT t = 150 SEC (HEIGHT OF THE BLADE = 10.0 cm.)

2.15  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION

AT t = 150 SEC (HEIGHT OF THE BLADE = 12.0 cm.)

2.16  TEMPERATURE GRADIENTS ACROSS THE BLADE

CROSS-SECTION AT t = 150 SEC.

(HEIGHT OF THE BLADE = 2.5 cm.)

2.17  TEMPERATURE GRADIENTS ACROSS THE BLADE

CROSS-SECTION AT t = 150 SEC.

(HEIGHT OF THE BLADE = 5.0 cm.)

2.18  TEMPERATURE GRADIENTS ACROSS THE BLADE

CROSS-SECTION AT t = 150 SEC.

x
2.19 TEMPERATURE GRADIENTS ACROSS THE BLADE CROSS-SECTION AT t = 150 SEC.

(HEIGHT OF THE BLADE = 10.0 cm.) 49

2.20 TEMPERATURE GRADIENTS ACROSS THE BLADE CROSS-SECTION AT t = 150 SEC.

(HEIGHT OF THE BLADE = 12.0 cm.) 50

2.21 TRANSIENT THERMAL STRESS ($\sigma_{xx}$) DISTRIBUTION IN THE BLADE 59

2.22 TRANSIENT THERMAL STRESS ($\sigma_{yy}$) DISTRIBUTION IN THE BLADE 60

2.23 TRANSIENT THERMAL STRESS ($\sigma_{zz}$) DISTRIBUTION IN THE BLADE 61

2.24 TRANSIENT THERMAL STRESS ($\sigma_{zz}$) DISTRIBUTION IN THE BLADE FOR THE FEASIBLE HEATING RATE OF 24° C/SEC AT DIFFERENT INSTANTS OF TIME 62

3.1 TURBINE BLADE AIRFOIL CROSS-SECTION AT THE ROOT AND THE TIP 66

3.2 VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 300 RPM 78

3.3 VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 900 RPM 79
3.4 VARIATION OF FIRST FIVE NATURAL FREQUENCIES
OF TURBINE BLADE AT ROTOR SPEED OF 1500 RPM 80
3.5 VARIATION OF FIRST FIVE NATURAL FREQUENCIES
OF TURBINE BLADE AT ROTOR SPEED OF 2100 RPM 81
3.6 VARIATION OF FIRST FIVE NATURAL FREQUENCIES
OF TURBINE BLADE AT ROTOR SPEED OF 2700 RPM 82
3.7 VARIATION OF FIRST FIVE NATURAL FREQUENCIES
OF TURBINE BLADE AT ROTOR SPEED OF 3300 RPM 83
3.8 VARIATION OF FIRST FIVE NATURAL FREQUENCIES
OF TURBINE BLADE AT ROTOR SPEED OF 3900 RPM 84
3.9 COMPARISON OF FIRST ACTUAL CRITICAL SPEED
WITH FIRST PROJECTED CRITICAL SPEED
USING STRESS STIFFNESS MATRIX 85
3.10 COMPARISON OF SECOND ACTUAL CRITICAL SPEED
WITH SECOND PROJECTED CRITICAL SPEED
USING STRESS STIFFNESS MATRIX 86
3.11 COMPARISON OF FIRST CRITICAL SPEED OBTAINED
FROM VARIOUS METHODS 87
3.12 COMPARISON OF SECOND CRITICAL SPEED OBTAINED
FROM VARIOUS METHODS 88
3.13 COMPARISON OF THIRD CRITICAL SPEED OBTAINED
FROM VARIOUS METHODS 89
3.14 COMPARISON OF FOURTH CRITICAL SPEED OBTAINED FROM VARIOUS METHODS 90
3.15 COMPARISON OF FIFTH CRITICAL SPEED OBTAINED FROM VARIOUS METHODS 91
4.1 REPRESENTATION OF THE NOZZLE EXCITATION FORCES AS A SINUSOIDAL PULSE 97
4.2 VIBRATORY STRESSES AT VARIOUS ROTOR SPEEDS 105
4.3 CENTRIFUGAL STRESSES AT VARIOUS ROTOR SPEEDS 106
4.4 THERMAL STRESSES AT VARIOUS TIMES 107
4.5 TRANSIENT THERMAL STRESS ($\sigma_{zz}$) DISTRIBUTION IN THE BLADE FOR THE FEASIBLE HEATING RATE OF 24°C/SEC AT DIFFERENT INSTANTS OF TIME 108
4.6 FATIGUE FAILURE SURFACE DEFINED BY BAGCI LINE 113
4.7 BLADE LOADING PATTERN 114
4.8 MAGNIFIED STRESS LEVELS BETWEEN 3100-3200 RPM DURING START-UP AT ROTOR ACCELERATION OF 800 RPM/MIN 116
4.9 MAGNIFIED STRESS LEVELS BETWEEN 3100-3200 RPM DURING START-UP AT ROTOR ACCELERATION OF 1500 RPM/MIN 117
**NOMENCLATURE**

\[ k_{xx} \quad \text{thermal conductivity in the x-direction} \]

\[ k_{yy} \quad \text{thermal conductivity in the y-direction} \]

\[ k_{zz} \quad \text{thermal conductivity in the z-direction} \]

\[ Q \quad \text{heat generated within the body} \]

\[ \rho \quad \text{mass density of the material} \]

\[ c \quad \text{specific heat} \]

\[ l_x, l_y, l_z \quad \text{direction cosines normal to the surface} \]

\[ h \quad \text{convective heat transfer coefficient} \]

\[ T_w \quad \text{gas temperature} \]

\[ \sigma \quad \text{Stefan-Boltzmann constant} \]

\[ \epsilon \quad \text{emissivity of the body} \]

\[ \chi \quad \text{a functional} \]

\[ V \quad \text{volume of the blade} \]

\[ q \quad \text{specified heat flux} \]

\[ \delta T \quad \text{first variation of } T \]

\[ S_1 \quad \text{surface experiencing heat flux} \]

\[ S_2 \quad \text{surface experiencing convection and radiation boundary conditions} \]

\[ [D] \quad \text{thermal conductivity matrix} \]

\[ [B] \quad \text{temperature gradient interpolation matrix} \]
\[ [N] \quad \text{shape function matrix} \]
\[ \{T\} \quad \text{nodal temperature vector} \]
\[ [CP] \quad \text{capacitance matrix} \]
\[ [KC] \quad \text{conduction matrix} \]
\[ \{F_c\} \quad \text{force vector due to convection} \]
\[ \{F_r\} \quad \text{force vector due to radiation} \]
\[ \xi, \eta, \zeta \quad \text{local coordinate directions} \]
\[ [J] \quad \text{jacobian matrix} \]
\[ |J| \quad \text{determinant of jacobian matrix} \]
\[ W_p, W_p, W_k \quad \text{weights used in Gauss-quadrature} \]
\[ G \quad \text{Global} \]
\[ e \quad \text{element} \]
\[ t \quad \text{time} \]
\[ \Delta t \quad \text{time increment} \]
\[ \{g\} \quad \text{temperature gradient vector} \]
\[ \{\sigma\} \quad \text{stress vector} \]
\[ [D,] \quad \text{material property matrix} \]
\[ \{e\} \quad \text{strain vector} \]
\[ \{e_0\} \quad \text{initial strain vector} \]
\[ \sigma_{xx}, \sigma_{yy}, \sigma_{zz} \quad \text{normal stresses} \]
\[ \sigma_{xy}, \sigma_{yz}, \sigma_{zx} \quad \text{shear stresses} \]
\[ \varepsilon_{xx}, \varepsilon_{yy}, \varepsilon_{zz} \quad \text{normal strains} \]
\[ \varepsilon_{xy}, \varepsilon_{xx}, \varepsilon_{yy} \] shear strains

\[ [B_{ij}] \] strain displacement matrix

\[ E \] modulus of elasticity

\[ \nu \] Poisson's ratio of the material

\[ \alpha \] coefficient of thermal expansion

\[ \{f\} \] force vector due to thermal expansion

\[ \{d\} \] displacement vector

\[ [K_{ij}] \] conventional stiffness matrix

\[ [K_{st}] \] stress stiffness matrix

\[ \{c\} \] nodal local coordinate vector

\[ U \] strain energy

\[ [M] \] mass matrix

\[ \lambda \] eigenvalue of the system

\[ [C] \] damping matrix

\[ \omega \] angular speed of the rotor

\[ \omega_{nk} \] natural frequency of \( k^{th} \) mode

\[ \omega_{dk} \] damped natural frequency of \( k^{th} \) mode

\[ \zeta_k \] damping ratio of \( k^{th} \) mode

\[ [\Phi] \] matrix of modal vectors

\[ \{F\} \] dynamic force vector

\[ \{Q\} \] force vector transformed in modal coordinates by normalized modal matrix
\{QF\} \quad \text{force vector transformed in modal coordinates by}

\text{m-orthonormalized modal matrix}

n \quad \text{number of elements}

G \quad \text{global}

e \quad \text{elemental}
CHAPTER 1
INTRODUCTION AND LITERATURE REVIEW

1.1 INTRODUCTION

In the fast few decades, the turbines have found an ever increasing service in the field of power generation and in the propulsion of jet engines. The major gas turbine components are shown in the Fig. 1.1. The air, as we see in this figure, is let in through an air-inlet or a diffuser, it is then compressed and led to combustor where the temperature of high-pressure air is increased. The air-fuel mixture is directed through the nozzles onto the turbine blade and thus converting energy of the gas, evidenced by a high pressure and temperature, into kinetic energy and then to the shaft work.

As is evident, the major brunt of the energy conversion is thus taken up by the turbine blade which is exposed to a variety of hostile environment and forces such as thermal, centrifugal and vibratory. All these causes (separately and collectively) can lead to turbine blade failure which is a major cause of breakdown in the turbomachinery. Figs. 1.2 and 1.3 show some of the examples of turbine blades which have undergone failure. Thus, it is essential to have an accurate estimate of the damage done by these forces on the turbine blade in the early stages of design which, in turn, can avoid the causes of the early blade failure, which is the objective of this investigation. This objective is achieved in two stages: at first, the heat transfer analysis of turbine blade is done and transient temperatures of the blade at different time steps are found. These temperatures help in finding the thermal gradients and transient thermal stresses.
Fig. 1.1  CUTAWAY DIAGRAM OF LYCOMING T53 - L - 13 [Treager, 1970]
Fig. 1.2  BURNT FIRST STAGE TURBINE BLADES[Boyce, 1982]
Fig. 1.3 CROSS SECTION OF FATIGUED TURBINE BLADE.

MARKINGS NEAR THE TRAILING EDGE INDICATE CYCLIC FATIGUE [Boyce, 1982]
The second part consists in the determination of natural frequencies of the turbine blade in the transient phase. This is used in finding the critical speeds, the vibratory stresses and the centrifugal stresses. When combined together, this gives the overall picture of the stresses in the turbine blade and finally, the fatigue life estimation of the turbine blade is done.

1.2 THE LITERATURE SURVEY

1.2.1 HEAT TRANSFER ANALYSIS

Since the advent of gas turbines and their subsequent application in various sectors of industry, there has been a rising demand for more and more power output. The blade loadings have continually risen typically in increased hostile environment due to temperature, wetness, speed and corrosion. These developments demand (a) greater output per blade, (b) smaller frame size or greater output per unit volume (c) higher operating speeds, and (d) higher processing temperatures and pressures. Above all, the reliability is the most demanding requirement. In general, simultaneous satisfaction of these demands has called for an increased attention to the accurate heat transfer and vibratory analysis of turbine blades.

Daniels et al (1981) and Mohanty et al (1977) reported some of the latest developments in heat transfer analysis of turbine blades. Maya et al (1978) and Mukherjee (1978) calculated the transient thermal stresses of the gas turbine blades. However, their studies included the heat exchange process by convection only and thus their temperature and thermal stress distribution studies of the turbine blade were
inadequate. One of the major deficiencies of these studies was the non-inclusion of the radiative heat exchange terms which are quite significant at the higher operating temperatures. Moreover, their investigation was based on the assumption that the maximum stress would take place only at one of the points at leading edge, trailing edge or mid-section of the turbine blade. Allen (1982) studied the effect of temperature dependent mechanical properties on thermal stresses in turbine blades. The influence of turbine blade geometry on its thermal stress state was studied by Bogov (1978). The experimental verification of finite element calculation of the thermal stress state of gas turbine blades was done by Gryaznov et al (1979). Bahree (1987) carried out the two-dimensional heat transfer analysis of the turbine blade using finite element analysis. His investigation was based on the assumption that the thermal gradients along the height of the turbine blade are negligible during the transient state as compared to the thermal gradients along the airfoil cross-section of the blade.

1.2.2 VIBRATORY ANALYSIS

The dynamic analysis of the turbine blade has been a challenging field for the engineers for a long time. Numerous mathematical approaches have been used to derive the dynamic equations. Most of the mathematical techniques were based on the Newtonian approach and, as a first approximation, the geometry of the turbine blade was modelled as a tapered beam having a rectangular cross-section. Rao and Carnegie (1970) used the Ritz-Galerkin method to obtain the bending frequencies for the first three modes of vibration of tapered cantilever blades having rectangular cross-section. Mabie
and Rogers (1974) analyzed the effect of various boundary conditions on the natural frequency of the turbine blade. Their method of solution was based on the use of the Bessel's functions. Sato (1980) used Ritz method to carry out the effect of axial force on the frequencies of blades with ends restrained elastically against rotation. Sisto and Chang (1984) formulated a finite element model to calculate the natural frequencies of the blade. Their model, however, was appropriate for thin and high aspect ratio blades only. Nagarajan and Alwar (1984) used the twenty-noded finite element to analyze the free vibration behaviour of the blade without temperature variation. Bahree (1987) did the transient free vibration analysis of the turbine blade using twenty-noded finite element for the turbine blade which included the effect of the change in temperature. The effect of the stress stiffness matrix, caused by the rotation of the turbine blade, was not carried out in his study. Warikoo (1989) analyzed the propeller-shaft transverse vibrations. His work included the non-linear stiffness variation but, it did not include the effect of change of the temperature on the transient natural frequencies of the turbine blade.

1.2.3 FATIGUE LIFE ANALYSIS

In spite of the fact that various design procedures for preventing failure of turbine blades due to fatigue are being used, there is yet no indication that a satisfactory situation exists with reference to basic data that will lead to establishing sound procedures in this regard. This is because of the fact that blade fatigue is a multi-dimensional problem and as such has been undergoing continuous investigations for a long time. The various reasons which lead the blade to failure are: high gas temperature, blade excitation.

1.3 OBJECTIVES

Based on the survey of the previous research works, and more specifically, Bahree’s (1987) work where (a) two-dimensional heat transfer analysis was done, (b) the non-linear terms in the stiffness matrix were neglected, and (c) the gas force excitations were modelled as impulses of an arbitrary height, in this work a three dimensional heat transfer model is analyzed; the non-linear stiffness terms have been included, and the gas force excitations are based on fluid flow over an airfoil cross section.

1. To derive the transient temperature distribution equations in a turbine blade using twenty-noded isoparametric finite elements.

2. To determine the temperatures and thermal gradients at various points on the
airfoil cross-section and along the height of the turbine blade by solving a non-linear system of equations. The non-linearity arises due to the radiative boundary conditions along the surface of the turbine blade and also due to the variation of the material properties.

3. To find the transient thermal stresses due to the transient temperature obtained.

4. To study the free vibration characteristics of the turbine blade using twenty-noded isoparametric finite elements.

5. To study the effect of the stress stiffness matrix and the change in the material properties of the blade (as temperature changes) on the natural frequencies of the turbine blade.

6. To study the dynamic stresses in the turbine blade due to nozzle excitation forces modelled as a sinusoidal pulse.

7. To study the effect of rotor speed on centrifugal forces.

8. To study the combined stress analysis in the turbine blade due to the effects of all of the stresses mentioned above.

9. To study the fatigue life estimation of the turbine blade due to the combination of thermal, vibratory and centrifugal stresses.

In Chapter 2, a mathematical model for the three-dimensional heat transfer analysis of the turbine blade is established. The non-linear equations for the transient temperature determination in the turbine blade are obtained by using a twenty-noded isoparametric finite element formulation. These non-linear system of differential equations are
expressed as a system of algebraic equations in the time domain by using the Crank-Nicolson finite difference scheme. The temperatures, thermal gradients, and stresses at various points on the airfoil cross-section and along the height of the blade are then determined by solving these non-linear system of equations. A feasible heating path for the gas turbine is established to carry out the thermal stress analysis.

In Chapter 3, the three-dimensional mathematical model for the turbine blade is used to study the free vibration characteristics of the turbine blade. The effect of stress stiffness matrix caused by the rotation of the turbine blade, on the natural frequency of the turbine blade is studied. The effect of the change in the material properties of the blade (as temperature changes) on the natural frequencies of the turbine blade is also studied.

In Chapter 4, the nozzle excitation forces have been modelled as a sinusoidal pulse. The responses due to nozzle excitation and centrifugal forces are used to calculate the dynamic stresses in the turbine blade. The fatigue life of the turbine blade is estimated based on the thermal and dynamic stresses obtained.

Finally, the conclusions and recommendations for future work are presented in Chapter 5.
CHAPTER 2

TRANSIENT HEAT TRANSFER ANALYSIS

2.1 INTRODUCTION

In order to increase the efficiency of a turbine, it is very essential that higher inlet pressures and temperatures be used. However, higher inlet temperatures, though important from thermodynamic point of view, severely increase the thermal loading on the turbine blade, and as a result, high thermal stresses are induced in the turbine blade. The impact of these thermal stresses is known to peak both, during the acceleration, and the deceleration stages of the gas turbine engine. The reason is that there is a significant temperature difference which is sufficiently large to cause plastic deformation (Bahree, 1987) at various points of the blade. Hence, it becomes very important to calculate the temperature distribution within the turbine blade with great accuracy. The present investigation is based on the three-dimensional analysis of the turbine blade. The mathematical model for the non-linear transient heat transfer analysis is formulated using curved, solid, \( C^0 \) continuity, serendipity, twenty-noded isoparametric finite elements. This type of element is chosen because of its versatility in accurately mapping the complex geometry of the turbine blade. The results obtained from three-dimensional model show that in addition to the thermal gradients along the airfoil cross-section of the turbine blade, there are significant thermal gradients along the height of the turbine blade, as well. These elements of the turbine blade (shown in Fig. 2.1) exchange heat with the surrounding high temperature gas by convection and radiation.
Fig. 2.1  THREE DIMENSIONAL FINITE ELEMENT MODEL FOR TURBINE BLADE
Because of the radiation term, we get a non-linear system of differential equations. The Crank-Nicolson finite difference scheme is then used to transform these non-linear differential equations into non-linear algebraic equations in the time domain. Then, using an iteration technique, the non-linear algebraic equations are solved to obtain the nodal and elemental temperatures. Once these transient nodal temperatures are known, the thermal gradients (a) across the airfoil cross-section and (b) along the height of the turbine blade, are calculated. Using these transient temperatures, one can calculate the thermal stresses. In order to keep these thermal stresses well within the yield stress limit, a feasible heating path is also obtained in the present investigation.

2.2 TRANSIENT TEMPERATURE DETERMINATION - THE MATHEMATICAL MODEL

In the following derivation, the general form of the three-dimensional equations are obtained first and then the appropriate conditions relevant to our problem are applied. The governing three-dimensional partial differential equation for heat flow in any solid can be written as

\[ \frac{k_x \partial^2 T}{\partial x^2} + \frac{k_y \partial^2 T}{\partial y^2} + \frac{k_z \partial^2 T}{\partial z^2} + Q = \rho c \frac{\partial T}{\partial t} \]  

(2.1)

and its boundary condition as

\[ k_x \frac{\partial T}{\partial x} l_x + k_y \frac{\partial T}{\partial y} l_y + k_z \frac{\partial T}{\partial z} l_z + h(T - T_m) + q + \sigma \varepsilon (T^4 - T_m^4) = 0 \]

(2.2)

We make use of the calculus of variations to solve the partial differential equation, Eq.
(2.1) and the boundary condition, Eq. (2.2) associated with it. This solution is obtained by minimizing the corresponding variational functional and solving the resulting system of algebraic equations. We can rearrange the Eq. (2.1) and write it as

$$-k_x \frac{\partial^2 T}{\partial x^2} - k_y \frac{\partial^2 T}{\partial y^2} - k_z \frac{\partial^2 T}{\partial z^2} - \left\{Q - \rho c \frac{\partial T}{\partial t}\right\} = 0 \quad (2.3)$$

Now the variational functional can be set up by multiplying Eq. (2.2) and Eq. (2.3) by the first variation of $T$ i.e., $\delta T$ and integrating over the whole domain. This results in the equation

$$\delta_x = \int_v \left[ -k_x \frac{\partial^2 T}{\partial x^2} - k_y \frac{\partial^2 T}{\partial y^2} - k_z \frac{\partial^2 T}{\partial z^2} - \left\{Q - \rho c \frac{\partial T}{\partial t}\right\} \right] \delta T \, dV \quad (2.4)$$

$$+ \int_s \left[ k_x \frac{\partial T}{\partial x} I_x + k_y \frac{\partial T}{\partial y} I_y + k_z \frac{\partial T}{\partial z} I_z + q + h(T - T_\infty) + \sigma E(T^4 - T_\infty^4) \right] \delta T \, dS$$

Rearranging Eq.(2.4) one gets

$$\delta_x = \int_s k_x \frac{\partial T}{\partial x} I_x \delta T \, dS - \int_v k_x \frac{\partial^2 T}{\partial x^2} \delta T \, dV + \int_s k_y \frac{\partial T}{\partial y} I_y \delta T \, dS - \int_v k_y \frac{\partial^2 T}{\partial y^2} \delta T \, dV$$

$$+ \int_s k_z \frac{\partial T}{\partial z} I_z \delta T \, dS - \int_v k_z \frac{\partial^2 T}{\partial z^2} \delta T \, dV + \int q \delta T \, dS - \int_v \left( Q - \rho c \frac{\partial T}{\partial t} \right) \delta T \, dV$$

$$+ \int h(T - T_\infty) \delta T \, dS + \int \sigma E(T^4 - T_\infty^4) \delta T \, dS \quad (2.5)$$
Using Green's Divergence Theorem, we can transform surface integral into volume integral. Thus the integral in Eq. (2.5) can be written as:

\[ \int k_x \frac{\partial T}{\partial x} l_x \delta T \, dS = \int k_x \frac{\partial^2 T}{\partial x^2} \delta T \, dV + \int \frac{k_x}{2} \delta \left( \frac{\partial T}{\partial x} \right)^2 dV \quad (2.6) \]

\[ \int k_y \frac{\partial T}{\partial y} l_y \delta T \, dS = \int k_y \frac{\partial^2 T}{\partial y^2} \delta T \, dV + \int \frac{k_y}{2} \delta \left( \frac{\partial T}{\partial y} \right)^2 dV \quad (2.7) \]

and

\[ \int k_z \frac{\partial T}{\partial z} l_z \delta T \, dS = \int k_z \frac{\partial^2 T}{\partial z^2} \delta T \, dV + \int \frac{k_z}{2} \delta \left( \frac{\partial T}{\partial z} \right)^2 dV \quad (2.8) \]

Substituting the integral from the Eqs. (2.6) to (2.8) into Eq. (2.5), the new equation will be

\[ \delta_x = \int \frac{k_x}{2} \delta \left( \frac{\delta T}{\delta x} \right)^2 dV + \int \frac{k_y}{2} \delta \left( \frac{\delta T}{\delta y} \right)^2 dV \]

\[ + \int \frac{k_z}{2} \delta \left( \frac{\delta T}{\delta z} \right)^2 dV - \int q \delta T \, dV \]

\[ + \int \rho c \frac{\delta T}{\delta t} \delta T \, dV \]

\[ + \int q \delta T \, dS + \int \frac{h}{2} \delta \left( T - T_m \right)^2 ds + \int \sigma \varepsilon \delta \left( \frac{T^5}{5} - T_m^4 \right) \, dS \quad (2.9) \]

Removing the variational operator \( \delta \) from both sides of the equation above, we get
\[
\chi = \int \frac{1}{2} \left[ k_x \left( \frac{\delta T}{\delta x} \right)^2 + k_y \left( \frac{\delta T}{\delta y} \right)^2 + k_z \left( \frac{\delta T}{\delta z} \right)^2 - 2 Q T + 2 \rho c T \frac{\delta T}{\delta t} \right] \\
+ \int_{S_1} q \ T \ dS + \int_{S_2} \left[ \frac{h}{2} (T - T_{in})^2 + \sigma \ v \left( \frac{T^5}{5} - T_{in}^4 \right) \right] \ dS
\]

where \( S_1 \) is the surface experiencing heat flux and \( S_2 \) is the surface experiencing convection and radiation boundary conditions.

Referring to Fig. 2.2 where there are 20-nodes per element, one can use the finite element equalities mentioned below to formulate the finite element equations corresponding to Eq. (2.10). Also, the temperature \( T \) is assumed continuous not over the whole domain but it is defined over an individual element. The equations are:

\[
[D^e] = \begin{bmatrix}
  k_x^e & 0 & 0 \\
  0 & k_y^e & 0 \\
  0 & 0 & k_z^e \\
\end{bmatrix}
\]

if \( k_x = k_y = k_z = k \) then

\[
[D^e] = \begin{bmatrix}
  1 & 0 & 0 \\
  0 & 1 & 0 \\
  0 & 0 & 1 \\
\end{bmatrix}
\]

(2.11)
Fig. 2.2 SOLID ISOPARAMETRIC, SERENDIPITY, C° CONTINUITY, 20-NODED ELEMENT (LOCAL COORDINATE SYSTEM)
Fig. 2.3  SOLID ISOPARAMETRIC, SERENDIPITY, C° CONTINUITY, 20-NODED ELEMENT (GLOBAL COORDINATE SYSTEM)
where \( T_1, T_2, \ldots, T_{20} \) are the nodal temperatures of a twenty-noded finite element. Also, both the terms \( Q \) and \( q \) in the Eq (2.10) are zero as there is no heat generation and no specified heat fluxes in the present investigation. Therefore, substituting Eqs. (2.11-2.13) into Eq. (2.10), we obtain

\[
\chi^e = \int_{V^e} \frac{1}{2} \left\{ \begin{array}{c} T^e \\ \end{array} \right\}^T [ D^e ] [ B^e ] \{ T^e \} \ dV
\]

\[
+ \int_{S^e} \frac{h^e}{2} \left( \left[ \begin{array}{c} N^e \\ \end{array} \right] \{ T^e \} \right)^2 \ dS
\]
The functional $\chi$ is defined for individual 20-noded element. If we minimize $\chi$ with respect to the nodal temperature vector, $\{T^e\}$, we will get a stationary value of $\chi$ which will satisfy Eqs. (2.1) and (2.2). The minimization of $\chi$ can be written as

$$\frac{\partial \chi}{\partial \{T^e\}} = \frac{\partial \chi^1}{\partial \{T^e\}} + \frac{\partial \chi^2}{\partial \{T^e\}} + \ldots + \frac{\partial \chi^n}{\partial \{T^e\}} = 0$$

where $n$ stands for the number of elements.

The following relationships as given by Segerlind (1976) are used to differentiate Eq. (2.14) with respect to $\{T^e\}$:

$$\frac{\partial}{\partial \{T^e\}} \left( \frac{1}{2} \{ T^e \}^T \{ B^e \}^T \{ D^e \} \{ B^e \} \{ T^e \} \right) dV$$
\[ \int_V \left[ \mathbf{B}^e \right]^T \left[ \mathbf{D}^e \right] \left[ \mathbf{B} \right] \left[ \mathbf{T}^e \right] \, dV \quad ; \quad (2.16) \]

\[ \frac{\partial}{\partial \{T^e\}} \int_V \rho \, c^e \left[ \left[ \mathbf{N}^e \right] \{T^e\} \right] \, dV \quad ; \quad (2.17) \]

\[ \int_V \rho \, c^e \left[ \left[ \mathbf{N}^e \right] \frac{\partial \{T^e\}}{\partial t} \right] \left[ \mathbf{N}^e \right]^T \, dV \quad ; \quad (2.17) \]

\[ \frac{\partial}{\partial \{T^e\}} \int_{S^e} \frac{h^e}{2} \left( \left[ \left[ \mathbf{N}^e \right] \{T^e\} \right]^2 \right) \, dS = \int_{S^e} h^e \left[ \left[ \mathbf{N}^e \right] \{T^e\} \right] \left[ \mathbf{N}^e \right]^T \{T^e\} \, dS \quad ; \quad (2.18) \]

\[ \frac{\partial}{\partial \{T^e\}} \int_{S^e} h^e \, T_+ \left[ \left[ \mathbf{N}^e \right] \{T^e\} \right] \, dS = \int_{S^e} h^e \, T_+ \left[ \mathbf{N}^e \right]^T \{T^e\} \, dS \quad ; \quad (2.19) \]

\[ \frac{\partial}{\partial \{T^e\}} \int_{S^e} \frac{\sigma \, \varepsilon}{5} \left( \left[ \left[ \mathbf{N}^e \right] \{T^e\} \right]^s \right) \, dS = \int_{S^e} \sigma \, \varepsilon \left[ \left[ \mathbf{N}^e \right]^T \left( \left[ \mathbf{N}^e \right] \{T^e\} \right)^s \right] \, dS \quad ; \quad (2.20) \]
Substituting these relationships into the Eq. (2.14) and summing up the contributions of each element as per Eq. (2.15), we get

\[
\frac{\partial \chi}{\partial \{T^e\}} = \sum_{i} \int \left[ B^e \right]^T \left[ D^e \right] \left[ B^e \right] \left\{ T^e \right\} \ dV \\
+ \int \rho \ c^e \left[ N^e \right] \frac{\partial \left\{ T^e \right\}}{\partial t} \left[ N^e \right]^T \ dS \\
+ \int h^e \left[ N^e \right]^T \left[ N^e \right] \left\{ T^e \right\} \ dS - \int h^e \left[ T^e \right]^T \left[ N^e \right]^T \ dS \\
+ \int \sigma \ \varepsilon \left[ N^e \right]^T \left( \left[ N^e \right] \left\{ T^e \right\} \right) \ dS \\
- \int \sigma \ \varepsilon \left[ T^e \right]^T \left[ N^e \right]^T \ dS = 0
\]  

(2.23)

The expressions for various elemental matrices are:

**ELEMENTAL CAPACITANCE MATRIX** \([CP^e]\) can be expressed as

\[
\left[ CP^e \right] = \int \rho \ c^e \left[ N^e \right]^T \left[ N^e \right] \ dV ,
\]  

(2.24)

**ELEMENTAL CONDUCTION MATRIX** \([KC^e]\) can be expressed as
\[
\mathbf{K} \mathbf{C}^\circ = \int_{V^*} \left( \mathbf{B}^\circ \right)^T \mathbf{D}^\circ \mathbf{B}^\circ \, dV + \int_{S^*} \mathbf{h}^\circ \left( \mathbf{N}^\circ \right)^T \mathbf{N}^\circ \, dS ,
\]

(2.25)

The FORCE VECTOR \( \{F_c^0\} \) for the convection process can be written as

\[
\{F_c^0\} = \int_{S^*} \mathbf{h}^\circ \mathbf{T}_{c0} \left( \mathbf{N}^\circ \right)^T \, dS ,
\]

and

(2.26)

the FORCE VECTOR \( \{F_r^0\} \) for the radiation process will be

\[
\{F_r^0\} = \int_{S^*} \mathbf{\sigma} \mathbf{e} \mathbf{T}_{r0} \left( \mathbf{N}^\circ \right)^T \, dS - \int_{S^*} \mathbf{\sigma} \mathbf{e} \left( \mathbf{N}^\circ \right) \left( \left[ \mathbf{N}^\circ \right] \{T^0\} \right)^T \, dS
\]

(2.27)

In case of isoparametric finite elements (shown in Fig. 2.3), the most striking feature is that their sides may be curved and that they make use of a special coordinate system \((\xi, \eta, \zeta)\). The shape functions, \([N_i]\), are defined for the variation of the local coordinate system from +1 to -1. Therefore we can have: \(\xi = \pm 1\); \(\eta = \pm 1\); and \(\zeta = \pm 1\). The shape functions \([N_i]\), for the 20-noded isoparametric element can be expressed as [Zienkiewicz, 1971]:

**Corner Nodes**

\[
N_i = \frac{1}{8} \left( 1 + \xi \xi_i \right) \left( 1 + \eta \eta_i \right) \left( 1 + \zeta \zeta_i \right) \left( \xi \xi_i + \eta \eta_i + \zeta \zeta_i - 2 \right)
\]

(2.28)

The details of Eq. (2.28) are given in APPENDIX A.
Typical Mid-side nodes

\[ N_i = \frac{1}{4} \left( 1 - \xi_i^2 \right) \left( 1 + \eta_i \eta_i \right) \left( 1 + \zeta_i \zeta_i \right) \]  \hspace{1cm} \text{(2.29)}

for the case when \( \xi_i = 0, \eta_i = \pm 1, \) and \( \zeta_i = \pm 1. \) The subscript \( i \) stands for the \( i^\text{th} \) node of the twenty-noded element in Eqs. (2.28) and (2.29).

Similarly, one can write the equations corresponding to Eq. (2.29) as shown in APPENDIX B.

The expression for elemental matrix \([CP^e]\) in terms of \( \xi, \eta, \zeta \) can be written as

\[
[CP^e] = \int_{-1}^{1} \int_{-1}^{1} \int_{-1}^{1} \rho c^e \left[ N^e(\xi, \eta, \zeta) \right]^T \left[ N^t(\xi, \eta, \zeta) \right] |J(\xi, \eta, \zeta)| \, d\xi d\eta d\zeta
\]  \hspace{1cm} \text{(2.30)}

The term \( |J(\xi, \eta, \zeta)| \) is the determinant of Jacobian matrix which relates the derivatives of the temperature \( \{T\} \) with respect to the local coordinate system \( \{\xi, \eta, \zeta\} \) to the derivatives of \( \{T\} \) with respect to the global coordinate system \( \{x, y, z\} \). The following chain rule evaluates the Jacobian matrix:

\[
\frac{\partial T}{\partial \xi} = \frac{\partial T}{\partial x} \frac{\partial x}{\partial \xi} + \frac{\partial T}{\partial y} \frac{\partial y}{\partial \xi} + \frac{\partial T}{\partial z} \frac{\partial z}{\partial \xi}
\]  \hspace{1cm} \text{(2.31)}

where \( T \) is a function of \( x, y, \) and \( z. \)

Hence, the Jacobian matrix \([J]\) can be expressed as

24
The element $J_{11}$ of $[J]$ can be written as

$$J_{11} = \frac{\partial x}{\partial \xi} = \frac{\partial}{\partial \xi} ([N]^e)^T \{x\}^e$$

(2.33)

$$1 \times 20 \quad 20 \times 1$$

where $\{x\}^e$ contains all the Cartesian coordinates along the x-direction of the 20-noded element with $i = 1, 2, 3, ..., 20$ and $[\{N\}]^e = \{N_1, N_2, N_3, ..., N_{20}\}$. The similar procedure is followed for the other elements of matrix $[J]$.

Therefore for the twenty-noded isoparametric element

$$[J(\xi, \eta)] = \begin{bmatrix}
\sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \xi} x_i & \sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \xi} y_i & \sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \xi} z_i \\
\sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \eta} x_i & \sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \eta} y_i & \sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \eta} z_i \\
\sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \zeta} x_i & \sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \zeta} y_i & \sum_{i=1}^{i=20} \frac{\partial N_i(\xi, \eta, \zeta)}{\partial \zeta} z_i
\end{bmatrix}$$

(2.34)

The inverse of the Jacobian matrix will be indicated as
The expression for elemental conductive matrix \( [KC^o] \) can be expressed as

\[
[KC^o] = \int \int \int \left[ B^o(\xi, \eta, \zeta) \right]^T \left[ D^o \right] \left[ B^o(\xi, \eta, \zeta) \right] |J(\xi, \eta, \zeta)| \, d\xi \, d\eta \, d\zeta
\]

\[ + \int \int h^o \left[ N^o(\xi, \eta) \right]^T \left[ N^o(\xi, \eta) \right] |J(\xi, \eta)| \, d\xi \, d\eta \]  

(2.36)

The matrix \( [B^o] \) in the equation above is expressed as

\[
[B(\xi,\eta,\zeta)] = \left[ J(\xi,\eta,\zeta) \right]^{-1}
\]

(2.37)

The expressions for elemental vectors \( \{F^s_o\} \) and \( \{F^e_o\} \) can be expressed as

\[
\{F^s_o\} = \int \int h^o \, T \left[ N^o(\xi, \eta) \right]^T \left| J(\xi, \eta) \right| \, d\xi \, d\eta
\]

(2.38)
\[
\{ F_\ell \} = \int \int \sigma \in \mathcal{E} \left[ N^e(\xi, \eta) \right]^T |J(\xi, \eta)| \ d\xi \ d\eta
- \int \int \sigma \in \left[ N^e(\xi, \eta) \right]^T \left( \left[ N^e(\xi, \eta) \right]\{ T^e(\xi, \eta) \} \right)^T |J(\xi, \eta)| \ d\xi \ d\eta
\]  

\[ (2.39) \]

2.3 **DERIVATION OF THE ELEMENTAL EQUATIONS FOR \([CP^e], [K^e], \{F^c\}, \{F^r\}\)**

Using the symbolic software package 'Maple', the evaluations of the elemental capacitance matrix \([CP^e]\), conduction matrix \([K^e]\), the force vector for the convection \((F^c)\), and the force vector for the radiation \((F^r)\) in terms of \(\xi, \eta, \zeta\) were done numerically by evaluating the volume integral in these equations. For isoparametric formulation, the elemental matrices and vectors mentioned above were integrated using Gauss quadrature technique. A quadratic polynomial for each side of the 20-noded element was used. Thus 2 Gauss points on each side of the element and 8 Gauss points in total were sufficient to integrate each of the elemental matrices. The expressions for the elemental matrices \([CP^e]\), \([K^e]\), \(\{F^c\}\), \(\{F^r\}\) for the case of a twenty-noded are given in APPENDIX C.

Using Gauss quadrature, final expressions for various matrices would be:

\[
[CP^e] = \sum_{i=1}^{2} \sum_{\lambda=1}^{2} \sum_{\mu=1}^{2} W_i W_\lambda W_\mu \rho c^e \left[ N^e(\xi, \eta, \zeta) \right]^T \left[ N^e(\xi, \eta, \zeta) \right] \ |J(\xi, \eta, \zeta)|
\]

\[ (2.40) \]
\[
[KC^g] = \sum_{i=1}^{2} \sum_{j=1}^{2} \sum_{k=1}^{2} W_i W_j W_k [B^g(\xi, \eta, \zeta)]^T [D^g] [B^g(\xi, \eta, \zeta)] J(\xi, \eta, \zeta) \\
+ \sum_{i=1}^{2} \sum_{j=1}^{2} W_i W_j h^g [N^g(\xi, \eta)]^T [N^g(\xi, \eta)] J(\xi, \eta) \\
\]
(2.41)

\[
\{F^c\} = \sum_{i=1}^{2} \sum_{j=1}^{2} W_i W_j h^g T_{\cdots} [N^g(\xi, \eta)]^T J(\xi, \eta) \\
(2.42)
\]

\[
\{F^g\} = \sum_{i=1}^{2} \sum_{j=1}^{2} W_i W_j \sigma e T^g \{N^g(\xi, \eta)\}^T J(\xi, \eta) \\
- \sum_{i=1}^{2} \sum_{j=1}^{2} W_i W_j \sigma e [N^g(\xi, \eta)]^T \left( [N^g(\xi, \eta)] \{T^g(\xi, \eta)\} \right)^T J(\xi, \eta) \\
(2.43)
\]

where \(W_p, W_p\) and \(W_k\) are corresponding weighting functions in the \(\xi, \eta,\) and \(\zeta\) directions respectively.

After getting the expressions for the elemental matrices and substituting them into the Eq. (2.23), we get the following global equations:

\[
[C^{P^g}] \frac{\partial \{T^g\}}{\partial t} + [KC^g] \{T^g\} = \{F^c\} + \{F^g\} \\
(2.44)
\]

In the Eq. (2.44), because of the non-linearity introduced by the radiation term, we get a non-linear set of partial differential equation. In the present work, we use the Crank-Nicolson finite difference method to transform the equations above into a system of non-
linear algebraic equations. Therefore in the time domain, for the time step \( \Delta t \), the first derivative of the nodal temperature vector will be

\[
\frac{d\{T^g\}_t}{dt} = \frac{\{T^g\}_{t+\frac{\Delta t}{2}} - \{T^g\}_{t-\frac{\Delta t}{2}}}{\Delta t}
\]  

(2.45)

Similar expressions can be written for \( \{T^g\}_t \) and \( \{F^g\}_t \) in the following way:

\[
\{T^g\}_t = \frac{\{T^g\}_{t+\frac{\Delta t}{2}} + \{T^g\}_{t-\frac{\Delta t}{2}}}{2}
\]  

and

\[
\{F^g\}_t = \frac{\{F^g\}_{t+\frac{\Delta t}{2}} + \{F^g\}_{t-\frac{\Delta t}{2}}}{2}
\]  

(2.47)

Substituting Eqs. (2.45), (2.46), and (2.47) into the Eq. (2.44), we get

\[
\frac{1}{\Delta t} [CP^g] \{T^g\}_{t+\frac{\Delta t}{2}} - \frac{1}{\Delta t} [CP^g] \{T^g\}_{t-\frac{\Delta t}{2}} + \frac{1}{2} [KC^g] \{T^g\}_{t+\frac{\Delta t}{2}} + \frac{1}{2} [K^g] \{T^g\}_{t-\frac{\Delta t}{2}}
\]

\[
= \frac{1}{2} \{F^g\}_{t+\frac{\Delta t}{2}} + \frac{1}{2} \{F^g\}_{t-\frac{\Delta t}{2}} + \frac{1}{2} \{F^g\}_{t+\frac{\Delta t}{2}} + \frac{1}{2} \{F^g\}_{t-\frac{\Delta t}{2}}
\]  

(2.48)

The unknown terms in the equation above are the nodal temperature vector \( \{T^g\} \) and the vectors \( \{F^g\}_C \), \( \{F^g\}_R \) at time \( t+\Delta t/2 \). All the vectors and the matrices at the previous instant of time \( t-\Delta t/2 \) are known. Therefore the Eq. (2.48) can be written as
After substituting all the known quantities at time \((t - \Delta t/2)\) one obtains the above equation in the form (the vector \(\{A_t\}\) contains values at time \((t - \Delta t/2)\))

\[
\left( \left[ KC^q \right] + \frac{2}{\Delta t} \left[ CP^q \right] \right) \{ T^q \}_{t-\frac{\Delta t}{2}} - \left( \frac{2}{\Delta t} \left[ CP^q \right] - \left[ KC^q \right] \right) \{ T^q \}_{t+\frac{\Delta t}{2}} + \{ F^c \}_{t-\frac{\Delta t}{2}} + \{ F^r \}_{t-\frac{\Delta t}{2}} = \{ A_t \} 
\]

(2.49)

The vector \(\{A_t\}\) is known at this point. The vectors \(\{F^c\}_{t-\frac{\Delta t}{2}}\) and \(\{F^r\}_{t-\frac{\Delta t}{2}}\) are to be evaluated using the nodal temperatures at time \(t+\Delta t/2\). We can use an iteration procedure to solve the Eq. (2.50) for the nodal temperatures. In this procedure, we assume a nodal temperature vector \(\{T^q\}\) at time \(t+\Delta t/2\), which is same as the nodal temperature at time \(t-\Delta t/2\). This assumed value is substituted on the right hand side of the Eq. (2.50). Then the Eq.(2.50) reduces to

\[
\left( \left[ KC^q \right] + \frac{2}{\Delta t} \left[ CP^q \right] \right) \{ T^q \}_{t+\frac{\Delta t}{2}} = \{ A_t \} + \{ F^c \}_{t-\frac{\Delta t}{2}} + \{ F^r \}_{t-\frac{\Delta t}{2}} 
\]

(2.50)

where \(\{A_2\}\) is now known. Solution of Eq. (2.51) will give us the unknown nodal temperature vector \(\{T^q\}_{t+\frac{\Delta t}{2}}\). The resultant nodal temperatures are compared with the assumed nodal temperatures and if the convergence criterion is not met, then for the next iteration the calculated nodal temperatures become the assumed nodal temperatures. The iterations continue till we get the converged transient nodal temperatures of the turbine blade. The transient nodal
temperatures are found in the airfoil cross-section at different heights of the turbine blade. Once the transient temperature distribution is known, the transient temperature gradients along the x and y directions at different heights can be calculated by using the following equation (2.52) for a twenty-noded element:

\[
\{g^e\} = \begin{bmatrix}
\frac{\partial T}{\partial x} \\
\frac{\partial T}{\partial y}
\end{bmatrix} = \begin{bmatrix}
T_1 \\
T_2 \\
T_3 \\
\vdots \\
T_{20}
\end{bmatrix} 
\]

\[ (2.52) \]

2.4 ILLUSTRATION OF THEORY

The heat transfer process within a turbine blade was studied on a blade made of AISI 4140, which is a chromium alloy steel, and is used in the manufacturing of aircraft gas turbine blades. In order to carry out the finite element analysis of this turbine blade, the blade was divided into 35 curved, solid, C^0 continuity, quadratic, serendipity, twenty noded isoparametric elements as shown in Fig. 2.1. The convergence in the nodal temperatures was achieved by refining the time increment. A Fortran code was written in order to carry out the convergence studies. The convective heat transfer coefficient were assumed to vary with temperature of the blade. The values for these variations were obtained from Mukherjee (1978). All the material properties such as α, E, c, k, etc
of the blade in the present analysis were selected as a function of temperature and were obtained from (Cubberly, 1980). Table in APPENDIX D shows the values of these parameters at different temperatures.

2.5 TEMPERATURE GRADIENT DISTRIBUTION ACROSS THE AIRFOIL CROSS-SECTION AT DIFFERENT HEIGHTS OF THE TURBINE BLADE

Bahree (1987) has shown that the effect of radiative heat flux is quite significant at elevated temperatures as the radiative heat flux contains terms made up of higher powers of the nodal temperatures. Therefore all the results include the radiative terms.

Assuming the room temperature as the initial condition, one of the most important findings of this investigation was to observe the change in transient temperature and temperature gradients along the z-direction of the turbine blade modelled as a three-dimensional problem. Fig. 2.4 to Fig. 2.10 show the change in temperature variation in the nodes along z-axis at different times at different cross-sections of the turbine blade. From these graphs, it is clear that there is a change in temperature along the height of the blade. Fig. 2.11 to Fig. 2.15 show the temperature contours along different heights of the blade at time t = 150 sec. Fig. 2.16 to Fig. 2.20 show the variation of temperature gradients along the x and y directions at time t = 150 seconds as we go along the height of the blade. The careful study of these gradients is very important to understand the dynamics of the heat transfer process. An interesting observation can be made from these figures that the temperature gradient increases along the height of the blade which is in accordance with the direction of the heat flux, which is from the top to the bottom. This helps us to conclude that, in addition to the
thermal gradients along $x$ and $y$ axes of the blade, the thermal gradients along the height of the turbine blade are also very significant during the transient state and as such cannot be neglected in calculating the overall stress distribution in the blade.

2.6 TRANSIENT THERMAL STRESS DETERMINATION - THE MATHEMATICAL MODEL

The next step in the heat transfer analysis is the calculation of thermal stresses caused by the temperature variation across the cross-section and along the height of the turbine blade.

The elemental stress is calculated using Hooke's law. Using this law, the elemental stress and strain vectors are related as given by [Segerlind, 1976]

$$\{\sigma^*\} = [D_i^*] \{e^*\} - [D_i^*] \{e_o^*\}$$  \hspace{1cm} (2.53)

where

$\{e^*\}$ is the elemental strain vector

$\{e_o^*\}$ is the elemental initial strain vector

$\{\sigma^*\}$ is the elemental stress vector

$[D_i^*]$ is the material property matrix.

Stress components for thermal stresses analysis can be written as

$\{\sigma\}^T = [\sigma_{xx}, \sigma_{yy}, \sigma_{zz}, 0, 0, 0]$  

Similarly, the strain components are

$\{e\}^T = [e_{xx}, e_{yy}, e_{zz}, 0, 0, 0]$
Fig. 2.4 NODAL TEMPERATURE VARIATION ALONG THE Z-AXIS AT LEADING EDGE
Fig. 2.5  NODAL TEMPERATURE VARIATION ALONG THE Z-AXIS AT 15 % CHORD LENGTH
Fig. 2.6  NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS AT 30 % CHORD LENGTH
Fig. 2.7  NODAL TEMPERATURE VARIATION ALONG THE Z-AXIS AT 45% CHORD LENGTH
Fig. 2.8  NODAL TEMPERATURE VARIATION ALONG THE Z - AXIS AT 60 %

CHORD LENGTH
Fig. 2.9

NODAL TEMPERATURE VARIATION ALONG THE Z-AXIS AT 75% CHORD LENGTH

NODAL TEMPERATURE IN CENTIGRADE

CHORD LENGTH

NODES ALONG Z-AXIS

TIME = 230 SEC
TIME = 155 SEC
TIME = 95 SEC
TIME = 90 SEC

57
59
105
180
213
288
0
100
200
300
400
500
600
700
800
900
Fig. 2.10  NODAL TEMPERATURE VARIATION ALONG THE Z-AXIS AT TRAILING EDGE
Fig. 2.11  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION AT

\[ t = 150 \text{ SEC (HEIGHT OF THE BLADE} = 2.5 \text{ cm.)} \]
Fig. 2.12  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION AT

\( t = 150 \text{ SEC} \) (HEIGH OF THE BLADE = 5.0 cm.)
Fig. 2.13  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION AT

t = 150 SEC (HEIGHT OF THE BLADE = 7.5 cm.)
Fig. 2.14  TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION AT 

\[ t = 150 \text{ SEC} \] (HEIGHT OF THE BLADE = 10.0 cm.)
Fig. 2.15 TEMPERATURE CONTOURS ACROSS THE AIRFOIL SECTION AT

\[ t = 150 \text{ SEC} \text{ (HEIGHT OF THE BLADE} = 12.0 \text{ cm.)} \]
Fig. 2.16  TEMPERATURE GRADIENTS ACROSS THE BLADE CROSS-SECTION
AT t = 150 SEC. (HEIGHT OF THE BLADE = 2.5 cm.)
Fig. 2.17  TEMPERATURE GRADIENTS ACROSS THE BLADE CROSS-SECTION

AT t = 150 SEC.  (HEIGHT OF THE BLADE = 5.0 cm.)
Fig. 2.18  TEMPERATURE GRADIENTS ACROSS THE BLADE CROSS-SECTION

AT $t = 150$ SEC.  (HEIGHT OF THE BLADE = 7.5 cm.)
Fig. 2.19  TEMPERATURE GRADIENTS ACROSS THE BLADE CROSS-SECTION AT $t = 150$ SEC. (HEIGHT OF THE BLADE = 10.0 cm.)
Fig. 2.20  TEMPERATURE GRADIENTS ACROSS THE BLADE CROSS-SECTION

AT \( t = 150 \) SEC.  (HEIGHT OF THE BLADE = 12.0 cm.)
In order to evaluate thermal stress analysis, we have to know the matrix \([B_1,0]\) which relates the strain in the element to the nodal displacements. The matrix \([B_1,0]\) for the stress analysis can be evaluated in the following way:

1. The strain vector in the global coordinate system can be related to the derivative of the displacement field as

\[
\begin{pmatrix}
\varepsilon_{xx} \\
\varepsilon_{yy} \\
\varepsilon_{zz} \\
0 \\
0 \\
0
\end{pmatrix}
= [P]
\begin{pmatrix}
\frac{\partial u}{\partial x} \\
\frac{\partial v}{\partial x} \\
\frac{\partial w}{\partial x} \\
\frac{\partial u}{\partial y} \\
\frac{\partial v}{\partial y} \\
\frac{\partial w}{\partial y} \\
\frac{\partial u}{\partial z} \\
\frac{\partial v}{\partial z} \\
\frac{\partial w}{\partial z}
\end{pmatrix}
\]  

(2.54)

where the matrix \([P]\) is given as
2. The vector containing derivatives of the displacement field in the global coordinate system is mapped to the vector containing derivatives of the displacement field in the local coordinate system as

\[
\begin{cases}
\frac{\partial u}{\partial x} \\
\frac{\partial u}{\partial y} \\
\frac{\partial u}{\partial z} \\
\frac{\partial v}{\partial x} \\
\frac{\partial v}{\partial y} \\
\frac{\partial v}{\partial z} \\
\frac{\partial w}{\partial x} \\
\frac{\partial w}{\partial y} \\
\frac{\partial w}{\partial z}
\end{cases} = [Q] \begin{cases}
\frac{\partial u}{\partial \xi} \\
\frac{\partial u}{\partial \eta} \\
\frac{\partial u}{\partial \zeta} \\
\frac{\partial v}{\partial \xi} \\
\frac{\partial v}{\partial \eta} \\
\frac{\partial v}{\partial \zeta} \\
\frac{\partial w}{\partial \xi} \\
\frac{\partial w}{\partial \eta} \\
\frac{\partial w}{\partial \zeta}
\end{cases}
\]

(2.56)

The matrix \([Q]\) contains terms of the inverse of the Jacobian \([J]\) and can be written as
The Jacobian matrix is formed by the chain rule of differential calculus as below:

\[ \frac{\partial u}{\partial \xi} = \frac{\partial u}{\partial x} \frac{\partial x}{\partial \xi} + \frac{\partial u}{\partial y} \frac{\partial y}{\partial \xi} + \frac{\partial u}{\partial z} \frac{\partial z}{\partial \xi} \]  

Therefore, the Jacobian is given as follows:

\[ [J] = \begin{bmatrix} \frac{\partial x}{\partial \xi} & \frac{\partial y}{\partial \xi} & \frac{\partial z}{\partial \xi} \\ \frac{\partial x}{\partial \eta} & \frac{\partial y}{\partial \eta} & \frac{\partial z}{\partial \eta} \\ \frac{\partial x}{\partial \zeta} & \frac{\partial y}{\partial \zeta} & \frac{\partial z}{\partial \zeta} \end{bmatrix} \]  

where \( J_{11} \) is given as:

\[ J_{11} = \frac{\partial x}{\partial \xi} = \frac{\partial}{\partial \xi} \left( [N]^n \right) \left( x \right)^n \]  

3. The vector containing the derivatives of the displacement field in the local coordinate system can be related to the nodal displacement vector as
The matrix $[R]$ contains the derivative of the interpolation functions with respect to the local coordinate system and this matrix can be represented as

\[
\begin{bmatrix}
\frac{\partial}{\partial x_1} & \frac{\partial}{\partial x_2} & \cdots & \frac{\partial}{\partial x_n}
\end{bmatrix} = [R]
\]
Therefore, the $[B,^\circ]$ matrix for an element can be expressed as

$$[B,^\circ] = [P] \quad [Q] \quad [R]$$

$$[6 \times 60] \quad [6 \times 9] \quad [9 \times 9] \quad [9 \times 60] \quad (2.63)$$

The materials property matrix $[D,^\circ]$ for a three-dimensional isotropic material is given by
The initial strain vector for a thermal load is

\[ \varepsilon_0 = \alpha \Delta T_{sv} \]  

The turbine blade material is subjected to thermal expansion and the thermal force vector generated as a result of this expansion is written as follows:
Just like before (as in Eq 2.44) one would have to obtain the global stiffness matrix \([KS^g]\) and the global force vector \([F^g]\) (the equations will be linear here) and write the equations for any time \(t\) as

\[
[KS^g] \{\Delta g\} = \{F^g\}
\] (2.67)

The solution of this system will yield the displacements \(d_1, d_2, \ldots\) etc. The stresses within any element will then be obtained using Eqs. (2.53), (2.54), (2.56) and (2.61). The modulus of elasticity \(E\) in Eq. (2.64), and the coefficient of thermal expansion \(\alpha\) in Eq. (2.65) are evaluated at the average temperature of the element. The \(T_{av}\) is the average temperature of the twenty nodes of an element. \(\Delta T_s\) is the difference in \(T_{av}\) at any instant of time and \(T_{av}\) initially when there is a stress free state. The elemental temperatures obtained from the analysis before thus help us to calculate the thermal stress \(\sigma_{xx}, \sigma_{yy},\) and \(\sigma_{zz}\) using Eq. (2.53). These stresses at different instants of time are shown in Figs. 2.21 to 2.23. It is clear from these figures that stresses along \(x\) and \(y\) directions are also significant and thus must be included in the design analysis of the turbine blade. It was also observed that with the present heating path, the maximum stress curve for \(\sigma_{zz}\) would
exceed unity as shown in Fig. 2.23 and as such would be an infeasible heating path. In order that maximum stress curve does not exceed the value of unity, the heating path has to be modified so that the maximum stresses remain below the yield stress throughout the heating period (Fig. 2.24). These stresses have been normalized with respect to the yield stress, \( \sigma_{\text{yield}} \). This yield stress is a function of temperature and therefore the normalizing values would be different at different instants of time. From the stress-time graph, one can clearly observe that the peak stress occurs at time \( t = 200 \) seconds. The stresses in these figures increase to a maximum value and then decrease because different points in the blade undergo differential rise in temperature to a maximum value and then the entire blade is expected to reach a steady state temperature given sufficient time. During the steady state, the thermal stresses would be zero because \( \Delta T \) from the reference state at two instants of time in the steady state will be the same for any element. These results show that \( \sigma_{xx} \) and \( \sigma_{yy} \) are not negligible as compared to \( \sigma_{zz} \). This is due to the three-dimensional variation of the temperatures.
Fig. 2.21 TRANSIENT THERMAL STRESS ($\sigma_{x_1}$) DISTRIBUTION IN THE BLADE
Fig. 2.22 TRANSIENT THERMAL STRESS ($\sigma_y$) DISTRIBUTION IN THE BLADE
Fig. 2.23  TRANSIENT THERMAL STRESS ($\sigma_{zz}$) DISTRIBUTION IN THE BLADE
Fig. 2.24 TRANSIENT THERMAL STRESS ($\sigma_{zz}$) DISTRIBUTION IN THE BLADE FOR THE FEASIBLE HEATING RATE OF 24°C/SEC AT DIFFERENT INSTANTS OF TIME
2.7 CONCLUSION

The main objective of the investigation done in this chapter was to determine the transient temperature distribution, temperature gradient distributions and the thermal stresses in a turbine blade modelled as a three-dimensional non-linear finite element problem. The observation made from the study done in this chapter clearly shows that a significant temperature gradients exist along the height of the blade which are very essential in the performance of an accurate analysis of the temperature and thermal stress distribution in a turbine blade.
CHAPTER 3
FREE VIBRATION ANALYSIS

3.1 INTRODUCTION

The free vibration analysis of the turbine blade has been a challenging area to the engineers for a long time. Before designing a turbine blade, it is very essential to know its undamped natural frequencies and there are various reasons for it. Right from the start and until it comes to a constant operating speed, the rotor goes through the various blade natural frequencies. In order to avoid resonance, the steady state operating speed should not match with the system natural frequencies. Also, as the temperature of the turbine blade changes with time, there is a change in the material properties of the turbine blade. This change in the material property of the turbine blade is reflected in the change in the conventional stiffness $[K_c]$ matrix of the blade and this causes a change in the natural frequencies of the turbine blade. There is also a significant change in the natural frequencies because of the variational effect of stress stiffness matrix $[K_s]$, which is non-linear in nature and is caused by the rotation of the turbine blade.

The present investigation is to study the combined effects of the two non-linearities on the natural frequencies of the turbine blade. The mathematical model for the study of the vibration analysis of the turbine blade is formulated using curved, three-dimensional, $C^0$ continuity, serendipity, twenty-noded isoparametric finite elements. This type of element is chosen because of its versatility in accurately mapping the complex geometry of the turbine blade. The blade is basically an airfoil cross-section, being asymmetric,
pre-twisted and having taper along its length. The two sections of the blade along its length are shown in Fig. 3.1.

3.2 MATHEMATICAL FORMULATION

The finite element analysis of the blade was done by dividing it into 35 elements, as shown in Fig 2.1. These elements are curved, solid, $C^0$ continuity, serendipity, twenty-noded isoparametric finite elements. Such type of an element was chosen because it gives a non-linear displacement field, which in turn takes care of the non-linear geometry of the turbine blade. The element chosen was a $C^0$ continuity element because the degrees of freedom at the nodes of the element are translational in nature along all the three axes. Hence, by selecting the degrees of freedom for displacements at the nodes along all the three axes, one can get the other modes of vibration also. Also, in case of isoparametric elements, the generalised coordinates and the generalised displacements are related to the nodal coordinates and nodal displacements respectively by the same shape function. The striking feature of the isoparametric elements is that they make use of a special coordinate system ($\xi, \eta, \zeta$) as shown in Fig. 2.3. The shape functions are defined for the variation of the local coordinate system from +1 to -1. Therefore, we can have: $\xi = \pm 1$; $\eta = \pm 1$; and $\zeta = \pm 1$. The shape functions $[N_j]$ for the 20-noded isoparametric element can be expressed as (Zienkiewicz, 1971):
Fig 3.1  TURBINE BLADE AIRFOIL CROSS-SECTION AT THE ROOT AND THE TIP [Bahree, 1987]
Corner Nodes

\[ N_i = \frac{1}{8} (1 + \xi \xi_i) (1 + \eta \eta_i) (1 + \zeta \zeta_i) (\xi \xi_i + \eta \eta_i + \zeta \zeta_i - 2) \]  \hspace{1cm} (3.1)

The details of Eq. (3.1) are given in APPENDIX A.

Typical Mid-side nodes

\[ N_i = \frac{1}{4} \left( 1 - \xi_i^2 \right) (1 + \eta \eta_i) (1 + \zeta \zeta_i) \]  \hspace{1cm} (3.2)

for the case when \( \xi_i = 0, \eta_i = \pm 1, \) and \( \zeta_i = \pm 1. \) The subscript \( i \) stands for the \( i^{th} \) node of the twenty-noded element in Eqs. (3.1) and (3.2).

Similarly one can write the equations corresponding to Eq. (3.2) as shown in APPENDIX B.

The mathematical formulation is arrived in the following way:

1. The local displacement vector \( \{u, v, w\} \) of a point in the element is related to nodal displacement vector \( \{d\} \) through matrix \( [N] \):

\[ \begin{bmatrix} u \\ v \\ w \end{bmatrix} = [N] \begin{bmatrix} d \end{bmatrix} \]  \hspace{1cm} (3.3)

or
2. The global coordinate vector \( \{ x, y, z \} \) at the same point in the element is related to nodal local coordinates \( \{ c \} \) through \( [N] \) as

\[
\begin{bmatrix}
  x \\
  y \\
  z
\end{bmatrix} = [N] \begin{bmatrix}
  c
\end{bmatrix}
\]

(3.5)
\[
\begin{pmatrix}
  x \\
  y \\
  z
\end{pmatrix} =
\begin{bmatrix}
  N_1 & 0 & 0 & N_2 & 0 & 0 & N_3 & 0 & 0 & \ldots & 0 & 0 & N_{20} & 0 & 0 \\
  0 & N_1 & 0 & 0 & N_2 & 0 & 0 & N_3 & 0 & \ldots & 0 & 0 & N_{20} & 0 \\
  0 & 0 & N_1 & 0 & 0 & N_2 & 0 & 0 & N_3 & \ldots & 0 & 0 & N_{20} & 0 \\
\end{bmatrix}
\begin{pmatrix}
  x_1 \\
  y_1 \\
  z_1 \\
  x_2 \\
  y_2 \\
  z_2 \\
  \vdots \\
  x_{20} \\
  y_{20} \\
  z_{20}
\end{pmatrix}
\] (3.6)

where

- \(u, v, w\) are generalized displacements along global X, Y, and Z axes,
- \(x, y, z\) are generalized coordinates along global X, Y, and Z axes,
- \(\{d\}\) is nodal displacement vector,
- \(\{c\}\) is nodal coordinate vector,
- \([N]\) and \([\tilde{N}]\) are shape function matrices which are the same for isoparametric elements, and both \([N]\) and \([\tilde{N}]\) are functions in terms of \(\xi, \eta, \) and \(\zeta\).

The eigenvalues and eigenvectors of the system are obtained from the stiffness and the mass matrices which are described below.
3.2.1 \([K^e_s]\) ELEMENTAL CONVENTIONAL STIFFNESS MATRIX - THE MATHEMATICAL FORMULATION

This stiffness matrix which depends upon the elastic properties of the turbine blade is given by the following relationship:

\[
[K^e_s] = \int_v [B^e_s]^T [D_e] [B^e_s] \, dv
\]  

(3.7)

In order to formulate \([K^e_s]\), we have to know the matrix \([B^e_s]\) which relates the strain in the element to the nodal displacements and the material property matrix \([D_e]\). The matrix \([B^e_s]\) for the stress analysis can be evaluated in the same way as explained in the Eq. (2.54) to Eq. (2.63).

The materials property matrix \([D_e]\) for a three-dimensional isotropic material is same as given by Eq. (2.64).

Thus, elemental stiffness matrix can be expressed as

\[
[K^e_s] = \int_v [B^e_s]^T [D_e] [B^e_s] \, dv = \int_v [B^e_s]^T [D_e] [B^e_s] \, dx \, dy \, dz
\]  

(3.8)

The \([K^e_s]\) matrix in terms of \(\xi\), \(\eta\), and \(\zeta\) can be expressed as

\[
[K^e_s] = \int_v [B^e_s]^T [D_e] [B^e_s] \, |J| \, d\xi \, d\eta \, d\zeta
\]  

(3.9)

where \(|J|\) is the determinant of Jacobian matrix \([J]\).

Since the local coordinates \(\xi\), \(\eta\), and \(\zeta\) vary from -1 to +1, for the isoparametric formulation, the Eq. (3.9) can also be expressed as
The matrix $[K^e]$ is evaluated numerically by making use of Gaussian quadrature. Because of the quadratic polynomial used for each side of the twenty-noded element, the expression for the elemental matrix is also quadratic. Therefore, for a general second degree polynomial $\Phi = a + b\xi + c\xi^2$, two Gauss points will give an accurate solution. This gives rise to 8 Gauss points within each element which is adequate to integrate each element of the matrix $[K^e]$.

Therefore,

$$
[K^e] = \sum_{i=1}^{2} \sum_{j=1}^{2} \sum_{k=1}^{2} W_i W_j W_k [B, (\xi, \eta, \zeta)]^T [D] [B, (\xi, \eta, \zeta)] \mu(\xi, \eta, \zeta)
$$

(3.11)

where $W_i$, $W_j$, and $W_k$ are corresponding weighting functions in the $\xi$, $\eta$, and $\zeta$ directions respectively.

3.2.2 $[K^e]$ ELEMENTAL STRESS STIFFNESS MATRIX - THE MATHEMATICAL FORMULATION

We define another elemental matrix $[K^0_o]$ that accounts for the change in potential energy associated with rotation of volume elements under load. This matrix is called the stress stiffness matrix. It is independent of material properties and depends only on the element's geometry, the displacement field, and state of stress. APPENDIX E explains how this matrix comes into the overall picture and why it is important to include this matrix in the present analysis.

The most important thing in deriving the mathematical formulation for elemental
Stress Stiffness Matrix \([K^e]\), for a complicated geometry (as in our case), is to make sure that both linear and non-linear terms have been included. Thus, a general expression for Elemental Stress Stiffness Matrix \([K^e]\) is derived as follows:

Strains can be written as

\[
\{ \varepsilon \} = \{ \varepsilon_L \} + \{ \varepsilon_{NL} \}
\]  

(3.12)

where \(\{ \varepsilon_L \}\) is the strain linear in the displacement derivatives and the higher order terms give rise to \(\{ \varepsilon_{NL} \}\), the non-linear strains. The strain energy \(U\) is stored as a result of constant stresses \(\{ \sigma_o \}\) acting through strains \(\{ \varepsilon \}\) and is given as:

\[
\int \{ \varepsilon \}^T \{ \sigma_o \} dV = \int \{ \varepsilon_L \}^T \{ \sigma_o \} dV + \int \{ \varepsilon_{NL} \}^T \{ \sigma_o \} dV
\]  

(3.13)

or

\[
U = U_L + U_{NL}
\]

APPENDIX E explains that \([K_o]\) is produced by stresses acting through displacements associated with higher-order contributions to strain.

Therefore the \([K_o]\) is extracted from \(U_{NL}\) as follows:

Expressions for strain are written in conventional notation as

\[
\varepsilon_x = u_x + \frac{1}{2}(u_{x,x}^2 + v_{x,x}^2 + w_{x,x}^2)
\]  

(3.14)

and

\[
\gamma_{xy} = u_y + v_x + (u_{x,y} + v_{x,y} + w_{x,y})
\]  

(3.15)
The terms within the parenthesis give rise to $\{\varepsilon_{\text{NL}}\}$ where $x$, $y$, $z$ are global directions.

In order to formulate $[K_e]$, we define the following:

$\{\delta\} = [G][d]$ where

$\{d\}$ is the nodal displacement vector as described before. We can also write

$\{\delta\} = \{u_x, u_y, u_z, v_x, v_y, v_z, w_x, w_y, w_z\}$

where $u_x$, $u_y$, and $u_z$ are $\partial u/\partial x$, $\partial u/\partial y$, and $\partial u/\partial z$ respectively.

Differentiating shape functions in $[N]$ give us the coefficients in $[G]$ given by Eq. (2.62).

We define one more matrix here which is

$$ [Q_1] = \begin{bmatrix}
    u_x & 0 & 0 & v_x & 0 & 0 & w_x & 0 & 0 \\
    0 & u_y & 0 & 0 & v_y & 0 & 0 & w_y & 0 \\
    0 & 0 & u_z & 0 & 0 & v_z & 0 & 0 & w_z \\
    u_y & 0 & v_y & 0 & v_x & 0 & w_y & 0 & w_z \\
    0 & u_z & 0 & v_z & 0 & v_y & 0 & w_z & w_y \\
    u_z & 0 & u_x & 0 & v_x & 0 & w_z & 0 & w_x \\
  \end{bmatrix} $$

(3.16)

Therefore Eq. (3.13) can be written as

$$ \{ \varepsilon \} = \{ \varepsilon_L \} + \{ \varepsilon_{\text{NL}} \} = \{ \varepsilon_L \} + \frac{1}{2} [Q_1] [G] \{d\} $$

(3.17)

Displacements $u$, $v$, and $w$ are functions of $x$, $y$, $z$ where $x$, $y$, and $z$ is the position of a point in the continuum in the unstrained configuration. Therefore $\{\varepsilon\}$ is called Lagrangian strain and $\{\varepsilon\} = 0$ for any rigid-body motion, no matter how large is the rotation or the displacement (Cook, 1981).

The vector $\{\sigma_0\}$ is defined as
Combining Eqs. (3.13), (3.17) and (3.18), we get

\[
U_{NL} = \frac{1}{2} \int_\Omega \left[ \begin{array}{c} a^T \end{array} \right]^T \left[ \begin{array}{c} Q_1 \end{array} \right] \left\{ \sigma_0 \right\} dv = \frac{1}{2} \int_\Omega \left[ \begin{array}{c} a^T \end{array} \right]^T \left[ \begin{array}{c} G \end{array} \right] \left\{ \delta \right\} dv
\]

(3.19)

where

\[
\left[ Q \right]^T \left\{ \sigma_0 \right\} = \left[ \begin{array}{ccc} [s] & 0 & 0 \\ 0 & [s] & 0 \\ 0 & 0 & [s] \end{array} \right] \left\{ \delta \right\}
\]

(3.20)

\[
\begin{bmatrix} \sigma_{x0} & \tau_{xy0} & \tau_{xzo} \\ \tau_{xy0} & \sigma_{y0} & \tau_{yzo} \\ \tau_{xzo} & \tau_{yzo} & \sigma_{z0} \end{bmatrix}
\]

(3.21)

Thus elemental stress stiffness matrix is given as

\[
\left[ K_0 \right] = \int_\Omega \left[ G \right]^T \left[ \begin{array}{ccc} [s] & 0 & 0 \\ 0 & [s] & 0 \\ 0 & 0 & [s] \end{array} \right] \left[ G \right] dv
\]

(3.22)

The element stress stiffness matrix \( [K_0] \) is a symmetric matrix like the conventional stress
stiffness matrix \([K_a]\) and the global \([K_a]\) is obtained by the usual assembly of elemental \([K_a^e]\) matrices.

### 3.2.3 \([M^e]\) ELEMENTAL MASS MATRIX - THE MATHEMATICAL FORMULATION

The elemental mass matrix for the system is given by:

\[
[M^e] = \int \left[ N \right]^T \rho \left[ N \right] d\nu \tag{3.23}
\]

where \([N]\) is the shape function matrix,

\(\rho\) is the material density.

In terms of local coordinate system \(\xi, \eta,\) and \(\zeta,\) the Eq. (3.34) can be represented as

\[
[M^e] = \int_{-1}^{1} \int_{-1}^{1} \int_{-1}^{1} \rho \left[ N \right]^T \left[ N \right] d\xi d\eta d\zeta \tag{3.24}
\]

Using Gauss quadrature, the elemental expression for \([M^e]\) can be written as

\[
[M^e] = \sum_{i=1}^{2} \sum_{j=1}^{2} \sum_{k=1}^{2} W_i W_j W_k \left[ N(\xi, \eta, \zeta) \right]^T \left[ N(\xi, \eta, \zeta) \right] J(\xi, \eta, \zeta) \tag{3.25}
\]

### 3.3 FREE VIBRATION ANALYSIS

The global stiffness matrix \([K^G]\) obtained from assembling global \([K_a^G]\) and global \([K_o^G]\), and global mass matrix \([M^G]\) were evaluated after assembling the elemental stiffness and mass matrices respectively. The natural frequencies were found by solving the following equation for the eigenvalue problem:

\[
[K^G][x] - \lambda[M^G][x] = 0 \tag{3.26}
\]
Figs. 3.4 to 3.10 show the variation of first five natural frequencies of the turbine blade both without and with the inclusion of stress stiffness matrix $[K_n]$ to the conventional matrix $[K_a]$ at different rotor speeds. It is clear from these graphs that there is a significant change in the frequency of the turbine blade by including stress stiffness matrix. Because of the change in the frequency there is also a change in the actual resonance of the system. Fig. 3.11 and 3.12 show the projected and actual resonance for first and second natural frequencies. Frequencies beyond second are of only academic interest because the maximum speed at which the rotor operates does not go beyond 4000 rpm.

### 3.4 EFFECT OF TEMPERATURE VARIATION ON THE NATURAL FREQUENCIES OF THE BLADE

One of the objectives of the present investigation was to show the effect of various factors on the natural frequencies of the turbine blade. Besides the effect of stress stiffness matrix, it was observed that the frequencies of the turbine blade also changed with the change in the temperature. This is due to the change in the elastic properties of the blade which resulted a change in the stiffness matrix of the system. The heating path chosen for finding the effect of change in temperature on frequency was the one which put the yield stress below one as shown in Fig. (2.24). Figs. 3.13 to 3.17 show the variation of first five natural frequencies of the turbine blade at different instants of time during the heating process. It is very clear from these figures that as the material is heated, the frequencies start decreasing due to a decrease in the value of the modulus of elasticity of the material of the turbine blade. If the effect of stress stiffness matrix is
not included, then the frequencies decrease only with increase in temperature of the turbine blade. The frequencies increase due to large deflections and, at the same time, these decrease due to increased temperature due to lowering of $E$, the modulus of elasticity. Thus, it is clear from these figures that there is a significant change in the natural frequencies in the transient state and is thus an important factor to be taken into account while designing a turbo blade. Therefore, the frequency analysis of the turbine blade must include the combined effect of

i) inclusion of the stress stiffness matrix to the conventional matrix and

ii) change in the material properties of the blade resulting in change in overall stiffness matrix of the turbine blade due to change in temperature in the transient phase.
Fig. 3.2  VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 300 RPM
Fig. 3.3  VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 900 RPM
Fig. 3.4 VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 1500 RPM
Fig. 3.5  VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 2100 RPM
Fig. 3.6  VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 2700 RPM
Fig. 3.7  VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 3300 RPM
Fig. 3.8  VARIATION OF FIRST FIVE NATURAL FREQUENCIES OF TURBINE BLADE AT ROTOR SPEED OF 3900 RPM
Fig. 3.9  COMPARISON OF FIRST ACTUAL CRITICAL SPEED WITH FIRST PROJECTED CRITICAL SPEED USING STRESS STIFFNESS MATRIX
Fig. 3.10  COMPARISON OF SECOND ACTUAL CRITICAL SPEED WITH SECOND PROJECTED CRITICAL SPEED USING STRESS STIFFNESS MATRIX

A2  CRITICAL SPEED = 1994.911 (IF S.S.M = 0)

B2  ACTUAL CRITICAL SPEED = 3042.239
CONVENTIONAL STIFFNESS (S.S.M. = 0, NO CHANGE IN TEMPERATURE)
- CONVENTIONAL STIFFNESS + CHANGE IN TEMPERATURE
- CONVENTIONAL STIFFNESS + STRESS STIFFNESS
- CONVENTIONAL STIFFNESS + STRESS STIFFNESS + CHANGE IN TEMPERATURE

Fig. 3.11 COMPARISON OF FIRST CRITICAL SPEED OBTAINED FROM VARIOUS METHODS
Fig. 3.12   COMPARISON OF SECOND CRITICAL SPEED OBTAINED FROM VARIOUS METHODS
Fig. 3.13  COMPARISON OF THIRD CRITICAL SPEED OBTAINED FROM VARIOUS METHODS
Fig. 3.14  COMPARISON OF FOURTH CRITICAL SPEED OBTAINED FROM VARIOUS METHODS
Fig. 3.15  COMPARISON OF FIFTH CRITICAL SPEED OBTAINED FROM VARIOUS METHODS
3.5 CONCLUSIONS

In this chapter, the eigenvalues and eigenvectors for the blade were found by discretizing of the turbine blade into finite elements. The turbine blade had a height of 12 cm. and was discretized into 35 elements with 7 elements across the cross-section and 5 layers along the height. The dynamics of the system was represented by 308 nodes and 924 degrees of freedom. The natural frequencies of the turbine blade were calculated after taking into account the effect of including stress stiffness matrix to the conventional stiffness matrix and the effect of change in the material properties of the turbine blade with the change in temperature and it was observed that inclusion of these two factors resulted in a significant change in the natural frequency of the turbine blade.
CHAPTER 4
TRANSIENT STRESS ANALYSIS
& FATIGUE LIFE ESTIMATION

4.1 INTRODUCTION

The present chapter deals with the transient stress analysis due to thermal, vibratory, and centrifugal loading on the blade. Primarily, there are three different reasons for the existence of the transient conditions of operation. They are due to: (a) unsteady flow conditions on the blade arising out of gas admission to the stator, (b) rising gas temperature till it reaches to steady-state, and (c) acceleration and deceleration of the rotor during starting and shutting operations of the machine. Under the transient conditions of gas admission, the blade experiences variable magnitudes of excitation forces. The resulting stresses may be high and can lower the life cycle of the blade. These stresses, however, can be minimized by good operating procedures. The present investigation studies the separate and collective effect of transient conditions arising out of thermal, vibratory, and centrifugal stress conditions. The transient conditions of operation that exist during the machine start-up and shut-down conditions generate resonant stresses at certain rotor speeds. The forces due to gas pressure are assumed to be distributed along the entire length of the blade. The actual magnitude of the pressure distribution would depend upon the type and operating conditions of the turbine. As has been seen in the Chapter 3 that the overall stiffness matrix \([K]\) changes due to a change in the stress stiffness matrix \([K_o]\) which changes as a result of change in the
deflection of the turbine blade. In order to economize on the CPU time, the stress stiffness matrix \([K_s]\) is computed as the average value of \([K_s]\) at 12 different speeds near the critical speed zone of the turbine blade. In addition, the value of \(E\) also changes due to the change in the temperature of the turbine blade. This value of \(E\) has also been kept constant as a first approximation towards saving on the CPU time for calculating the vibratory stresses (Bahre, 1987). The boundary conditions at the root of the blade correspond to zero deflection for the applicable nodes. This averaging approximates the linearization of the system.

### 4.2 TRANSIENT RESPONSE OF A ROTOR BLADE DUE TO NOZZLE EXCITATION FORCES

The equations of motion for a damped system can be expressed as

\[
[M^o] \ddot{x} + [C^o] \dot{x} + [K^o] x = \{f(t)^o\}
\]  

(4.1)

The response of each mode is found by decoupling the global equation of motion given in Eq. (4.1). The procedure for decoupling the equations of motion is given below. Introducing a state vector

\[
\{y\} = \begin{bmatrix} \{\dot{x}\}_s \\ \{x\}_s \end{bmatrix}
\]  

(4.2)

Eq. (4.1) can be reduced to a set of simultaneous first-order equations which are expressed as:
\[ [AA][\dot{y}] + [BB][y] = \{EE\} \] (4.3)

where

\[
[AA] = 
\begin{bmatrix}
[0] & [M^0] \\
(n \times n) & (n \times n)
\end{bmatrix}
\begin{bmatrix}
[M^0] \\
(n \times n)
\end{bmatrix}
\]

(4.4)

\[
[BB] = 
\begin{bmatrix}
[-M^0] & [0] \\
(n \times n) & (n \times n)
\end{bmatrix}
\begin{bmatrix}
[0] \\
(n \times n)
\end{bmatrix}
\]

(4.5)

and

\[
\{EE\} = 
\begin{bmatrix}
\{0\} \\
(n \times 1)
\end{bmatrix}
\begin{bmatrix}
\{F^0\} \\
(n \times 1)
\end{bmatrix}
\]

(4.6)

The damped natural frequencies are then found by finding the eigenvalues of the dynamical matrix \([H]\) which can be expressed as

\[
[H] = -[AA]^{-1}[BB]
\]

(4.7)

The dynamical matrix of the transposed system will be
Eq. (4.3) can be transformed using the relation

\[ \{ y \} = \{ \phi \} \{ z \} \]  

(4.9)

where \( \{ \Phi \} \) is the modal matrix of the system defined in Eq. (4.3).

Substituting Eq. (4.9) into Eq. (4.3) and premultiplying by \( \{ \Phi^* \}^T \), we obtain


(4.10)

This leads to the diagonalization of the global mass and stiffness matrices respectively. Thus the entire system is expressed by individual single degree of freedom systems by decoupling the equations of motion.

In case of mass-orthonormalization of the matrix of eigenvectors, the Eq. (4.1) can be written as

\[ \{ \ddot{q} \} + 2 \zeta \omega_n \{ \dot{q} \} + \omega_n^2 \{ q \} = \{ GF \} \]  

(4.11)

For individual modes, Eq. (4.11) can be expressed as

\[ \ddot{q}_k(t) + 2 \zeta \omega_k \dot{q}_k(t) + \omega_k^2 \ddot{q}_k(t) = QF_k(t) \]  

(4.12)

where \( k \) represents the mode number.

The present investigation models the nozzle excitation force as a sinusoidal pulse as a superposition of two sine waves with a delay of time \( t_i \). Consequently, there will be two types of responses; one for time \( t < t_i \) (due to first wave) and second for time \( t > t_i \) (due to both waves), as shown in Fig. (4.1). Now, we will consider a rotor blade excited by a force \( F(\omega,t) \). Since the frequency
Fig. 4.1 REPRESENTATION OF THE NOZZLE EXCITATION FORCES AS A
SINUSOIDAL PULSE
is time dependent, we have

\[ \omega = \omega_c + at \]  \hfill (4.13)

where

\( \omega_c \) = angular velocity

\( \alpha \) = angular acceleration, and

\( \omega \) = instantaneous angular velocity

The excitation force for sinusoidal pulse is

\[ F = F_0 \sin(\omega t) \]  \hfill (4.14)

The response of each mode can be calculated as follows:

\[ \ddot{x} + \omega_n^2 x = \frac{F_0}{m} \sin(\omega t) \]  \hfill (4.15)

Since \( \omega = \frac{\pi}{t_i} \), we have

\[ \ddot{x} + \omega_n^2 x = \frac{F_0}{m} \sin \left( \frac{\pi}{t_i} t \right) \]  \hfill (4.16)

The general solution for Eq. (4.16) is

\[ x(t) = A \sin(\omega_n t) + B \cos(\omega_n t) + \frac{F_0}{m} \sin \left( \frac{\pi}{t_i} t \right) \frac{\left( \omega_n^2 - \left( \frac{\pi}{t_i} \right)^2 \right)^{1/2}}{\left( \omega_n^2 - \left( \frac{\pi}{t_i} \right)^2 \right)^{1/2}} \]  \hfill (4.17)
The initial conditions are

For \( x(0) = \dot{x}(0) = 0 \), we get

\[
B = 0 \text{ and } A = \frac{-F_o}{m} \frac{\pi}{l_1} \left( \frac{\pi}{\omega_n} - \left( \frac{\pi}{l_1} \right)^2 \right) \tag{4.18}
\]

Since \( \omega_n = \frac{n \pi}{l_1} \), we can write

\[
\frac{\pi}{l_1} \omega_n = \frac{\pi \tau}{2n} = \frac{\tau}{2 l_1} \tag{4.19}
\]

Therefore Eq. (4.17) becomes

\[
x(t) = \frac{F_o}{m} \left[ \frac{-\pi \sin (\omega_n t)}{\omega_n \left( \omega_n^2 - \left( \frac{\pi}{l_1} \right)^2 \right)} + \frac{\sin \left( \frac{n t}{l_1} \right)}{\omega_n \left( \omega_n^2 - \left( \frac{n}{l_1} \right)^2 \right)} \right] \tag{4.20}
\]

or

\[
x(t) = \frac{F_o}{m} \left[ \frac{-\pi \sin \left( \frac{2\pi t}{\tau} \right)}{\left( \frac{2\pi}{\tau} \right)^2 - \left( \frac{\pi}{l_1} \right)^2} - \frac{n t}{\tau} \sin \left( \frac{2\pi t}{\tau} \right) + \sin \left( \frac{n t}{l_1} \right) \right] \tag{4.21}
\]

The Eq. (4.21) is the response for the time \( t < t_1 \).

For time \( t > t_1 \), the solution above is added with \( t \) replaced by \( (t - t_1) \). The response in
that case will be as:

\[
x(t) = \frac{F_0}{m} \left\{ \frac{2\pi}{T} \sin \left( 2\pi \frac{t-t_1}{T} \right) - \frac{\pi}{t_1} \sin \left( \frac{\pi}{t_1} \right) \right\}
\]

or

\[
x(t) = \frac{F_0}{m} \left\{ \frac{T}{2t_1} \sin \left( \omega_n (t-t_1) \right) + \sin(\omega(t-t_1)) \right\}
\]

The gas forces have been calculated from the circulation and the lift over the airfoil.

The circulation over the airfoil is given by

\[
\Gamma = c \int_{\theta_0}^{\pi} \gamma(\theta) \sin \theta d\theta
\]

\[
= 2V_\infty c \left\{ A_0 (1 + \cos \theta) + \sum A_i \sin n \theta \sin \theta \right\} d\theta
\]

\[
= 2V_\infty c \left\{ \pi A_0 + \frac{1}{2} \pi A_1 \right\}
\]

(4.24)
where
\[ V_\infty = 718.05 \text{ m/sec}. \]
\[ c = 0.07 \]
\[ A_0 = 0.40151 \]
\[ A_1 = 0.08146 \]

\( A_0 \) and \( A_1 \) are Glauert constants for a thin cambered airfoil.

The lift \( L \) is thus calculated as

\[ L = \pi \rho V_\infty^2 (2A_0 + A_1) \]  \hspace{1cm} (4.25)

where \( \rho = 2.89 \). \( F_o \) was 10\% of the value of the lift obtained in Eq. (4.25).

4.3 TRANSIENT RESPONSE DUE TO CENTRIFUGAL FORCE

The radial location of the element with reference to the axis of rotation of the rotor shaft causes the centrifugal forces in the turbine blade. The centrifugal force is given as:

\[ F_c = m \omega^2 (R + z) \]  \hspace{1cm} (4.26)

where

\( m \) = the mass of the element

\( R \) = radius of the rotor disk

\( z \) = distance from the root to the centre of gravity of element.

This force acts at the centre of gravity and such can be replaced by eight equivalent forces each of which act at the corner and mid nodes. Also, these centrifugal forces are time variant because \( \omega \) in Eq. (4.26) is a variable which is calculated from Eq. (4.14).
For calculating the dynamic response due to the centrifugal forces, the forcing function corresponding to the Eq. (4.19) is given as

\[ F_c = (R + z) (\alpha t)^2 m \]  \hspace{1cm} (4.27)

by assuming \( \omega_0 = 0 \) in Eq. (4.13).

The modal response is given by:

\[ q_k(t) = \frac{1}{\omega_{dk}} \int_0^1 (R + z) (\alpha t)^2 m e^{\xi_k \omega_{dk} (t - \tau)} \sin(\omega_{dk} (t - \tau)) d\tau \]  \hspace{1cm} (4.28)

where \( \omega_{nk} \) = natural frequency of \( k^{th} \) mode

\( \omega_{dk} \) = damped natural frequency of \( k^{th} \) mode

The solution for this equation can be written as (Weast, 1975):

\[ q_k(t) = \frac{(R + z)}{\omega_{dk}} \alpha^2 \sin(\omega_{dk} t) e^{\xi_k \omega_{dk} t} \]

\[ \left[ e^{\xi_k \omega_{nk} t} \sum_{r=0}^{2} \frac{(-1)^r 2!(\tau)^{(2-r)}}{\left( \frac{\xi_k^2 \omega_{nk}^2 + \omega_{dk}^2 r+1}{2} (2 - r)! \right)^{r+1}} \cos \left( \omega_{dk} \tau - (r + 1) \tan^{-1} \left( \frac{\omega_{dk}}{\xi_k \omega_{nk}} \right) \right) \right] \]

\[ \left. - \frac{(R + z)}{\omega_{dk}} \alpha^2 \cos(\omega_{dk} t) e^{\xi_k \omega_{nk} t} \right) \]  \hspace{1cm} (4.29)

\[ \left[ e^{\xi_k \omega_{nk} t} \sum_{r=0}^{2} \frac{(-1)^r 2!(\tau)^{(2-r)}}{\left( \frac{\xi_k^2 \omega_{nk}^2 + \omega_{dk}^2 r+1}{2} (2 - r)! \right)^{r+1}} \sin \left( \omega_{dk} \tau - (r + 1) \tan^{-1} \left( \frac{\omega_{dk}}{\xi_k \omega_{nk}} \right) \right) \right] \]
To calculate the response one has to decouple the equations of motion using the matrix of eigenvectors in the same fashion as shown from Eq. (4.1) to Eq. (4.12).

4.4 VIBRATORY STRESSES

The dynamic loading on the turbine blade is caused by nozzles placed at regular intervals along the periphery of the rotor. For the purpose of calculating the vibratory stresses, the turbine blade is divided into 35 twenty-noded elements shown in Fig. (2.4). Defining \( \{ \gamma \} \) and \( \{ x \} \) as elemental strains and displacements, from the total response of the turbine blade due to nozzle excitations, the strains in each element are obtained from the isoparametric finite element formulation as

\[
\{ \gamma \} = [ B ] \{ x \}
\]

\[
6 \times 1 \quad 6 \times 60 \quad 60 \times 1
\]

and the dynamic stresses are calculated as

\[
\{ \sigma \} = [ D ] \{ \gamma \}
\]

\[
6 \times 1 \quad 6 \times 6 \quad 6 \times 1
\]

The vibratory stresses were calculated using Eqs. (4.30) and (4.31). The results obtained are shown in Fig. 4.2. This figure shows that the peak stresses occur at the critical speeds.

4.5 CENTRIFUGAL STRESSES

Similar procedure was applied to find centrifugal stresses at different rotor speeds.
In this case response was obtained using Eq. (4.29). The Fig. (4.3) shows that centrifugal stresses increase as the rotor speed increases. This follows from the fact that the normal acceleration increases with the increase in the angular velocity.

4.6 THERMAL STRESSES

The temperature distribution for the three-dimensional turbine blade finite element model was obtained from Eqs. (2.1) to (2.52). The thermal stress distribution was found from this temperature distribution using Eqs. (2.53) to (2.67). It was also established in the Chapter 2 that if gas temperature is raised from room temperature to $870^\circ$ C instantaneously, the thermal stresses in the blade for that particular heating rate exceed the yield stress as shown in Fig. (4.4). This problem is solved by increasing the gas temperature in stages; first to $400^\circ$ C, then to $870^\circ$ C and finally maintaining the gas temperature at $870^\circ$ C. The thermal stresses for this heating path were then calculated and found to be within the yield stress limit. These stresses are shown in Fig. (4.5).
Fig. 4.2  VIBRATORY STRESSES AT VARIOUS ROTOR SPEEDS
Fig. 4.3 CENTRIFUGAL STRESSES AT VARIOUS ROTOR SPEEDS
Fig. 4.4 THERMAL STRESSES AT VARIOUS TIMES
Fig. 4.5  TRANSIENT THERMAL STRESS ($\sigma_{th}$) DISTRIBUTION IN THE BLADE FOR THE FEASIBLE HEATING RATE OF 24°C/SEC AT DIFFERENT INSTANTS OF TIME
4.7 STRESS ANALYSIS OF BLADE DUE TO COMBINED EFFECTS

From the Figs. 4.2 to 4.5, one can determine the critical instants of time from the stress point of view. The thermal stresses are maximum at time $t = 200$ sec., and then these stresses decrease slowly with time. At time $t = 200$ sec. when the rotor speed is 2900 rpm, centrifugal stresses are also high and are still increasing with increase in rotor speed. An interesting feature is that the vibratory stresses show their peak at time $t = 204$ sec. when rotor speed is about 3000 rpm. From these observations, it is clear that between time $t = 200$ sec. and $t = 205$ sec., and between rotor speeds of 2800 rpm and 3200 rpm, all the three types of stresses are at their peak. All the stresses mentioned here are normalized stresses (normalizing factor is the yield stress). By selecting a feasible heating path for thermal stresses, the normalized thermal stresses are kept below one. Also the centrifugal stresses can be kept within the safe limit by operating the rotor at a speed within 4000 rpm. However, in case of vibratory stresses, we see normalized stresses going beyond one. From all these discussions, one can conclude that, in addition to vibratory and centrifugal stresses, the thermal stresses are also quite important in the design of the blade. In the design of an actual turbine blade, the transient forces have to be exactly known, and, therefore, the actual design would involve at first the summation of all the stresses at any instant of time, and then finding the global maxima of these types of combined stresses as the time varies. To do this, the stresses are added up and the principal stresses in each element of the turbine blade are obtained by solving the following cubic equation given by Timoshenko and Goodier (1970):
Solving the Eq. (4.32) will give the three principal stresses as \( \sigma_1, \sigma_2, \text{ and } \sigma_3 \). In order to know the state of stress in each element of the turbine blade, the design stress \( (\sigma_d) \) is found using distortion energy theory. The \( \sigma_d \) is given by the following equation:

\[
\sigma_d = \frac{1}{\sqrt{2}} \left[ (\sigma_1 - \sigma_2) + (\sigma_2 - \sigma_3) + (\sigma_3 - \sigma_1) \right]^{1/2}
\]  

(4.33)

This stress \( \sigma_d \) comprises of both mean and alternating stresses arising from thermal, centrifugal, and vibratory stresses. This gives us a clear picture about the stress history of the turbine blade and their effect on fatigue life of the turbine blade which is discussed in next article.

4.8 FATIGUE LIFE ESTIMATION

The present investigation makes use of the fatigue failure surface line by Bagci (1981). Fig. (4.6) shows the failure surface and the Bagci line where the mean stress is represented by \( \sigma_m \). This fatigue line gives the fatigue design of machine elements subjected to cyclic combined stresses having non-vanishing stress as in the present case. It is reported to fit the most recent fatigue data well and defines the two design zones, namely, the zone of fatigue failure experiencing brittle fracture and
the zone of failure due to yielding in one, and thus, providing a unified fatigue design
equation for machine members.

Bagci's fatigue failure surface line is of fourth order and is given by

\[ \sigma_{sf} = \sigma_e \left(1 - \left(\frac{\sigma_{mf}}{\sigma_y}\right)^4\right) \]  

(4.34)

where

\( \sigma_{sf} \) = failure value of alternating stress

\( \sigma_e \) = endurance limit

\( \sigma_{mf} \) = failure value of mean stress

\( \sigma_y \) = yield stress for the material

The endurance limit \( \sigma_e \) of the actual blade may be lower than the endurance limit \( \sigma_e' \) of
the standard rotating fatigue specimen due to a number of factors discussed below.

The factors are accounted for by using

\[ \sigma_e = R_i \sigma_{e}' \]  

(4.35)

where the endurance limit modifying factor

\( R_i = k_s k_b k_c k_d k_e k_f \)

with

\( k_s \) = surface factor

\( = 0.46 \) for the turbine blade material with tensile strength of 1183 MPa

\( k_b \) = size factor

\( = 1.00 \) for the thickness dimension \( d \leq 7.6 \text{ mm.} \) of the root-section of the turbine blade
\( k_c = \) reliability factor

\[ k_c = 0.86 \]

\( k_d = \) temperature factor

\[ k_d = \frac{3100}{(2400 + 9T)} \text{ for } T > 70^\circ C. \text{ Therefore for } T = 435^\circ C, k_d = 0.50 \]

\( k_n = \) stress concentration factor

\[ k_n = \frac{1}{1 + q(k_i - 1)}, q \text{ is the notch sensitivity factor, where } k_i \text{ is the factor which depends on the geometric values of the member. For a notch of 0.2 cm. radius, } \]

\[ q = 0.85, k_i = 2.6 \text{ which gives } k_n = 0.424. \]

\( k_i = \) miscellaneous factor

\[ k_i = 1.0 \text{ (assuming the blade to be undamaged).} \]

Therefore, \( R_i = 0.46 \times 1.0 \times 0.86 \times 0.50 \times 0.424 \times 1.0 = 0.084 \)

Using

\( \sigma_u \text{(ultimate stress)} = 1183 \text{ MPa} \)

\( \sigma_y \text{(yield stress)} = 855 \text{ MPa} \)

\( \sigma_e \text{(endurance limit)} = 591 \text{ MPa}, \text{ and} \)

from the values detailed above, the following stresses for the material can be calculated:

using Eq. (4.27).

\( \sigma_{af} \text{(alternating failure stress)} = 49.686 \text{ MPa} \)

\( \sigma_{mf} \text{(mean failure stress)} = 89.4348 \text{ MPa} \)
Fig. 4.6  FATIGUE FAILURE SURFACE DEFINED BY BAGCI LINE
Fig. 4.7  BLADE LOADING PATTERN
4.9 CUMULATIVE FATIGUE DAMAGE

The turbine blades discussed in the present investigation are assumed to be operating in a machine at an acceleration rate of 800 rpm/min and working for 8 hours (480 minutes) at 4000 rpm. Fig. (4.7) gives blade loading pattern. If the blade was to operate at 4000 rpm without being ever shut down, the blade will have infinite life. However, that is not the case due to the actual blade loading pattern shown in Fig. 4.7. The turbine blade is excessively stressed while passing through resonant rotor speeds, both during start-up and shut-down, in each load block. Since it was established before that the resonant stresses would be between 2800 rpm and 3200 rpm at time t = 200 sec, a check of the stress levels experienced between these two rotor speeds and at these particular time instants were made. In order to calculate the fatigue damage, the calculations of the stress amplitudes between 2800 rpm and 3200 rpm were made. The peak alternating stresses within this speed zone are shown in Fig. (4.8). From these graphs, it is clear that the resonant stresses lie above the failure value and the fatigue damage would be caused to the blade. For the estimation of fatigue damage, the resonant peaks are divided into 11 stress blocks. The rotor speeds and the corresponding stress levels are shown Table 4.1. The number of cycles N to failure is obtained by locating the load point on the failure surface and the number of cycles, n, experienced n is obtained from calculating the cycles in each stress blocks. Thus fatigue damage is obtained as a ratio of n/N.

The total fatigue damage during start-up in each stress block as can be seen in Table 4.1 is \( \sum \frac{n}{N} = 25 \times 10^{-5} \)
Fig. 4.8 MAGNIFIED STRESS LEVELS BETWEEN 3100-3200 RPM DURING START-UP AT ROTOR ACCELERATION OF 800 RPM/MIN.
Fig. 4.9  MAGNIFIED STRESS LEVELS BETWEEN 3100-3200 RPM DURING START-UP AT ROTOR ACCELERATION OF 1500 RPM/MIN.
<table>
<thead>
<tr>
<th>STRESS BLOCK NUMBER (DURING START-UP)</th>
<th>ALTERNATING STRESS (Pa)</th>
<th>MEAN SPEED (RPM)</th>
<th>CYCLES TO FAILURE (N)</th>
<th>CYCLES EXPERIENCED (N)</th>
<th>FATIGUE DAMAGE (NIN *10^6)</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>525000000</td>
<td>2800</td>
<td>1605000</td>
<td>23</td>
<td>1.433021807</td>
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<td>2</td>
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<td>1665000</td>
<td>23</td>
<td>1.381381381</td>
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<tr>
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<td>558000000</td>
<td>2880</td>
<td>1350000</td>
<td>22</td>
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<td>975000</td>
<td>27</td>
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<td>2960</td>
<td>449000</td>
<td>27</td>
<td>6.013363029</td>
</tr>
<tr>
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<td>3000</td>
<td>456500</td>
<td>13</td>
<td>2.847754655</td>
</tr>
<tr>
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<td>3040</td>
<td>585000</td>
<td>20</td>
<td>3.418803419</td>
</tr>
<tr>
<td>8</td>
<td>585000000</td>
<td>3080</td>
<td>1105000</td>
<td>21</td>
<td>1.899592944</td>
</tr>
<tr>
<td>9</td>
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<td>3120</td>
<td>1400000</td>
<td>25</td>
<td>1.785714286</td>
</tr>
<tr>
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<td>3160</td>
<td>1605000</td>
<td>20</td>
<td>1.246105919</td>
</tr>
<tr>
<td>11</td>
<td>505000000</td>
<td>3200</td>
<td>1906300</td>
<td>20</td>
<td>1.049152809</td>
</tr>
<tr>
<td>STRESS BLOCK NUMBER (DURING SHUT-DOWN)</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>1</td>
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<td>2800</td>
<td>1765000</td>
<td>22</td>
<td>1.246458924</td>
</tr>
<tr>
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<td>1832000</td>
<td>20</td>
<td>1.091703057</td>
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<tr>
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<td>557415800</td>
<td>2880</td>
<td>1490000</td>
<td>21</td>
<td>1.40935973</td>
</tr>
<tr>
<td>4</td>
<td>585965000</td>
<td>2920</td>
<td>1075200</td>
<td>18</td>
<td>1.674107143</td>
</tr>
<tr>
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<td>2960</td>
<td>493950</td>
<td>19</td>
<td>3.84653172</td>
</tr>
<tr>
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<td>3000</td>
<td>502200</td>
<td>19</td>
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<td>643750</td>
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<tr>
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<td>1540500</td>
<td>12</td>
<td>0.77867868</td>
</tr>
<tr>
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<td>52447399</td>
<td>3160</td>
<td>1766300</td>
<td>23</td>
<td>1.302378256</td>
</tr>
<tr>
<td>11</td>
<td>504459000</td>
<td>3200</td>
<td>2096390</td>
<td>13</td>
<td>0.620113624</td>
</tr>
</tbody>
</table>

Table 4.1  STRESS-BLOCK APPROXIMATION OF FATIGUE DAMAGE FOR START-UP AND SHUT DOWN OPERATION
In this table the fatigue damage during the shut-down is

\[ \Sigma \frac{n}{N} = 21 \times 10^{-5} \]

The total fatigue damage due to start-up and shut-down in each load block is thus given as

\[ 25 \times 10^{-5} + 21 \times 10^{-5} = 46 \times 10^{-5}. \]

Similar calculations with different acceleration were done. A new value of acceleration (1500 rpm/min) of blade rpm was found to be safe as the peak of the resonant stresses obtained for the rotor speed zone between 2800 rpm and 3200 rpm, were found to be below the failure value indicating no fatigue damage caused to the blade. Fig. 4.9 shows the peak alternating stress levels for a value of 1500 rpm/min. The total blade fatigue life is given by applying Miner's rule:

\[ \frac{1}{46 \times 10^{-5}} = 2.173 \times 10^3 \text{ load blocks} \]

4.10 CONCLUSIONS

In spite of the fact that various design procedures for preventing failure of turbine blades due to fatigue are being used, there is yet no indication that a satisfactory situation exists with reference to basic data that will lead to establishing sound procedures in this regard. This is because of the fact that blade fatigue is a multi-dimensional problem and as such, has been undergoing continuous investigations for long time. However, the approach discussed in the present investigation can be used effectively to identify the time instants and rotor speeds which can cause fatigue damage due to start-up and shut-down operations.
CHAPTER 5

CONCLUSIONS AND RECOMMENDATIONS

5.1 A BRIEF DISCUSSION ABOUT THE INVESTIGATION AND THE CONCLUSIONS

The objective of this investigation was to find the fatigue life of turbine blades due to combined effects of thermal and vibratory loadings. This was achieved by separately carrying out the heat transfer and vibration analysis of a blade and then adding the effects due to each of these. The heat transfer analysis was done using non-linear finite element analysis in three-dimensions. The non-linearity was due to change in material properties of the blade with temperature. The transient temperatures obtained from the heat transfer analysis were then used in the calculation of the temperature gradients and transient thermal stresses. The dynamic analysis was done at first, for finding the undamped natural frequencies of the turbine blade. These free vibration studies included the effect of the non-linearity in the stiffness matrix caused by the rotation of the turbine blade. The effect of change in the frequencies arising from the change in the material properties of the blade with temperature were also studied. Thereafter, the vibratory stresses due to (a) the nozzle excitation and (b) the centrifugal stresses at different rotor speeds were calculated. Finally, the total effect of all the three different types of stresses (transient thermal, centrifugal, and vibratory due to the nozzle excitation) on the fatigue life of the turbine blade was studied. The mathematical model for both the heat transfer and the vibratory analysis was formulated using curved, solid, C^0 continuity, quadratic, serendipity, twenty noded isoparametric finite elements.
The studies carried out in this investigation, therefore, helps one to draw the following conclusions:

1. In addition to temperature gradients along the cross-section of the blade, there exists a significant gradient along the z-direction as well.

2. In addition to stresses in z-direction, there are significant stresses in x and y directions as well, which are due to the three-dimensional temperature distribution.

3. The thermal stresses are maximum at time $t = 200$ sec. These stresses decrease slowly with time.

4. The transient thermal stress distribution in the blade can be limited well within yield stress for the feasible heating rate of $24^\circ$ C/sec.

5. The solid isoparametric finite element can be successfully used to accurately predict the three-dimensional temperature distribution.

6. The effect of including the stress stiffness matrix along with the conventional matrix makes a significant change in the natural frequencies of the blade.

7. The blade natural frequencies also vary quite significantly during the transient period.

8. The kinematic equations can be successfully used to model the nozzle excitation forces as a sinusoidal pulse.

9. The centrifugal stresses increase with increase in rotor speed.

10. The vibratory stresses show their peak at time $t = 204$ sec. when the rotor speed is about 3000 rpm.

11. Between time $t = 200$ sec. and $t = 205$ sec., and between rotor speeds of 2800
rpm and 3200 rpm, all the three types of stresses are at their peak.

12. In addition to vibratory and centrifugal stresses, the thermal stresses are also quite important in the design of the blade.

13. The turbine blade is excessively stressed while passing through resonant rotor speeds both during start-up and shut-down in each load block.

14. A new value of acceleration (1500 rpm/min) of blade was found to be safe as the peak of the resonant stresses obtained for the rotor speed zone between 2800 rpm and 3200 rpm, were found to be below the failure value indicating no fatigue damage caused to the blade.

5.2 LIMITATIONS OF THE PRESENT INVESTIGATION AND RECOMMENDATIONS FOR FUTURE WORK:

1. The calculations for non-linear temperature distribution were done assuming surrounding gases to be at uniform temperature while the gas temperatures in practice are not so. The present finite element model can be amended to include the effect of non-uniform gas temperature distribution around the blade.

2. The creep under high temperature conditions was not taken into account.

3. Fatigue being a multi-facet problem, data collection on broad blade-operational aspects is essential for definition of a specific fatigue problem.

4. The dynamic response was calculated keeping viscous damping as a function of rotor speed. The non-linearity in the damping due to dry-friction damping at the root of the blade, where it is attached to the rotor disk, should also be taken into
account.
REFERENCES


along the Blade on its Thermal Stress State", Energomashinostroenie, No. 8, pp. 10--12.


APPENDIX A

EXPRESSIONS FOR SHAPE FUNCTIONS OF CORNER NODES FOR THE
TWENTY-NODED ELEMENT

| N1 = (1/8)\*(1-\(\xi\))\*(1-\(\eta\))\*(1-\(\zeta\))\*(-\(\xi\)-\(\eta\)-\(\zeta\)-2) |
| N2 = (1/8)\*(1+\(\xi\))\*(1-\(\eta\))\*(1-\(\zeta\))\*(\(\xi\)-\(\eta\)-\(\zeta\)-2) |
| N3 = (1/8)\*(1+\(\xi\))\*(1+\(\eta\))\*(1-\(\zeta\))\*(\(\zeta\)+\(\eta\)-\(\zeta\)-2) |
| N4 = (1/8)\*(1-\(\xi\))\*(1+\(\eta\))\*(1-\(\zeta\))\*(-\(\xi\)+\(\eta\)-\(\zeta\)-2) |
| N5 = (1/8)\*(1-\(\xi\))\*(1-\(\eta\))\*(1+\(\zeta\))\*(-\(\xi\)-\(\eta\)+\(\zeta\)-2) |
| N6 = (1/8)\*(1+\(\xi\))\*(1-\(\eta\))\*(1+\(\zeta\))\*(\(\zeta\)-\(\eta\)+\(\zeta\)-2) |
| N7 = (1/8)\*(1+\(\xi\))\*(1+\(\eta\))\*(1+\(\zeta\))\*(\(\zeta\)+\(\eta\)+\(\zeta\)-2) |
| N8 = (1/8)\*(1-\(\xi\))\*(1+\(\eta\))\*(1+\(\zeta\))\*(-\(\xi\)+\(\eta\)+\(\zeta\)-2) |
**APPENDIX B**

**EXPRESSIONS FOR SHAPE FUNCTIONS OF MID-SIDE NODES FOR THE TWENTY-NODED ELEMENT**

\[
\begin{array}{|l|}
\hline
N9 = \frac{1}{4}(1-\xi^2)(1-\eta) \\
N10 = \frac{1}{4}(1-\xi^2)(1+\zeta)(1-\eta) \\
N11 = \frac{1}{4}(1-\xi^2)(1+\zeta)(1+\eta) \\
N12 = \frac{1}{4}(1-\xi^2)(1-\xi)(1+\eta) \\
N13 = \frac{1}{4}(1-\xi^2)(1-\eta)(1-\zeta) \\
N14 = \frac{1}{4}(1-\eta^2)(1+\zeta)(1-\xi) \\
N15 = \frac{1}{4}(1-\xi^2)(1+\eta)(1-\zeta) \\
N16 = \frac{1}{4}(1-\eta^2)(1-\zeta)(1-\xi) \\
N17 = \frac{1}{4}(1-\xi^2)(1-\eta)(1+\zeta) \\
N18 = \frac{1}{4}(1-\eta^2)(1+\zeta)(1+\xi) \\
N19 = \frac{1}{4}(1-\xi^2)(1+\eta)(1+\zeta) \\
N20 = \frac{1}{4}(1-\eta^2)(1-\zeta)(1+\xi) \\
\hline
\end{array}
\]
APPENDIX C

EXPRESSIONS FOR ELEMENTAL MATRICES AND VECTORS FOR THE TWENTY-NOLED ELEMENT

DESCRIPTION OF TERMS IN ELEMENTAL CAPACITANCE MATRIX \([CP]\)

\[
\iint_{\zeta_{1}}^{\zeta_{2}} \iint_{\eta_{1}}^{\eta_{2}} h_{\zeta} \left[ N_{\zeta}(\zeta, \eta, \zeta) \right]^{T} \left[ N_{\eta}(\zeta, \eta, \zeta) \right] | J(\xi, \eta, \zeta) | \, d\zeta \, d\eta \, d\zeta
\]

- \( t_1 = \zeta^{**2} \)
- \( t_2 = -1 + t_1 \)
- \( t_3 = -1 + \zeta \)
- \( t_4 = t_2 \cdot t_3 \)
- \( t_5 = 1 + \eta \)
- \( t_6 = t_5^{**2} \)
- \( t_7 = \zeta^{**2} \)
- \( t_8 = -1 + t_7 \)
- \( t_9 = t_6 \cdot t_8 \)
- \( t_{10} = -1 + \zeta \)
- \( t_{12} = t_4 \cdot t_9 \cdot t_{10} \)
- \( t_{13} = t_{10} \cdot t_{12} \)
- \( t_{14} = 1 + \zeta \)
- \( t_{15} = t_{13} \cdot t_{14} \)
\[ t_{16} = t_9 \cdot t_{15} \]
\[ t_{17} = t_{14} \cdot t_5 \]
\[ t_{18} = t_{17} \cdot t_{10} \]
\[ t_{19} = -\xi - \eta + \zeta + 2 \]
\[ t_{20} = \eta^{**} 2 \]
\[ t_{21} = -1 + t_{20} \]
\[ t_{22} = t_{19} \cdot t_{21} \]
\[ t_{23} = 1 + \zeta \]
\[ t_{24} = t_3 \cdot t_{23} \]
\[ t_{26} = t_{18} \cdot t_{22} \cdot t_{24} \]
\[ t_{27} = t_{21} \cdot t_{14} \]
\[ t_{28} = t_{27} \cdot t_{23} \]
\[ t_{29} = -1 + \eta \]
\[ t_{31} = t_8 \cdot t_{29} \cdot t_{10} \]
\[ t_{32} = t_{28} \cdot t_{31} \]
\[ t_{33} = t_8^{**} 2 \]
\[ t_{34} = t_{29}^{**} 2 \]
\[ t_{35} = t_{33} \cdot t_{34} \]
\[ t_{36} = t_{10}^{**} 2 \]
\[ t_{38} = t_{36} \cdot t_8 \]
\[ t_{39} = t_{38} \cdot t_5 \]
\[ t_{40} = t_{27} \cdot t_{39} \]
\[ t_{41} = t_{4}t_{5} \]
\[ t_{42} = t_{41}t_{31} \]
\[ t_{43} = t_{14}t_{29} \]
\[ t_{44} = t_{43}t_{10} \]
\[ t_{45} = \xi + \eta + \zeta + 2 \]
\[ t_{46} = t_{45}t_{21} \]
\[ t_{48} = t_{44}t_{46}t_{24} \]
\[ t_{49} = t_{38}t_{29} \]
\[ t_{50} = t_{51}t_{49} \]
\[ t_{51} = t_{43}t_{36} \]
\[ t_{52} = t_{45}t_{8} \]
\[ t_{54} = t_{51}t_{52}t_{5} \]
\[ t_{53} = t_{8}t_{5} \]
\[ t_{56} = t_{55}t_{23} \]
\[ t_{57} = t_{2}t_{14} \]
\[ t_{58} = t_{57}t_{29} \]
\[ t_{59} = t_{56}t_{58} \]
\[ t_{60} = t_{2}^2 \]
\[ t_{61} = t_{3}^2 \]
\[ t_{62} = t_{60}t_{61} \]
\[ t_{63} = t_{5}t_{29} \]
\[ t_{64} = t_{62}t_{63} \]
t_{65} = 143 \cdot t_{23}
\nonumber
t_{66} = \zeta \cdot \eta + \zeta \cdot 2
\nonumber
t_{67} = t_{66} \cdot t_{12}
\nonumber
t_{68} = t_{3} \cdot t_{5}
\nonumber
t_{70} = t_{65} \cdot t_{67} \cdot t_{68}
\nonumber
t_{71} = t_{33} \cdot t_{5}
\nonumber
t_{72} = t_{23} \cdot t_{29}
\nonumber
t_{74} = t_{71} \cdot t_{72} \cdot t_{10}
\nonumber
t_{75} = t_{14} \cdot t_{2}
\nonumber
t_{76} = t_{75} \cdot t_{5}
\nonumber
t_{77} = t_{23} \cdot t_{2}
\nonumber
t_{79} = \xi + \eta + \zeta \cdot 2
\nonumber
t_{80} = t_{79} \cdot t_{29}
\nonumber
t_{82} = t_{76} \cdot t_{77} \cdot t_{80} \cdot t_{66}
\nonumber
t_{83} = t_{14} \cdot t_{34}
\nonumber
t_{85} = t_{66} \cdot t_{3}
\nonumber
t_{86} = -\xi \cdot \eta + \zeta \cdot 2
\nonumber
t_{88} = t_{83} \cdot t_{77} \cdot t_{85} \cdot t_{86}
\nonumber
t_{89} = t_{61} \cdot t_{34}
\nonumber
t_{90} = t_{86} \cdot t_{2}
\nonumber
t_{94} = t_{71} \cdot t_{77} \cdot t_{29}
\nonumber
t_{95} = t_{76} \cdot t_{23}
\nonumber
\begin{align*}
t_{96} &= t_{79} t_{21} \\
t_{98} &= t_{95} t_{96} t_{10} \\
t_{99} &= t_{3} t_{29} \\
t_{100} &= t_{99} t_{23} \\
t_{101} &= t_{86} t_{2} \\
t_{103} &= t_{100} t_{101} t_{17} \\
t_{104} &= t_{17} t_{23} \\
t_{105} &= t_{79} t_{8} \\
t_{106} &= t_{29} t_{10} \\
t_{108} &= t_{104} t_{105} t_{106} \\
t_{109} &= t_{55} t_{10} \\
t_{110} &= t_{109} t_{58} \\
t_{111} &= t_{36} t_{19} \\
t_{113} &= t_{76} t_{111} t_{21} \\
t_{114} &= t_{23} t_{2} \\
t_{115} &= t_{114} t_{14} \\
t_{116} &= t_{9} t_{115} \\
t_{117} &= t_{21} t_{2} \\
t_{120} &= t_{117} t_{3} t_{36} t_{14} \\
t_{121} &= t_{21} t_{3} \\
t_{122} &= t_{77} t_{8} \\
t_{123} &= t_{122} t_{29}
\end{align*}
\[ t_{124} = t_{121} \cdot t_{123} \]
\[ t_{125} = t_{28} \cdot t_{109} \]
\[ t_{126} = t_{75} \cdot t_{34} \]
\[ t_{127} = t_{10} \cdot t_{45} \]
\[ t_{129} = t_{126} \cdot t_{127} \cdot t_{2} \]
\[ t_{130} = t_{99} \cdot t_{10} \]
\[ t_{131} = \xi + \eta + \zeta + 2 \]
\[ t_{132} = t_{131} \cdot t_{8} \]
\[ t_{133} = t_{5} \cdot t_{23} \]
\[ t_{135} = t_{130} \cdot t_{132} \cdot t_{133} \]
\[ t_{136} = t_{17} \cdot t_{36} \]
\[ t_{137} = t_{19} \cdot t_{8} \]
\[ t_{139} = t_{136} \cdot t_{137} \cdot t_{29} \]
\[ t_{140} = t_{121} \cdot t_{23} \]
\[ t_{141} = t_{140} \cdot t_{109} \]
\[ t_{142} = t_{43} \cdot t_{77} \]
\[ t_{143} = t_{66} \cdot t_{8} \]
\[ t_{145} = t_{142} \cdot t_{143} \cdot t_{5} \]
\[ t_{146} = t_{21} \cdot t_{75} \]
\[ t_{147} = t_{13} \cdot t_{29} \]
\[ t_{148} = t_{146} \cdot t_{147} \]
\[ t_{149} = t_{14} \cdot t_{6} \]
\[ t_{150} = t_{149} * t_{23} \]
\[ t_{151} = t_{179} * t_{2} \]
\[ t_{153} = t_{150} * t_{151} * t_{3} \]
\[ t_{154} = t_{61} * t_{6} \]
\[ t_{155} = \xi - \eta + \zeta + 2 \]
\[ t_{156} = t_{10} * t_{155} \]
\[ t_{158} = t_{154} * t_{156} * t_{2} \]
\[ t_{159} = t_{121} * t_{49} \]
\[ t_{160} = t_{75} * t_{29} \]
\[ t_{161} = t_{160} * t_{23} \]
\[ t_{163} = t_{161} * t_{67} * t_{5} \]
\[ t_{164} = t_{110} * t_{19} \]
\[ t_{166} = t_{23} * t_{86} \]
\[ t_{168} = t_{17} * t_{164} * t_{99} * t_{166} \]
\[ t_{169} = t_{3} * t_{34} \]
\[ t_{170} = t_{169} * t_{23} \]
\[ t_{172} = t_{170} * t_{101} * t_{14} \]
\[ t_{173} = t_{117} * t_{61} \]
\[ t_{175} = t_{61} * t_{29} \]
\[ t_{176} = t_{175} * t_{10} \]
\[ t_{177} = t_{131} * t_{21} \]
\[ t_{179} = t_{176} * t_{177} * t_{23} \]
\[ t_{180} = t_{121} \cdot t_{10} \]
\[ t_{182} = t_{180} \cdot t_{57} \cdot t_{15} \]
\[ t_{183} = t_{121} \cdot t_{39} \]
\[ t_{184} = t_{83} \cdot t_{10} \]
\[ t_{186} = t_{184} \cdot t_{52} \cdot t_{23} \]
\[ t_{187} = t_{117} \cdot t_{75} \]
\[ t_{188} = t_{23} \cdot t_{10} \]
\[ t_{189} = t_{187} \cdot t_{188} \]
\[ t_{191} = -\xi + \eta + \zeta - 2 \]
\[ t_{194} = t_{154} \cdot t_{23} \cdot t_{191} \cdot t_{10} \cdot t_{155} \]
\[ t_{195} = t_{68} \cdot t_{10} \]
\[ t_{196} = t_{155} \cdot t_{12} \]
\[ t_{198} = t_{195} \cdot t_{196} \cdot t_{43} \]
\[ t_{199} = t_{77} \cdot t_{79} \]
\[ t_{201} = t_{76} \cdot t_{199} \cdot t_{21} \]
\[ t_{202} = t_{61} \cdot t_{5} \]
\[ t_{204} = t_{191} \cdot t_{29} \]
\[ t_{206} = t_{202} \cdot t_{77} \cdot t_{204} \cdot t_{86} \]
\[ t_{207} = t_{77} \cdot t_{86} \]
\[ t_{209} = t_{169} \cdot t_{207} \cdot t_{8} \]
\[ t_{210} = t_{23} \cdot t_{66} \]
\[ t_{212} = t_{126} \cdot t_{210} \cdot t_{2} \]
t213 = t28*t41
t214 = t99*t77
t215 = t86*t8
t217 = t214*t215*t5
t218 = t114*t5
t219 = t146*t218
t220 = t117*t14
t222 = t220*t24*t10
t223 = t8*t34
t224 = t223*t15
t225 = t169*t10
t226 = t131*t2
t228 = t225*t226*t14
t229 = t75*t6
t233 = t229*t23*t79*t10*t19
t234 = t36*t131
t236 = t169*t234*t8
t237 = t33*t6
t239 = t83*t23
t240 = t10*t131
t242 = t239*t85*t240
t243 = t17*t77
\[t_{244} = t_{79}t_3\]
\[t_{247} = t_{243}t_{244}t_{29}t_{86}\]
\[t_{249} = t_{149}t_{199}t_8\]
\[t_{251} = t_{130}t_{226}t_{17}\]
\[t_{253} = t_{95}t_{151}t_{29}\]
\[t_{255} = t_{36}t_{45}\]
\[t_{257} = t_{83}t_{255}t_8\]
\[t_{258} = t_3t_{16}\]
\[t_{259} = t_{36}t_{155}\]
\[t_{261} = t_{258}t_{259}t_8\]
\[t_{263} = t_{239}t_{143}t_{10}\]
\[t_{265} = t_{225}t_{132}t_{23}\]
\[t_{266} = t_{258}t_{23}\]
\[t_{267} = t_{191}t_{14}\]
\[t_{269} = t_{266}t_{267}t_{164}\]
\[t_{270} = t_{60}t_3\]
\[t_{272} = t_{270}t_{17}t_{29}\]
\[t_{273} = t_{77}t_{66}\]
\[t_{275} = t_{160}t_{273}t_{21}\]
\[t_{278} = t_{68}t_{156}t_{43}t_{210}\]
\[t_{280} = t_{149}t_{110}\]
\[t_{282} = t_{280}t_{137}t_{23}\]
t284 = t4\cdot t29

t285 = t56\cdot t284

t286 = t68\cdot t77

t287 = t191\cdot t8

t288 = t286\cdot t287\cdot t29

t289 = t122\cdot t5

t290 = t121\cdot t290

t291 = t60\cdot t75

t292 = t292\cdot t63

t293 = t202\cdot t259\cdot t21

t294 = t114\cdot t29

t295 = t146\cdot t296

t296 = t150\cdot t105\cdot t10

t300 = t122\cdot t23

t301 = t300\cdot t204\cdot t240

t302 = t191\cdot t21

t303 = t300\cdot t303\cdot t10

t304 = t66\cdot t21

t305 = t3\cdot t10

t306 = t65\cdot t306\cdot t307

t307 = t155\cdot t21

t311 = t14\cdot t23
\[ t_{133} = 195 \times t_{310} \times t_{311} \]
\[ t_{134} = 177 \times t_{191} \]
\[ t_{136} = 1258 \times t_{314} \times t_{18} \]
\[ t_{137} = 168 \times t_{23} \]
\[ t_{138} = 114 \times t_{10} \]
\[ t_{120} = 1317 \times t_{303} \times t_{318} \]
\[ t_{122} = 175 \times t_{234} \times t_{21} \]
\[ t_{126} = 189 \times t_{23} \times t_{86} \times t_{10} \times t_{131} \]
\[ t_{127} = 145 \times t_{2} \]
\[ t_{131} = 144 \times t_{152} \times t_{133} \]
\[ t_{132} = 1258 \times t_{10} \]
\[ t_{134} = 1332 \times t_{196} \times t_{14} \]
\[ t_{135} = 1131 \times t_{2} \]
\[ t_{138} = 121 \times t_{161} \]
\[ t_{139} = 1338 \times t_{296} \]
\[ t_{140} = 128 \times t_{284} \]
\[ t_{142} = 1266 \times t_{287} \times t_{10} \]
\[ t_{143} = 127 \times t_{290} \]
\[ t_{144} = 1270 \times t_{83} \]
\[ t_{145} = 176 \times t_{110} \]
\[ t_{146} = 119 \times t_{2} \]
\[ t_{148} = 1345 \times t_{346} \times t_{129} \]
\text{t350} = t243*t96*t3 \\
t352 = t229*t164*t2 \\
t354 = t83*t273*t8 \\
t356 = t160*t255*t21 \\
t357 = t338*t218 \\
t358 = t23*t79 \\
t361 = t17*t358*t99*t240 \\
t363 = t142*t306*t3 \\
t364 = t68*t36 \\
t365 = t155*t14 \\
t368 = t364*t365*t29*t45 \\
t371 = t170*t86*t14*t127 \\
t372 = t45*t2 \\
t374 = t184*t372*t3 \\
t375 = t86*t21 \\
t377 = t100*t375*t318 \\
t379 = t317*t287*t106 \\
t380 = t202*t10 \\
t382 = t360*t196*t29 \\
t384 = t380*t310*t23 \\
t388 = t136*t19*t13*t29*t131 \\
t390 = t155*t29
\[ \begin{align*}
1392 &= t_{202} \cdot t_{36} \cdot t_{390} \cdot t_{131} \\
1394 &= t_{189} \cdot t_{166} \cdot t_{12} \\
1395 &= t_{155} \cdot t_{18} \\
1397 &= t_{364} \cdot t_{395} \cdot t_{29} \\
1398 &= t_{160} \cdot t_{110} \\
1400 &= t_{398} \cdot t_{46} \cdot t_{23} \\
1402 &= t_{118} \cdot t_{137} \cdot t_{72} \\
1404 &= t_{44} \cdot t_{372} \cdot t_{168} \\
1406 &= t_{338} \cdot t_{147} \\
1407 &= t_{119} \cdot t_{29} \\
1409 &= t_{345} \cdot t_{407} \cdot t_{210} \\
1410 &= t_{113} \cdot t_{5} \\
1411 &= t_{338} \cdot t_{410} \\
1412 &= t_{237} \cdot t_{188} \\
1414 &= t_{118} \cdot t_{346} \cdot t_{99} \\
1416 &= t_{214} \cdot t_{375} \cdot t_{14} \\
1417 &= t_{5} \cdot t_{10} \\
1419 &= t_{100} \cdot t_{215} \cdot t_{417} \\
1421 &= t_{364} \cdot t_{310} \cdot t_{14} \\
1423 &= t_{1195} \cdot t_{395} \cdot t_{172} \\
1424 &= t_{119} \cdot t_{2} \\
1429 &= t_{286} \cdot t_{267} \cdot t_{29} \cdot t_{66}
\end{align*} \]
t431 = t229 \cdot t358 \cdot t2

t432 = t114 \cdot t3

t433 = t223 \cdot t432

t435 = t95 \cdot t80 \cdot t127

t436 = t40 \cdot t31

t437 = t191 \cdot t2

t439 = t300 \cdot t437 \cdot t29

t440 = t191 \cdot t29

t444 = t345 \cdot t22 \cdot t23

t447 = t76 \cdot t36 \cdot t407 \cdot t45

t450 = t71 \cdot t36 \cdot t29

t452 = t104 \cdot t96 \cdot t307

t453 = t175 \cdot t23

t455 = t453 \cdot t101 \cdot t5

t457 = t202 \cdot t314 \cdot t21

t459 = t130 \cdot t177 \cdot t311

t460 = t270 \cdot t149

t462 = t4 \cdot t223 \cdot t10

t464 = t176 \cdot t226 \cdot t5

t465 = t180 \cdot t58

t466 = t146 \cdot t410

t468 = t266 \cdot t437 \cdot t14
\[ t_{470} = 1280 \times 1346 \times t_3 \]
\[ t_{471} = 135 \times t_{188} \]
\[ t_{472} = 127 \times t_{123} \]
\[ t_{474} = 1332 \times 1395 \times t_{23} \]
\[ t_{475} = 179^{**2} \]
\[ t_{479} = 1170 \times t{215} \times t_{10} \]
\[ t_{480} = t_{23} \times t_{191} \]
\[ t_{482} = 1154 \times t_{480} \times t_{12} \]
\[ t_{484} = 1380 \times 1390 \times t_{166} \]
\[ t_{487} = 1149 \times t_{77} \times t_{244} \times t_{191} \]
\[ t_{489} = 1286 \times t_{303} \times t_{14} \]
\[ t_{491} = 1220 \times t_{77} \times t_{3} \]
\[ t_{493} = 165 \times t_{143} \times t_{417} \]
\[ t_{495} = 1243 \times t_{105} \times t_{129} \]
\[ t_{497} = 189 \times t_{240} \times t_{2} \]
\[ t_{499} = t_{51} \times t_{46} \times t_{3} \]
\[ t_{501} = 1398 \times t_{372} \times t_{5} \]
\[ t_{503} = 1149 \times t_{111} \times t_{18} \]
\[ t_{505} = 1173 \times t_{188} \]
\[ t_{506} = 1155^{**2} \]
\[ t_{510} = 1175 \times t_{207} \times t_{21} \]
\[ t_{514} = 1169 \times t_{36} \times t_{131} \times t_{14} \times t_{45} \]
t516 = t136*t22*t3

rt 517 = t99*t36
t519 = t517*t132*t5

rt 521 = t150*t244*t156

rt 522 = t9*t432

rt 524 = t161*t306*t10

rt 525 = t223*t115

rt 527 = t317*t437*t43

rt 529 = t453*t375*t10

rt 533 = t126*t23*t66*t10*t45

rt 535 = t104*t151*t99

rt 537 = t517*t177*t14

rt 540 = t68*t480*t43*t127

rt 542 = t66**2

rt 546 = t239*t67*t13

rt 551 = t258*t36*t365*t19
## ELEMENTAL CAPACITANCE MATRIX [CP^2]

### [20 x 20]

|   |   |   |   |   |   |   |   |   |   |   |   |   |   |   |   |   |   |   |   |
| 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 | 13 | 14 | 15 | 16 | 17 | 18 | 19 | 20 |
| 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 | 13 | 14 | 15 | 16 | 17 | 18 | 19 |
| 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 | 13 | 14 | 15 | 16 | 17 | 18 |
| 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 | 13 | 14 | 15 | 16 | 17 |
| 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 | 13 | 14 | 15 | 16 |
| 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 | 13 | 14 | 15 |
| 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 | 13 | 14 |
| 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 | 13 |
| 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 | 12 |
| 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 | 11 |
| 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 | 10 |
| 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 | 9 |
| 13 | 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 | 8 |
| 14 | 13 | 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 | 7 |
| 15 | 14 | 13 | 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 | 6 |
| 16 | 15 | 14 | 13 | 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 | 5 |
| 17 | 16 | 15 | 14 | 13 | 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 | 3 | 4 |
| 18 | 17 | 16 | 15 | 14 | 13 | 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 | 2 |
| 19 | 18 | 17 | 16 | 15 | 14 | 13 | 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 |
| 20 | 19 | 18 | 17 | 16 | 15 | 14 | 13 | 12 | 11 | 10 | 9 | 8 | 7 | 6 | 5 | 4 | 3 | 2 | 1 |
DESCRIPTION OF TERMS IN ELEMENTAL TEMPERATURE-GRADIENT

INTERPOLATION MATRIX $[B^e]$:

$$
\begin{bmatrix}
\frac{\partial N_1}{\partial \xi} & \frac{\partial N_2}{\partial \xi} & \cdots & \frac{\partial N_n}{\partial \xi} \\
\frac{\partial N_1}{\partial \eta} & \frac{\partial N_2}{\partial \eta} & \cdots & \frac{\partial N_n}{\partial \eta} \\
\frac{\partial N_1}{\partial \zeta} & \frac{\partial N_2}{\partial \zeta} & \cdots & \frac{\partial N_n}{\partial \zeta}
\end{bmatrix}^{-1}
$$

Constants:

- $t_1 = \xi^2$
- $t_2 = 1 - t_1$
- $t_3 = 1 - \eta$
- $t_4 = t_2 \cdot t_3$
- $t_5 = 1 - \zeta$
- $t_6 = 1 + \eta$
- $t_7 = t_5 \cdot t_6$
- $t_8 = -\xi + \eta - \zeta - 2$
- $t_{10} = 1 - \zeta$
- $t_{11} = t_7 \cdot t_{10}$
- $t_{13} = \xi \cdot t_6$
- $t_{14} = 1 + \zeta$
- $t_{16} = t_2 \cdot t_{15}$
- $t_{17} = \xi \cdot t_{15}$
t_{19} = \eta \cdot t_{10}
\hfill (19)

\hfill (20)

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\hfill (43)

\hfill (44)

\hfill (45)

\hfill (46)

\hfill (47)

\hfill (48)

\hfill (49)

\hfill (50)

\hfill (51)

\hfill (52)

\hfill (53)
\[ t_{54} = t_{53} t_{14} \]
\[ t_{55} = \eta t_{5} \]
\[ t_{60} = t_{22} t_{14} \]
\[ t_{61} = \xi - \eta + \zeta - 2 \]
\[ t_{63} = t_{23} t_{14} \]
\[ t_{65} = t_{53} t_{22} \]
\[ t_{68} = t_{3} t_{14} \]
\[ t_{71} = t_{5} t_{14} \]
\[ t_{72} = -\xi - \eta + \zeta - 2 \]
\[ t_{74} = t_{5} t_{3} \]
\[ t_{75} = t_{74} t_{14} \]
\[ t_{77} = t_{53} t_{15} \]
\[ t_{78} = t_{6} t_{14} \]
\[ t_{79} = \xi + \eta + \zeta - 2 \]
\[ t_{81} = t_{38} t_{14} \]
\[ t_{89} = t_{53} t_{10} \]
\[ t_{94} = t_{5} t_{10} \]
\[ t_{95} = -\xi - \eta - \zeta - 2 \]
\[ t_{97} = t_{74} t_{10} \]
\[ t_{99} = -\xi + \eta + \zeta - 2 \]
\[ t_{101} = t_{7} t_{14} \]
\[ t_{110} = t_{30} t_{14} \]
ELEMENTAL TEMPERATURE-GRADIENT INTERPOLATION MATRIX \([B^*]\)

\[[3 \times 20]\]

\[
\begin{array}{cccccccccccccccccccccc}
0.059 & 0.118 & 0.177 & 0.236 & 0.302 & 0.368 & 0.434 & 0.500 & 0.566 & 0.632 & 0.698 & 0.764 & 0.830 & 0.896 & 0.962 & 1.028 & 1.094 & 1.160 \\
0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 & 0.000 \\
\end{array}
\]
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL CONDUCTION

MATRIX FOR $\xi = 1$

$$\int_{-1}^{1} \int_{-1}^{1} \left[ \mathbf{N}^\eta(\eta, \zeta) \right]^T \left[ \mathbf{N}^\eta(\eta, \zeta) \right] \left| \mathbf{J}(\eta, \zeta) \right| \, d\eta \, d\zeta$$

\[
\begin{align*}
t_1 &= 1 + \eta \\
t_2 &= t_1 \ast 2 \\
t_3 &= 1 + \zeta \\
t_4 &= t_3 \ast 2 \\
t_6 &= -1 + \eta + \zeta \\
t_7 &= t_6 \ast 2 \\
t_9 &= t_2 \ast t_3 \\
t_{10} &= -1 + \zeta \\
t_{12} &= 1 - \eta + \zeta \\
t_{14} &= t_9 \ast t_6 \ast t_{10} \ast t_{12} \\
t_{15} &= t_1 \ast t_4 \\
t_{16} &= -1 + \eta \\
t_{18} &= -1 - \eta + \zeta \\
t_{20} &= t_{15} \ast t_6 \ast t_{16} \ast t_{18} \\
t_{21} &= t_1 \ast t_3 \\
t_{23} &= t_{16} \ast t_{10} \\
t_{24} &= 1 + \eta + \zeta
\end{align*}
\]
l26 = l21 * t6 * l23 * t24
l27 = η**2
l28 = -1 + t27
l29 = l6 * t28
l30 = l15 * t29
l32 = l21 * l29 * t10
l33 = ζ**2
l34 = -1 + t33
l35 = l6 * t34
l36 = l9 * t35
l38 = l21 * l35 * t16
l39 = l10 ** 2
l41 = l12 ** 2
l43 = l1 * t10
l45 = l16 * t13
l47 = l43 * t12 * l45 * t18
l48 = l1 * t39
l51 = l48 * t12 * l16 * t24
l52 = t12 * t28
l54 = l43 * t52 * t3
l55 = l48 * t52
l57 = t12 * t34
t58 = t2*t10*t57


t60 = t43*t57*t16


t61 = t16**2


t63 = t18**2


t65 = t61*t3


t68 = t65*t18*t10*t24


t70 = t18*t28


t71 = t16*t4*t70


t73 = t45*t70*t10


t74 = t18*t34


t76 = t45*t74*t11


t77 = t65*t74


t79 = t24**2


t81 = t24*t28


t83 = t23*t81*t3


t85 = t16*t39*t81


t86 = t24*t34


t88 = t23*t86*t1


t90 = t61*t10*t86


t91 = t28**2


t94 = t91*t3*t10


t95 = t28*t3
t96 = t34*t1

197 = 195*t96

198 = t34*t116

199 = 195*t98

1101 = t28*t10

1102 = t101*t96

1103 = t101*t98

1104 = t34**2

1107 = 1104*t11*t16
SECOND PART OF ELEMENTAL CONDUCTION MATRIX FOR $\xi = 1$

$$[20 \times 20]$$

<table>
<thead>
<tr>
<th></th>
<th>t2 * t4 * t7/16</th>
<th>t14/16</th>
<th>-t20/16</th>
<th>0</th>
<th>0</th>
<th>-t26/16</th>
<th>0</th>
<th>0</th>
<th>t32/8</th>
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<th>-t36/8</th>
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<tr>
<td>t14/16</td>
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\begin{align*}
&\text{t2} \times \text{t4} \times \text{t7}/16 & 0 & 0 & \text{t14}/16 & -\text{t20}/16 & 0 & 0 & -\text{t26}/16 & 0 & 0 & \text{t32}/8 & 0 & 0 & -\text{t36}/8 & -\text{t30}/8 & 0 & \text{t38}/8 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&-\text{t20}/16 & 0 & 0 & -\text{t47}/16 & \text{t61} \times \text{t4} \times \text{t63}/16 & 0 & 0 & \text{t68}/16 & 0 & 0 & -\text{t73}/8 & 0 & 0 & \text{t76}/8 & \text{t71}/8 & 0 & -\text{t77}/8 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&-\text{t26}/16 & 0 & 0 & -\text{t51}/16 & \text{t68}/16 & 0 & 0 & \text{t61} \times \text{t39} \times \text{t79}/16 & 0 & 0 & -\text{t85}/8 & 0 & 0 & \text{t88}/8 & \text{t83}/8 & 0 & -\text{t90}/8 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&-\text{t32}/8 & 0 & 0 & \text{t55}/8 & -\text{t73}/8 & 0 & 0 & -\text{t85}/8 & 0 & 0 & \text{t91} \times \text{t39}/4 & 0 & 0 & -\text{t102}/4 & -\text{t94}/4 & 0 & 0 & \text{t103}/4 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&-\text{t36}/8 & 0 & 0 & -\text{t58}/8 & \text{t75}/8 & 0 & 0 & \text{t88}/8 & 0 & 0 & -\text{t1102}/4 & 0 & 0 & \text{t104} \times \text{t12}/4 & \text{t97}/4 & 0 & 0 & -\text{t107}/4 \\
&-\text{t30}/8 & 0 & 0 & -\text{t54}/8 & \text{t71}/8 & 0 & 0 & \text{t83}/8 & 0 & 0 & -\text{t194}/4 & 0 & 0 & \text{t197}/4 & \text{t91} \times \text{t14}/4 & 0 & 0 & -\text{t99}/4 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
&-\text{t38}/8 & 0 & 0 & \text{t60}/8 & -\text{t177}/8 & 0 & 0 & -\text{t90}/8 & 0 & 0 & \text{t103}/4 & 0 & 0 & -\text{t107}/4 & -\text{t99}/4 & 0 & 0 & \text{t104} \times \text{t61}/4
\end{align*}
$$
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL CONDUCTION

MATRIX FOR $\xi = -1$

$$\iint_{-1}^{1} \left[ N^\eta(\eta, \zeta) \right]^T \left[ N^\eta(\eta, \zeta) \right] | J(\eta, \zeta) | \ d\eta \ d\zeta$$

$t_1 = 1 + \eta$
$t_2 = t_1^2$
$t_3 = 1 + \zeta$
$t_4 = t_3^2$
$t_6 = -1 + \eta + \zeta$
$t_7 = t_6^2$
$t_9 = t_2^3$
$t_{10} = -1 + \zeta$
$t_{12} = 1 - \eta + \zeta$
$t_{14} = t_9^4 * t_{10} * t_{12}$
$t_{15} = t_1^4$
$t_{16} = -1 + \eta$
$t_{18} = -1 - \eta + \zeta$
$t_{20} = t_{15}^4 * t_{16} * t_{18}$
$t_{21} = t_1^3$
$t_{23} = t_{15}^2 * t_{10}$
$t_{24} = 1 + \eta + \zeta$
\[ t_{26} = t_{21} * t_6 * t_{23} * t_{24} \]
\[ t_{27} = \eta^{**2} \]
\[ t_{28} = -1 + t_{27} \]
\[ t_{29} = t_6 * t_{28} \]
\[ t_{30} = t_{15} * t_{29} \]
\[ t_{32} = t_{21} * t_{29} * t_{10} \]
\[ t_{33} = \zeta^{**2} \]
\[ t_{34} = -1 + t_{33} \]
\[ t_{35} = t_6 * t_{34} \]
\[ t_{36} = t_9 * t_{35} \]
\[ t_{38} = t_{21} * t_{35} * t_{16} \]
\[ t_{39} = t_{10}^{**2} \]
\[ t_{41} = t_{12}^{**2} \]
\[ t_{43} = t_{11} * t_{10} \]
\[ t_{45} = t_{16} * t_{3} \]
\[ t_{47} = t_{43} * t_{12} * t_{45} * t_{18} \]
\[ t_{48} = t_{11} * t_{39} \]
\[ t_{51} = t_{48} * t_{12} * t_{16} * t_{24} \]
\[ t_{52} = t_{12} * t_{28} \]
\[ t_{54} = t_{43} * t_{52} * t_{13} \]
\[ t_{55} = t_{46} * t_{52} \]
\[ t_{57} = t_{12} * t_{34} \]
\[ t_{58} = t_{12} \cdot t_{10} \cdot t_{57} \]
\[ t_{60} = t_{43} \cdot t_{57} \cdot t_{16} \]
\[ t_{61} = t_{16} \cdot t_{10} \]
\[ t_{63} = t_{18} \cdot t_{10} \]
\[ t_{65} = t_{61} \cdot t_{13} \]
\[ t_{68} = t_{65} \cdot t_{18} \cdot t_{10} \cdot t_{24} \]
\[ t_{70} = t_{18} \cdot t_{28} \]
\[ t_{71} = t_{16} \cdot t_{14} \cdot t_{70} \]
\[ t_{73} = t_{45} \cdot t_{70} \cdot t_{10} \]
\[ t_{74} = t_{18} \cdot t_{34} \]
\[ t_{76} = t_{45} \cdot t_{74} \cdot t_{1} \]
\[ t_{77} = t_{65} \cdot t_{74} \]
\[ t_{79} = t_{124} \cdot t_{28} \]
\[ t_{81} = t_{124} \cdot t_{28} \]
\[ t_{83} = t_{123} \cdot t_{81} \cdot t_{13} \]
\[ t_{85} = t_{116} \cdot t_{39} \cdot t_{81} \]
\[ t_{86} = t_{124} \cdot t_{34} \]
\[ t_{88} = t_{123} \cdot t_{86} \cdot t_{1} \]
\[ t_{90} = t_{61} \cdot t_{10} \cdot t_{86} \]
\[ t_{91} = t_{128} \cdot t_{28} \]
\[ t_{94} = t_{191} \cdot t_{3} \cdot t_{10} \]
\[ t_{95} = t_{128} \cdot t_{13} \]
t96 = t34*t1

t97 = t95*t96

t98 = t34*t16

t99 = t95*t98

t101 = t28*t10

t102 = t101*t96

t103 = t101*t98

t104 = t34**2

t107 = t104*t1*t16
SECOND PART OF ELEMENTAL CONDUCTION MATRIX FOR $\xi = -1$

$$[20 \times 20]$$

\begin{array}{cccccccccccccccccccc}
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & t2*t4*t7/16 & t14/16 & 0 & 0 & -t20/16 & -t26/16 & 0 & 0 & -t30/8 & 0 & 0 & -t36/8 & 0 & 0 & t38/8 & t32/8 & 0 \\
0 & t14/16 & t2*t39*t41/16 & 0 & 0 & -t47/16 & -t51/16 & 0 & 0 & -t54/8 & 0 & 0 & -t58/8 & 0 & 0 & t60/8 & t55/8 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & -t20/16 & -t47/16 & 0 & t61*t4*t63/16 & t68/16 & 0 & 0 & t71/8 & 0 & 0 & t76/8 & 0 & 0 & -t77/8 & -t73/8 & 0 \\
0 & -t26/16 & -t51/16 & 0 & t68/16 & t61*t39*t79/16 & 0 & t83/8 & 0 & 0 & t88/8 & 0 & 0 & -t90/8 & -t85/8 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & -t30/8 & -t54/8 & 0 & t71/8 & t83/8 & 0 & 0 & t91*t4/4 & 0 & 0 & t97/4 & 0 & 0 & -t99/4 & -t94/4 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & -t36/8 & -t58/8 & 0 & t76/8 & t88/8 & 0 & 0 & t97/4 & 0 & 0 & t1104*t2/4 & 0 & 0 & -t107/4 & -t102/4 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & t38/8 & 160/8 & 0 & -t77/8 & -t90/8 & 0 & 0 & -t99/4 & 0 & 0 & -t107/4 & 0 & 0 & t104*t61/4 & 1103/4 & 0 \\
0 & t32/8 & t55/8 & 0 & -t73/8 & -t85/8 & 0 & -t94/4 & 0 & 0 & -t1102/4 & 0 & 0 & t103/4 & 91*t39/4 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
\end{array}
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL CONDUCTION

MATRIX FOR \( \eta = 1 \)

\[
\int \int [N^T(\xi, \zeta)]^T [N(\xi, \zeta)] J(\xi, \zeta) \ d\xi \ d\zeta
\]

\[
t_1 = 1 + \xi
\]
\[
t_2 = t_1^2
\]
\[
t_3 = 1 + \zeta
\]
\[
t_4 = t_3^2
\]
\[
t_6 = \xi - 1 + \zeta
\]
\[
t_7 = t_6^2
\]
\[
t_9 = t_1 t_4
\]
\[
t_{10} = -1 + \xi
\]
\[
t_{12} = -\xi - 1 + \zeta
\]
\[
t_{14} = t_9 t_6 t_{10} t_{12}
\]
\[
t_{15} = t_1 t_3
\]
\[
t_{17} = -1 + \zeta
\]
\[
t_{18} = t_{10} t_{17}
\]
\[
t_{19} = \xi + 1 + \zeta
\]
\[
t_{21} = t_{15} t_6 t_{18} t_{19}
\]
\[
t_{22} = t_2 t_3
\]
\[
t_{24} = -\xi + 1 + \zeta
\]
t26 = t22*t6*t17*t24

t27 = ω**2

t28 = -1+t27

t29 = t6*t28

t30 = t9*t29

t31 = ω**2

t32 = -1+t31

t33 = t6*t32

t35 = t15*t33*t10

t37 = t15*t29*t17

t38 = t22*t33

t39 = t10**2

t41 = t12**2

t43 = t39*t3

t46 = t43*t12*t17*t19

t47 = t10*t3

t49 = t1*t17

t51 = t47*t12*t49*t24

t53 = t12*t28

t54 = t10*t4*t53

t55 = t12*t32

t56 = t43*t55
156 = 147*53*17
160 = 147*55*11
161 = 17**2
163 = 19**2
165 = 110*161
168 = 165*19*1*1*24
169 = 119*28
171 = 118*69*13
173 = 119*32
174 = 139*17*173
175 = 165*69
177 = 118*173*11
179 = 124**2
181 = 124*128
183 = 149*81*13
184 = 124*32
186 = 149*84*110
188 = 11*61*181
190 = 12*17*184
191 = 128**2
193 = 128*3
194 = 132*10
t95 = t93*t94

t97 = t91*t3*t17

t98 = t32*t1

t99 = t93*t98

t100 = t32**2

t102 = t28*t17

t103 = t94*t102

t105 = t100*t10*t1

t107 = t102*t98
SECOND PART OF ELEMENTAL CONDUCTION MATRIX FOR $\eta = 1$

$$[20 \times 20]$$

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<tr>
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<th>$t2^*t4^*t7/16$</th>
<th>$-t14/16$</th>
<th>$-t21/16$</th>
<th>$t26/16$</th>
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<td>$-t21/16$</td>
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<td>$t26/16$</td>
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<td>$t2^*t61^*t79/16$</td>
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<td>$-t83/8$</td>
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DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL CONDUCTION

MATRICES FOR $\eta = -1$

\[
\int_{-1}^{1} \int_{-1}^{1} \begin{bmatrix} N^\eta(\xi, \zeta) \end{bmatrix}^T \begin{bmatrix} N^\eta(\xi, \zeta) \end{bmatrix} \begin{bmatrix} J(\xi, \zeta) \end{bmatrix} \, d\xi \, d\zeta
\]

$t_1 = 1 + \xi$
$t_2 = t_1 \times 2$
$t_3 = 1 + \zeta$
$t_4 = t_3 \times 2$
$t_6 = \xi - 1 + \zeta$
$t_7 = t_6 \times 2$
$t_9 = t_1 \times t_4$
$t_{10} = -1 + \xi$
$t_{12} = -\xi - 1 + \zeta$
$t_{14} = t_9 \times t_6 \times t_{10} \times t_{12}$
$t_{15} = t_1 \times t_3$
$t_{17} = -1 + \zeta$
$t_{18} = t_{10} \times t_{17}$
$t_{19} = \xi + 1 + \zeta$
$t_{21} = t_{15} \times t_6 \times t_{18} \times t_{19}$
$t_{22} = t_2 \times t_3$
$t_{24} = -\xi + 1 + \zeta$
\[ t_{26} = t_{22} \cdot t_{16} \cdot t_{17} \cdot t_{24} \]
\[ t_{27} = \xi^2 \]
\[ t_{28} = -1 + t_{27} \]
\[ t_{29} = t_{6} \cdot t_{28} \]
\[ t_{30} = t_{9} \cdot t_{29} \]
\[ t_{31} = \zeta^2 \]
\[ t_{32} = -1 + t_{31} \]
\[ t_{33} = t_{6} \cdot t_{32} \]
\[ t_{35} = t_{15} \cdot t_{33} \cdot t_{10} \]
\[ t_{37} = t_{15} \cdot t_{29} \cdot t_{17} \]
\[ t_{38} = t_{22} \cdot t_{33} \]
\[ t_{39} = t_{10}^2 \]
\[ t_{41} = t_{12}^2 \]
\[ t_{43} = t_{39} \cdot t_{13} \]
\[ t_{46} = t_{43} \cdot t_{12} \cdot t_{17} \cdot t_{19} \]
\[ t_{47} = t_{10} \cdot t_{13} \]
\[ t_{49} = t_{1} \cdot t_{17} \]
\[ t_{51} = t_{47} \cdot t_{12} \cdot t_{49} \cdot t_{24} \]
\[ t_{53} = t_{12} \cdot t_{128} \]
\[ t_{54} = t_{10} \cdot t_{14} \cdot t_{153} \]
\[ t_{55} = t_{12} \cdot t_{32} \]
\[ t_{56} = t_{43} \cdot t_{155} \]
t58 = t47*t53*t17


t60 = t47*t55*t1


t61 = t17**2


t63 = t19**2


t65 = t10*t61


t68 = t65*t19*t1*t24


t69 = t19*t28


t71 = t18*t69*t3


t73 = t19*t32


t74 = t39*t17*t73


t75 = t65*t69


t77 = t18*t73*t1


t79 = t24**2


t81 = t24*t28


t83 = t49*t81*t3


t84 = t24*t32


t86 = t49*t84*t10


t88 = t1*t61*t11


t90 = t2*t17*t84


t91 = t28**2


t93 = t28*t3


t94 = t32*t10
\[ t_{95} = t_{93} \times t_{94} \]
\[ t_{97} = t_{91} \times t_{3} \times t_{17} \]
\[ t_{98} = t_{32} \times t_{1} \]
\[ t_{99} = t_{93} \times t_{98} \]
\[ t_{100} = t_{32} \times t_{2} \]
\[ t_{102} = t_{28} \times t_{17} \]
\[ t_{103} = t_{94} \times t_{102} \]
\[ t_{105} = t_{100} \times t_{10} \times t_{1} \]
\[ t_{107} = t_{102} \times t_{98} \]
SECOND PART OF ELEMENTAL CONDUCTION MATRIX FOR $\eta = -1$

[20 x 20]

$$
\begin{array}{ccccccccccccccccccc}
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & t2*t4*t7/16 & -t14/16 & -t21/16 & t26/16 & 0 & 0 & 0 & 0 & -t30/8 & t35/8 & t37/8 & -t38/8 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & -t14/16 & t39*t4*t41/16 & t46/16 & -t51/16 & 0 & 0 & 0 & 0 & t54/8 & -t56/8 & -t58/8 & t60/8 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & -t21/16 & t46/16 & t39*t61*t63/16 & -t68/16 & 0 & 0 & 0 & 0 & t71/8 & -t74/8 & -t75/8 & t77/8 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & t26/16 & -t51/16 & -t68/16 & t2*t61*t79/16 & 0 & 0 & 0 & 0 & -t83/8 & t86/8 & t88/8 & -t90/8 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & -t30/8 & t54/8 & t71/8 & -t83/8 & 0 & 0 & 0 & 0 & t91*t4/4 & -t95/4 & -t97/4 & t99/4 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & t35/8 & -t56/8 & -t74/8 & t86/8 & 0 & 0 & 0 & 0 & -t95/4 & t100*t39/4 & t103/4 & -t105/4 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & t37/8 & -t58/8 & -t75/8 & t88/8 & 0 & 0 & 0 & 0 & -t97/4 & t103/4 & t191*t61/4 & -t107/4 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & -t38/8 & t60/8 & t77/8 & -t90/8 & 0 & 0 & 0 & 0 & t99/4 & -t105/4 & -t107/4 & t100*t2/4 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
\end{array}
$$
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL CONDUCTION

MATRIX FOR $\zeta = 1$

$$
\int \int \left[ N^e(\xi, \eta) \right]^T \left[ N^e(\xi, \eta) \right] J(\xi, \eta) \, d\xi \, d\eta
$$

$t_1 = 1 + \xi$

$t_2 = t_1^2$

$t_3 = 1 + \eta$

$t_4 = t_3^2$

$t_6 = \xi + \eta - 1$

$t_7 = t_6^2$

$t_9 = t_1^3 t_4$

$t_{10} = -1 + \xi$

$t_{12} = \xi - \eta + 1$

$t_{14} = t_9 t_6 t_{10} t_{12}$

$t_{15} = t_2 t_3$

$t_{16} = -1 + \eta$

$t_{18} = \xi - \eta - 1$

$t_{20} = t_1 t_6 t_{16} t_{18}$

$t_{21} = t_1^3 t_3$

$t_{23} = t_10 t_{16}$

$t_{24} = \xi + \eta + 1$
\( t_{26} = t_{21} \cdot t_{6} \cdot t_{23} \cdot t_{24} \)
\( t_{27} = \eta^{**2} \)
\( t_{28} = -1 + t_{27} \)
\( t_{29} = t_{6} \cdot t_{28} \)
\( t_{30} = t_{15} \cdot t_{29} \)
\( t_{32} = t_{21} \cdot t_{29} \cdot t_{10} \)
\( t_{33} = \xi^{**2} \)
\( t_{34} = -1 + t_{33} \)
\( t_{35} = t_{6} \cdot t_{34} \)
\( t_{36} = t_{9} \cdot t_{35} \)
\( t_{38} = t_{21} \cdot t_{35} \cdot t_{16} \)
\( t_{39} = t_{10}^{**2} \)
\( t_{41} = t_{12}^{**2} \)
\( t_{43} = t_{10} \cdot t_{3} \)
\( t_{45} = t_{1} \cdot t_{16} \)
\( t_{47} = t_{43} \cdot t_{12} \cdot t_{45} \cdot t_{18} \)
\( t_{48} = t_{39} \cdot t_{3} \)
\( t_{51} = t_{48} \cdot t_{12} \cdot t_{16} \cdot t_{24} \)
\( t_{52} = t_{12} \cdot t_{28} \)
\( t_{54} = t_{43} \cdot t_{52} \cdot t_{1} \)
\( t_{55} = t_{48} \cdot t_{52} \)
\( t_{57} = t_{12} \cdot t_{134} \)
\( t_{58} = t_{10} \cdot t_{4} \cdot t_{57} \)
\( t_{60} = t_{43} \cdot t_{57} \cdot t_{116} \)
\( t_{61} = t_{16} \cdot t_{2} \)
\( t_{63} = t_{18} \cdot t_{2} \)
\( t_{65} = t_{1} \cdot t_{61} \)
\( t_{68} = t_{65} \cdot t_{118} \cdot t_{110} \cdot t_{24} \)
\( t_{70} = t_{18} \cdot t_{28} \)
\( t_{71} = t_{2} \cdot t_{16} \cdot t_{70} \)
\( t_{73} = t_{45} \cdot t_{70} \cdot t_{110} \)
\( t_{74} = t_{18} \cdot t_{34} \)
\( t_{76} = t_{45} \cdot t_{74} \cdot t_{13} \)
\( t_{77} = t_{65} \cdot t_{74} \)
\( t_{79} = t_{24} \cdot t_{2} \)
\( t_{81} = t_{24} \cdot t_{128} \)
\( t_{83} = t_{23} \cdot t_{181} \cdot t_{1} \)
\( t_{85} = t_{39} \cdot t_{116} \cdot t_{181} \)
\( t_{86} = t_{24} \cdot t_{34} \)
\( t_{88} = t_{23} \cdot t_{186} \cdot t_{3} \)
\( t_{90} = t_{10} \cdot t_{161} \cdot t_{186} \)
\( t_{91} = t_{28} \cdot t_{2} \)
\( t_{94} = t_{91} \cdot t_{1} \cdot t_{10} \)
\( t_{95} = t_{28} \cdot t_{1} \)
t96 = t34*t3

197 = 195*196

t98 = t34*t16

199 = 195*198

t101 = t28*t10

102 = t101*t96

103 = t101*t98

104 = t34**2

107 = t104*t3*t16
SECOND PART OF ELEMENTAL CONDUCTION MATRIX FOR $\zeta = 1$

$$[20 \times 20] $$

\[
\begin{array}{cccccccccccccccccc}
 t2^*t4^*7/16 & t14/16 & 0 & 0 & -t20/16 & -t26/16 & 0 & 0 & -t36/8 & 0 & 0 & t38/8 & 0 & 0 & -t30/8 & t32/8 & 0 & 0 \\
 t14/16 & t39^*t4^*t41/16 & 0 & 0 & -t47/16 & -t51/16 & 0 & 0 & -t58/8 & 0 & 0 & t60/8 & 0 & 0 & -t54/8 & t55/8 & 0 & 0 \\
 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 -t20/16 & -t47/16 & 0 & 0 & t2^*t61^*t63/16 & t68/16 & 0 & 0 & t76/8 & 0 & 0 & -t77/8 & 0 & 0 & t71/8 & -t73/8 & 0 & 0 \\
 -t26/16 & -t51/16 & 0 & 0 & t68/16 & t39^*t61^*t79/16 & 0 & 0 & t88/8 & 0 & 0 & -t90/8 & 0 & 0 & t83/8 & -t85/8 & 0 & 0 \\
 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 -t36/8 & -t58/8 & 0 & 0 & t76/8 & t88/8 & 0 & 0 & t104^*t4/4 & 0 & 0 & -t107/4 & 0 & 0 & t197/4 & -t102/4 & 0 & 0 \\
 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 -t30/8 & -t54/8 & 0 & 0 & t71/8 & t83/8 & 0 & 0 & t97/4 & 0 & 0 & -t99/4 & 0 & 0 & t91^*t2/4 & -t94/4 & 0 & 0 \\
 t32/8 & t55/8 & 0 & 0 & -t73/8 & -t85/8 & 0 & 0 & -t102/4 & 0 & 0 & t103/4 & 0 & 0 & -t94/4 & t91^*t39/4 & 0 & 0 \\
 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
\end{array}
\]
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL CONDUCTION

MATRIX FOR $\zeta = -1$

\[
\int \int_{-1}^{1} [N(x, \eta)]^T [N(x, \eta)] | J(x, \eta) | d\xi \, d\eta
\]

t1 = -1 + \xi

t2 = t1**2

t3 = 1 + \eta

t4 = t3**2

t6 = \xi - \eta + 1

t7 = t6**2

t9 = t1*t4

t10 = 1 + \xi

t12 = \xi + \eta - 1

t14 = t9*t6*t10*t12

t15 = t2*t3

t16 = -1 + \eta

t18 = \xi + \eta + 1

t20 = t15*t6*t16*t18

t21 = t1*t3

t23 = t10*t16

t24 = \xi - \eta - 1

182
\[
\begin{align*}
\tau_{26} &= t_{21}t_{6}t_{23}t_{24} \\
\tau_{27} &= \eta^{**2} \\
\tau_{28} &= -1 + \tau_{27} \\
\tau_{29} &= t_{6}\tau_{28} \\
\tau_{30} &= t_{15}\tau_{29} \\
\tau_{32} &= t_{21}\tau_{29}\tau_{10} \\
\tau_{33} &= \xi^{**2} \\
\tau_{34} &= -1 + \tau_{33} \\
\tau_{35} &= t_{6}\tau_{34} \\
\tau_{36} &= t_{9}\tau_{35} \\
\tau_{38} &= t_{21}\tau_{35}\tau_{16} \\
\tau_{39} &= t_{10}^{**2} \\
\tau_{41} &= t_{12}^{**2} \\
\tau_{43} &= t_{10}\tau_{3} \\
\tau_{45} &= t_{1}\tau_{16} \\
\tau_{47} &= t_{43}\tau_{12}\tau_{45}\tau_{18} \\
\tau_{48} &= \tau_{39}\tau_{3} \\
\tau_{51} &= \tau_{48}\tau_{12}\tau_{16}\tau_{24} \\
\tau_{52} &= \tau_{12}\tau_{28} \\
\tau_{54} &= \tau_{43}\tau_{52}\tau_{1} \\
\tau_{55} &= \tau_{48}\tau_{52} \\
\tau_{57} &= \tau_{12}\tau_{34} 
\end{align*}
\]
t58 = t10*t4*t57
t60 = t43*t57*t16
t61 = t16**2
t63 = t18**2
t65 = t1*t61
t68 = t65*t18*t10*t24
t70 = t18*t28
t71 = t2*t16*t70
t73 = t45*t70*t10
t74 = t18*t34
t76 = t45*t74*t3
t77 = t65*t74
t79 = t24**2
t81 = t24*t28
t83 = t23*t81*t1
t85 = t39*t16*t81
t86 = t24*t34
t88 = t23*t86*t3
t90 = t10*t61*t86
t91 = t28**2
t94 = t91*t1*t10
t95 = t28*t1
t96 = t34*t13
t97 = t95*t96
t98 = t34*t16
t99 = t95*t98
t101 = t28*t10
t102 = t101*t96
t103 = t101*t98
t104 = t34**2
t107 = t104*t3*t16
SECOND PART OF ELEMENTAL CONDUCTION MATRIX FOR $\zeta = -1$

$$[20 \times 20]$$

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]

\[
\begin{array}{ccccccccccccccccccc}
0 & 0 & t^2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{array}
\]
DESCRIPTION OF TERMS IN ELEMENTAL CONVECTION VECTOR \([F_{\xi, \eta}]\) FOR \(\xi = 1\)

\[
\int_{-1}^{1} \int_{-1}^{1} \left[ N^e(\eta, \xi) \right]^T |J(\eta, \xi)| d\eta \, d\xi
\]

\(t_1 = 1 + \eta\)
\(t_2 = 1 + \xi\)
\(t_6 = 1 - \xi\)
\(t_{10} = 1 - \eta\)
\(t_{17} = \eta^{**2}\)
\(t_{18} = 1 - t_{17}\)
\(t_{21} = \xi^{**2}\)
\(t_{22} = 1 - t_{21}\)
| \( \text{FC}1(1,1) \)  | \( \text{FC}1(2,1) = 0 \)  | \( \text{FC}1(3,1) = 0 \)  | \( \text{FC}1(4,1) = t1^2 t^2(-1+\eta+\zeta)/4 \)  | \( \text{FC}1(5,1) = t10^2 t^2(-1-\eta+\zeta)/4 \)  | \( \text{FC}1(6,1) = 0 \)  | \( \text{FC}1(7,1) = 0 \)  | \( \text{FC}1(8,1) = t10^6 t^6(-1-\eta-\zeta)/4 \)  | \( \text{FC}1(9,1) = 0 \)  | \( \text{FC}1(10,1) = 0 \)  | \( \text{FC}1(11,1) = 0 \)  | \( \text{FC}1(12,1) = t22 t^1/2 \)  | \( \text{FC}1(13,1) = 0 \)  | \( \text{FC}1(14,1) = 0 \)  | \( \text{FC}1(15,1) = 0 \)  | \( \text{FC}1(16,1) = t22 t^10/2 \)  | \( \text{FC}1(17,1) = t18 t^2/2 \)  | \( \text{FC}1(18,1) = 0 \)  | \( \text{FC}1(19,1) = 0 \)  | \( \text{FC}1(20,1) = t18 t^8/2 \)  |
DESCRIPTION OF TERMS IN ELEMENTAL CONVECTION VECTOR \( \mathbf{F}_c \) FOR \( \xi = -1 \)

\[
\int_{-1}^{1} \int_{-1}^{1} [N^c(\eta, \xi)]^T |J(\eta, \xi)| \, d\eta \, d\xi
\]

\( t1 = 1 + \eta \)
\( t2 = 1 + \xi \)
\( t6 = 1 - \xi \)
\( t10 = 1 - \eta \)
\( t17 = \eta^2 \)
\( t18 = 1 - t17 \)
\( t21 = \xi^2 \)
\( t22 = 1 - t21 \)
ELEMENTAL CONVECTION VECTOR (FC2) FOR $\xi = -1$

<table>
<thead>
<tr>
<th>$i$</th>
<th>$FC_2(i,1)$</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>0</td>
</tr>
<tr>
<td>2</td>
<td>$t_1^2(-1+\eta+\xi)/4$</td>
</tr>
<tr>
<td>3</td>
<td>$t_1^6(-1+\eta-\xi)/4$</td>
</tr>
<tr>
<td>4</td>
<td>0</td>
</tr>
<tr>
<td>5</td>
<td>0</td>
</tr>
<tr>
<td>6</td>
<td>$t_2^2(-1-\eta+\xi)/4$</td>
</tr>
<tr>
<td>7</td>
<td>$t_2^6(-1-\eta-\xi)/4$</td>
</tr>
<tr>
<td>8</td>
<td>0</td>
</tr>
<tr>
<td>9</td>
<td>0</td>
</tr>
<tr>
<td>10</td>
<td>$t_2^2t_1/2$</td>
</tr>
<tr>
<td>11</td>
<td>0</td>
</tr>
<tr>
<td>12</td>
<td>0</td>
</tr>
<tr>
<td>13</td>
<td>0</td>
</tr>
<tr>
<td>14</td>
<td>$t_2^2t_10/2$</td>
</tr>
<tr>
<td>15</td>
<td>0</td>
</tr>
<tr>
<td>16</td>
<td>0</td>
</tr>
<tr>
<td>17</td>
<td>0</td>
</tr>
<tr>
<td>18</td>
<td>$t_1^2t_2/2$</td>
</tr>
<tr>
<td>19</td>
<td>$t_1^2t_6/2$</td>
</tr>
<tr>
<td>20</td>
<td>0</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN ELEMENTAL CONVECTION VECTOR $[F_c^\eta]$ FOR $\eta = 1$

\[
\int_{-1}^{1} \int_{-1}^{1} \left[ N^\eta_0(\xi, \zeta) \right]^T \cdot V(\xi, \zeta) \, d\xi \, d\zeta
\]

\begin{align*}
11 &= 1 + \xi \\
12 &= 1 + \zeta \\
16 &= 1 - \xi \\
10 &= 1 - \zeta \\
17 &= \xi^2 \\
18 &= 1 - 17 \\
19 &= \zeta^2 \\
20 &= 1 - 21 \\
21 &= 1 - 20 \\
\end{align*}
E L E M E N T A L  C O N V E C T I O N  V E C T O R \{ F C 3 \}  F O R  \eta = 1

<p>| | | | | | | |</p>
<table>
<thead>
<tr>
<th></th>
<th></th>
<th></th>
<th></th>
<th></th>
<th></th>
<th></th>
</tr>
</thead>
<tbody>
<tr>
<td>FC3(1,1)</td>
<td>=</td>
<td>t1<em>t2</em>(\xi-1+\zeta)/4</td>
<td>FC3(2,1)</td>
<td>=</td>
<td>t6<em>t2</em>(-\xi+1+\zeta)/4</td>
<td>FC3(3,1)</td>
</tr>
<tr>
<td>FC3(4,1)</td>
<td>=</td>
<td>t1<em>t10</em>(-\xi-1-\zeta)/4</td>
<td>FC3(5,1)</td>
<td>=</td>
<td>0</td>
<td>FC3(6,1)</td>
</tr>
<tr>
<td>FC3(7,1)</td>
<td>=</td>
<td>0</td>
<td>FC3(8,1)</td>
<td>=</td>
<td>0</td>
<td>FC3(9,1)</td>
</tr>
<tr>
<td>FC3(10,1)</td>
<td>=</td>
<td>t21*t6/2</td>
<td>FC3(11,1)</td>
<td>=</td>
<td>t18*t10/2</td>
<td>FC3(12,1)</td>
</tr>
<tr>
<td>FC3(13,1)</td>
<td>=</td>
<td>0</td>
<td>FC3(14,1)</td>
<td>=</td>
<td>0</td>
<td>FC3(15,1)</td>
</tr>
<tr>
<td>FC3(16,1)</td>
<td>=</td>
<td>0</td>
<td>FC3(17,1)</td>
<td>=</td>
<td>0</td>
<td>FC3(18,1)</td>
</tr>
<tr>
<td>FC3(19,1)</td>
<td>=</td>
<td>0</td>
<td>FC3(20,1)</td>
<td>=</td>
<td>0</td>
<td></td>
</tr>
</tbody>
</table>

192
DESCRIPTION OF TERMS IN ELEMENTAL CONVECTION VECTOR \([F_c^\circ]\) FOR \(\eta = 1\)

\[
\int_{-1}^{1} \int_{-1}^{1} \left[N_c^\circ(\xi, \zeta) \right]^T \cdot J(\xi, \zeta) \cdot d\xi \cdot d\zeta
\]

\[
t_1 = 1 + \zeta
\]
\[
t_2 = 1 + \zeta
\]
\[
t_6 = 1 - \zeta
\]
\[
t_{10} = 1 - \zeta
\]
\[
t_{17} = \zeta^{**2}
\]
\[
t_{18} = 1 - t_{17}
\]
\[
t_{20} = \zeta^{**2}
\]
\[
t_{21} = 1 - t_{20}
\]
ELEMENTAL CONVECTION VECTOR \( \{ FC_4 \} \) FOR \( \eta = -1 \)

| \( FC_4(1,1) \) | 0 |
| \( FC_4(2,1) \) | 0 |
| \( FC_4(3,1) \) | 0 |
| \( FC_4(4,1) \) | 0 |
| \( FC_4(5,1) \) | \( t_1 t_2 (\xi - 1 + \zeta)/4 \) |
| \( FC_4(6,1) \) | \( t_6 t_2 (-\xi - 1 + \zeta)/4 \) |
| \( FC_4(7,1) \) | \( t_6 t_10 (-\xi - 1 - \zeta)/4 \) |
| \( FC_4(8,1) \) | \( t_11 t_10 (\xi - 1 - \zeta)/4 \) |
| \( FC_4(9,1) \) | 0 |
| \( FC_4(10,1) \) | 0 |
| \( FC_4(11,1) \) | 0 |
| \( FC_4(12,1) \) | 0 |
| \( FC_4(13,1) \) | \( t_{18} t_2/2 \) |
| \( FC_4(14,1) \) | \( t_{21} t_6/2 \) |
| \( FC_4(15,1) \) | \( t_{18} t_{10}/2 \) |
| \( FC_4(16,1) \) | \( t_{21} t_1/2 \) |
| \( FC_4(17,1) \) | 0 |
| \( FC_4(18,1) \) | 0 |
| \( FC_4(19,1) \) | 0 |
| \( FC_4(20,1) \) | 0 |
DESCRIPTION OF TERMS IN ELEMENTAL CONVECTION VECTOR $[F_{c,0}]$ FOR $\zeta = 1$

\[ \int_{-1}^{1} \int_{-1}^{1} \left[ N_{c}(\xi, \eta) \right]^T \mathcal{L}(\xi, \eta) \, d\xi \, d\eta \]

\begin{align*}
t1 &= 1 + \xi \\
t2 &= 1 + \eta \\
t6 &= 1 - \xi \\
t10 &= 1 - \eta \\
t17 &= \eta^{**2} \\
t18 &= 1 - t17 \\
t21 &= \xi^{**2} \\
t22 &= 1 - t21
\end{align*}
**ELEMENTAL CONVECTION VECTOR \{\text{FC5}\} FOR \( \zeta = 1 \)**

<table>
<thead>
<tr>
<th>( \text{FC5(1,1)} )</th>
<th>( = t_1^* t_2^* (\zeta + \eta - 1)/4 )</th>
</tr>
</thead>
<tbody>
<tr>
<td>( \text{FC5(2,1)} )</td>
<td>( = t_6^* t_2^* (-\zeta + \eta - 1)/4 )</td>
</tr>
<tr>
<td>( \text{FC5(3,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(4,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(5,1)} )</td>
<td>( = t_1^* t_{10}^* (\zeta - \eta - 1)/4 )</td>
</tr>
<tr>
<td>( \text{FC5(6,1)} )</td>
<td>( = t_6^* t_{10}^* (-\zeta - \eta - 1)/4 )</td>
</tr>
<tr>
<td>( \text{FC5(7,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(8,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(9,1)} )</td>
<td>( = t_{22}^* t_2^*/2 )</td>
</tr>
<tr>
<td>( \text{FC5(10,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(11,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(12,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(13,1)} )</td>
<td>( = t_{22}^* t_{10}^*/2 )</td>
</tr>
<tr>
<td>( \text{FC5(14,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(15,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(16,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(17,1)} )</td>
<td>( = t_{18}^* t_1^*/2 )</td>
</tr>
<tr>
<td>( \text{FC5(18,1)} )</td>
<td>( = t_{18}^* t_6^*/2 )</td>
</tr>
<tr>
<td>( \text{FC5(19,1)} )</td>
<td>( = 0 )</td>
</tr>
<tr>
<td>( \text{FC5(20,1)} )</td>
<td>( = 0 )</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN ELEMENTAL CONVECTION VECTOR $[F_\xi]$ FOR $\zeta = -1$

$$
\int \int \left[ N_{\eta}(\xi, \eta) \right]^T \left| J(\xi, \eta) \right| \, d\xi \, d\eta
$$

$t_1 = 1-\xi$
$t_2 = 1+\eta$
$t_6 = 1+\xi$
$t_{10} = 1-\eta$
$t_{17} = \eta^{**2}$
$t_{18} = 1-t_{17}$
$t_{21} = \xi^{**2}$
$t_{22} = 1-t_{21}$
ELEMENAL CONVECTION VECTOR (FC6) FOR $\zeta = -1$

<table>
<thead>
<tr>
<th>$\text{FC6}(i,j)$</th>
<th>Formula</th>
</tr>
</thead>
<tbody>
<tr>
<td>$\text{FC6}(1,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(2,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(3,1)$</td>
<td>$t1^<em>t2^</em>(-\xi+\eta-1)/4$</td>
</tr>
<tr>
<td>$\text{FC6}(4,1)$</td>
<td>$t6^<em>t2^</em>(\xi+\eta-1)/4$</td>
</tr>
<tr>
<td>$\text{FC6}(5,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(6,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(7,1)$</td>
<td>$t1^<em>t10^</em>(-\xi-\eta-1)/4$</td>
</tr>
<tr>
<td>$\text{FC6}(8,1)$</td>
<td>$t6^<em>t10^</em>(\xi-\eta-1)/4$</td>
</tr>
<tr>
<td>$\text{FC6}(9,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(10,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(11,1)$</td>
<td>$t22^*t2^2/2$</td>
</tr>
<tr>
<td>$\text{FC6}(12,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(13,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(14,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(15,1)$</td>
<td>$t22^*t10^2/2$</td>
</tr>
<tr>
<td>$\text{FC6}(16,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(17,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(18,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FC6}(19,1)$</td>
<td>$t18^*t1^2/2$</td>
</tr>
<tr>
<td>$\text{FC6}(20,1)$</td>
<td>$t18^*t6^2/2$</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN FIRST PART OF ELEMENTAL RADIATION VECTOR

$[E_i^a] \text{ FOR } \xi = 1$

\[ \int \int \left[ N_q(\eta, \xi) \right]^T | \mathbf{J}(\eta, \xi) | \, d\eta \, d\xi \]

\[ t_1 = 1+\eta \]
\[ t_2 = 1+\xi \]
\[ t_6 = 1-\xi \]
\[ t_{10} = 1-\eta \]
\[ t_{17} = \eta^{**2} \]
\[ t_{18} = 1-t_{17} \]
\[ t_{21} = \xi^{**2} \]
\[ t_{22} = 1-t_{21} \]
FIRST PART OF ELEMENTAL RADIATION VECTOR (FR1) FOR $\xi = 1$

<table>
<thead>
<tr>
<th>$\text{FR1}(n,1)$</th>
<th>Expression</th>
</tr>
</thead>
<tbody>
<tr>
<td>$\text{FR1}(1,1)$</td>
<td>$t1<em>t2</em>(-1+\eta+\xi)/4$</td>
</tr>
<tr>
<td>$\text{FR1}(2,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(3,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(4,1)$</td>
<td>$t1<em>t6</em>(-1+\eta-\xi)/4$</td>
</tr>
<tr>
<td>$\text{FR1}(5,1)$</td>
<td>$t10<em>t2</em>(-1-\eta+\xi)/4$</td>
</tr>
<tr>
<td>$\text{FR1}(6,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(7,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(8,1)$</td>
<td>$t10<em>t6</em>(-1-\eta-\xi)/4$</td>
</tr>
<tr>
<td>$\text{FR1}(9,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(10,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(11,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(12,1)$</td>
<td>$t22*t1/2$</td>
</tr>
<tr>
<td>$\text{FR1}(13,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(14,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(15,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(16,1)$</td>
<td>$t22*t10/2$</td>
</tr>
<tr>
<td>$\text{FR1}(17,1)$</td>
<td>$t18*t2/2$</td>
</tr>
<tr>
<td>$\text{FR1}(18,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(19,1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR1}(20,1)$</td>
<td>$t18*t6/2$</td>
</tr>
</tbody>
</table>

200
DESCRIPTION OF TERMS IN FIRST PART OF ELEMENTAL RADIATION VECTOR

\[ [F,^\eta] \] FOR \( \xi = -1 \)

\[
\int_{-1}^{1} \int_{-1}^{1} \left[ N^\eta(\eta, \zeta) \right]^T J(\eta, \zeta) \, d\eta \, d\zeta
\]

\( t_1 = 1 + \eta \)
\( t_2 = 1 + \zeta \)
\( t_6 = 1 - \zeta \)
\( t_{10} = 1 - \eta \)
\( t_{17} = \eta^2 \)
\( t_{18} = 1 - t_{17} \)
\( t_{21} = \zeta^2 \)
\( t_{22} = 1 - t_{21} \)
FIRST PART OF ELEMENTAL RADIATION VECTOR \( (FR2) \) FOR \( \xi = -1 \)

| \( FR2(1,1) \) | 0 |
| \( FR2(2,1) \) | \( t_1 t_2 (-1 + \eta + \zeta)/4 \) |
| \( FR2(3,1) \) | \( t_1 t_6 (-1 + \eta - \zeta)/4 \) |
| \( FR2(4,1) \) | 0 |
| \( FR2(5,1) \) | 0 |
| \( FR2(6,1) \) | \( t_{10} t_{12} (-1 - \eta + \zeta)/4 \) |
| \( FR2(7,1) \) | \( t_{10} t_{16} (-1 - \eta - \zeta)/4 \) |
| \( FR2(8,1) \) | 0 |
| \( FR2(9,1) \) | 0 |
| \( FR2(10,1) \) | \( t_{22} t_{11}/2 \) |
| \( FR2(11,1) \) | 0 |
| \( FR2(12,1) \) | 0 |
| \( FR2(13,1) \) | 0 |
| \( FR2(14,1) \) | \( t_{22} t_{10}/2 \) |
| \( FR2(15,1) \) | 0 |
| \( FR2(16,1) \) | 0 |
| \( FR2(17,1) \) | 0 |
| \( FR2(18,1) \) | \( t_{18} t_{12}/2 \) |
| \( FR2(19,1) \) | \( t_{18} t_{16}/2 \) |
| \( FR2(20,1) \) | 0 |
DESCRIPTION OF TERMS IN FIRST PART OF ELEMENTAL RADIATION VECTOR

\[ [F,\eta] \text{ FOR } \eta = 1 \]

\[
\int_{-1}^{1} \int_{-1}^{1} \left[ N^{T}(\xi, \zeta) \right] \left| J(\xi, \zeta) \right| d\xi \, d\zeta
\]

\[ t_1 = 1+\xi \]
\[ t_2 = 1+\xi \]
\[ t_6 = 1-\xi \]
\[ t_{10} = 1-\xi \]
\[ t_{17} = \xi^{*2} \]
\[ t_{18} = 1-t_{17} \]
\[ t_{20} = \xi^{*2} \]
\[ t_{21} = 1-t_{20} \]
FIRST PART OF ELEMENTAL RADIATION VECTOR (FR3) FOR $\eta = 1$

<table>
<thead>
<tr>
<th>$\text{FR3}(i,j)$</th>
<th>Equation</th>
</tr>
</thead>
<tbody>
<tr>
<td>$\text{FR3}(1,1)$</td>
<td>$t1<em>t2</em>(\xi-1+\zeta)/4$</td>
</tr>
<tr>
<td>$\text{FR3}(2,1)$</td>
<td>$t6<em>t2</em>(-\xi-1+\zeta)/4$</td>
</tr>
<tr>
<td>$\text{FR3}(3,1)$</td>
<td>$t6<em>t10</em>(-\xi-1-\zeta)/4$</td>
</tr>
<tr>
<td>$\text{FR3}(4,1)$</td>
<td>$t1<em>t10</em>(\xi-1-\zeta)/4$</td>
</tr>
<tr>
<td>$\text{FR3}(5,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(6,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(7,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(8,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(9,1)$</td>
<td>$t18*t2/2$</td>
</tr>
<tr>
<td>$\text{FR3}(10,1)$</td>
<td>$t21*t6/2$</td>
</tr>
<tr>
<td>$\text{FR3}(11,1)$</td>
<td>$t18*t10/2$</td>
</tr>
<tr>
<td>$\text{FR3}(12,1)$</td>
<td>$t21*t1/2$</td>
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<tr>
<td>$\text{FR3}(13,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(14,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(15,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(16,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(17,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(18,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(19,1)$</td>
<td>0</td>
</tr>
<tr>
<td>$\text{FR3}(20,1)$</td>
<td>0</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN FIRST PART OF ELEMENTAL RADIATION VECTOR

\[ [E_\bullet^\eta] \text{ FOR } \eta = -1 \]

\[
\int_{-1}^{1} \int_{-1}^{1} \left( \mathcal{N}(\xi, \zeta) \right)^T \left| J(\xi, \zeta) \right| d\xi \, d\zeta
\]

\begin{align*}
t1 &= 1 + \xi \\
t2 &= 1 + \zeta \\
t6 &= 1 - \xi \\
t10 &= 1 - \zeta \\
t17 &= \xi^* \xi \\
t18 &= 1 - t17 \\
t20 &= \zeta^* \zeta \\
t21 &= 1 - t20
\end{align*}
FIRST PART OF ELEMENTAL RADIATION VECTOR \{FR4\} FOR $\eta = -1$

<table>
<thead>
<tr>
<th>$FR4(1, 1)$</th>
<th>0</th>
</tr>
</thead>
<tbody>
<tr>
<td>$FR4(2, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(3, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(4, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(5, 1)$</td>
<td>$t1^<em>t2^</em>(\xi-1+\zeta)/4$</td>
</tr>
<tr>
<td>$FR4(6, 1)$</td>
<td>$16^<em>1^2^</em>(-\xi-1+\zeta)/4$</td>
</tr>
<tr>
<td>$FR4(7, 1)$</td>
<td>$t6^<em>10^</em>(-\xi-1-\zeta)/4$</td>
</tr>
<tr>
<td>$FR4(8, 1)$</td>
<td>$t1^<em>10^</em>(\xi-1-\zeta)/4$</td>
</tr>
<tr>
<td>$FR4(9, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(10, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(11, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(12, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(13, 1)$</td>
<td>$t18^*t2/2$</td>
</tr>
<tr>
<td>$FR4(14, 1)$</td>
<td>$t21^*t6/2$</td>
</tr>
<tr>
<td>$FR4(15, 1)$</td>
<td>$t18^*t10/2$</td>
</tr>
<tr>
<td>$FR4(16, 1)$</td>
<td>$t21^*t1/2$</td>
</tr>
<tr>
<td>$FR4(17, 1)$</td>
<td>0</td>
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<tr>
<td>$FR4(18, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(19, 1)$</td>
<td>0</td>
</tr>
<tr>
<td>$FR4(20, 1)$</td>
<td>0</td>
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</tbody>
</table>
DESCRIPTION OF TERMS IN FIRST PART OF ELEMENTAL RADIATION VECTOR

$[F_r^q]$ FOR $\zeta = 1$

$$
\int_1 \int_1 [N(\xi, \eta)]^T \cdot J(\xi, \eta) \cdot d\xi \cdot d\eta
$$

$t1 = 1 + \xi$

$t2 = 1 + \eta$

$t6 = 1 - \xi$

$t10 = 1 - \eta$

$t17 = \eta^2$

$t18 = 1 - t17$

$t21 = \xi^2$

$t22 = 1 - t21$
FIRST PART OF ELEMENTAL RADIATION VECTOR (FR5) FOR $\zeta = 1$

<table>
<thead>
<tr>
<th>Index (n, m)</th>
<th>Value</th>
</tr>
</thead>
<tbody>
<tr>
<td>FR5(1,1)</td>
<td>$t_1^* t_2^* (\xi + \eta - 1)/4$</td>
</tr>
<tr>
<td>FR5(2,1)</td>
<td>$t_6^* t_2^* (-\xi + \eta - 1)/4$</td>
</tr>
<tr>
<td>FR5(3,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(4,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(5,1)</td>
<td>$t_1^* t_10^* (\xi - \eta - 1)/4$</td>
</tr>
<tr>
<td>FR5(6,1)</td>
<td>$t_6^* t_10^* (-\xi - \eta - 1)/4$</td>
</tr>
<tr>
<td>FR5(7,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(8,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(9,1)</td>
<td>$t_{22}^* t_{2}/2$</td>
</tr>
<tr>
<td>FR5(10,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(11,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(12,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(13,1)</td>
<td>$t_{22}^* t_{10}/2$</td>
</tr>
<tr>
<td>FR5(14,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(15,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(16,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(17,1)</td>
<td>$t_{18}^* t_{1}/2$</td>
</tr>
<tr>
<td>FR5(18,1)</td>
<td>$t_{18}^* t_{6}/2$</td>
</tr>
<tr>
<td>FR5(19,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR5(20,1)</td>
<td>0</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN FIRST PART OF ELEMENTAL RADIATION VECTOR

$[F_{\varepsilon}^\circ]$ FOR $\zeta = -1$

\[
\int_{-1}^{1} \int_{-1}^{1} [N^\circ(\xi, \eta)]^T \mid J(\xi, \eta) \mid d\xi \, d\eta
\]

t1 = 1 - \xi

t2 = 1 + \eta

t6 = 1 + \xi

t10 = 1 - \eta

t17 = \eta^2

t18 = 1 - t17

t21 = \xi^2

t22 = 1 - t21
FIRST PART OF ELEMENTAL RADIATION VECTOR \( \{ Fr6 \} \) FOR \( \zeta = -1 \)

<table>
<thead>
<tr>
<th>( Fr6(i,1) )</th>
</tr>
</thead>
<tbody>
<tr>
<td>( Fr6(1,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(2,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(3,1) = t^{11} t^{2} (-\xi + \eta - 1)/4 )</td>
</tr>
<tr>
<td>( Fr6(4,1) = t^{6} t^{2} (\xi + \eta - 1)/4 )</td>
</tr>
<tr>
<td>( Fr6(5,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(6,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(7,1) = t^{1} t^{10} (-\xi - \eta - 1)/4 )</td>
</tr>
<tr>
<td>( Fr6(8,1) = t^{6} t^{10} (\xi - \eta - 1)/4 )</td>
</tr>
<tr>
<td>( Fr6(9,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(10,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(11,1) = t^{22} t^{2}/2 )</td>
</tr>
<tr>
<td>( Fr6(12,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(13,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(14,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(15,1) = t^{22} t^{10}/2 )</td>
</tr>
<tr>
<td>( Fr6(16,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(17,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(18,1) = 0 )</td>
</tr>
<tr>
<td>( Fr6(19,1) = t^{18} t^{1}/2 )</td>
</tr>
<tr>
<td>( Fr6(20,1) = t^{18} t^{6}/2 )</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL RADIATION

VECTOR $[F_{\xi}]$ FOR $\xi = 1 (C_1, C_4, C_5, C_8, C_{12}, C_{16}, C_{17}, C_{20}$ are the nodal

temperatures)

$$\int \int [N^o (\eta, \xi)]^T \left[ N^o (\eta, \xi) \{T^o (\eta, \xi)\}\right]^4 d\eta \, d\xi$$

$t_1 = 1 + \eta$
$t_2 = 1 + \zeta$
$t_5 = C_4 \eta$
$t_7 = C_{16} \eta$
$t_8 = \zeta^2$
$t_{10} = \eta^2$
$t_{11} = C_1 t_{10}$
$t_{13} = C_8 t_{10}$
$t_{15} = C_4 t_{10}$
$t_{17} = C_5 \eta$
$t_{19} = C_5 t_{10}$
$t_{21} = C_1 \eta$
$t_{29} = C_{17} t_{10}$
$t_{31} = -t_5 \zeta - 2 t_7 t_8 + t_{11} \zeta - t_{13} \zeta - t_{15} \zeta - t_{17} t_8 + t_{19} \zeta +$

$\# t_{21} t_8 + C_5 t_8 - 2 C_{16} t_8 + 2 t_7 + C_1 t_8 + t_{11} + C_4 t_8 + t_{15} + C_8 t_8 + t_{13} + 2 C_{17} \zeta$

$\# - 2 t_{29} - 2 C_{12} \zeta$
\[ t_{32} = C_{12} \cdot t_{10} \]
\[ t_{33} = C_{20} \cdot \eta \]
\[ t_{39} = C_{8} \cdot \eta \]
\[ t_{44} = -2 \cdot t_{32} - 2 \cdot t_{33} - 2 \cdot C_{20} \cdot t_{8} - 2 \cdot t_{29} \cdot \zeta + t_{5} \cdot t_{8} + 2 \cdot C_{17} + t_{19} - C_{8} + 2 \cdot t_{32} \cdot \xi \]
\[ a \cdot t_{17} \cdot \zeta + t_{39} \cdot \zeta - t_{39} \cdot t_{8} - C_{4} + 2 \cdot C_{12} + 2 \cdot C_{16} - C_{1} - C_{5} + t_{21} \cdot \zeta + 2 \cdot t_{33} \cdot t_{8} + 2 \cdot C_{20} \]
\[ a \cdot t_{46} = (t_{31} + t_{44})^2 \]
\[ t_{47} = t_{46}^2 \]
\[ t_{50} = -1 + \zeta \]
\[ t_{55} = -1 + \eta \]
\[ t_{64} = -1 + t_{10} \]
\[ t_{69} = -1 + t_{8} \]
SECOND PART OF ELEMENTAL RADIATION VECTOR (FR11) FOR $\zeta = 1$

<table>
<thead>
<tr>
<th>$\text{FR11}(1,1)$</th>
<th>$= t^1 t^2 (-1 + \eta + \zeta) t^{47/1024}$</th>
</tr>
</thead>
<tbody>
<tr>
<td>$\text{FR11}(2,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(3,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(4,1)$</td>
<td>$= t^1 t^50 (1 - \eta + \zeta) t^{47/1024}$</td>
</tr>
<tr>
<td>$\text{FR11}(5,1)$</td>
<td>$= -t^55 t^2 (-1 - \eta + \zeta) t^{47/1024}$</td>
</tr>
<tr>
<td>$\text{FR11}(6,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(7,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(8,1)$</td>
<td>$= -t^55 t^50 (1 + \eta + \zeta) t^{47/1024}$</td>
</tr>
<tr>
<td>$\text{FR11}(9,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(10,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(11,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(12,1)$</td>
<td>$= t^64 t^50 t^{47/512}$</td>
</tr>
<tr>
<td>$\text{FR11}(13,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(14,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(15,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(16,1)$</td>
<td>$= -t^69 t^1 t^{47/512}$</td>
</tr>
<tr>
<td>$\text{FR11}(17,1)$</td>
<td>$= -t^64 t^2 t^{47/512}$</td>
</tr>
<tr>
<td>$\text{FR11}(18,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(19,1)$</td>
<td>$= 0$</td>
</tr>
<tr>
<td>$\text{FR11}(20,1)$</td>
<td>$= t^69 t^55 t^{47/512}$</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL RADIATION
VECTOR $[F_{ij}]$ FOR $\xi = -1$ (C2, C3, C6, C7, C10, C14, C18, C19 are the nodal temperatures)

$$
\int_{-1}^{1} \int_{-1}^{1} \left[ N^i (\eta, \zeta) \right]^T \left( \left[ N^o (\eta, \zeta) \right] \left[ T^o (\eta, \zeta) \right] \right) \, d\eta \, d\zeta
$$

$\eta = 1 + \eta$
$\zeta = 1 + \zeta$
$t1 = C3*\eta$
$t2 = C2*\eta$
$t3 = \eta**2$
$t4 = C3*t9$
$t5 = C2*t9$
$t6 = \zeta**2$
$t7 = C3*t9$
$t8 = C14*\eta$
$t9 = C10*t9$
$t10 = C6*t9$
$t11 = C6*\eta$
$t12 = C7*\eta$
$t13 = t5*\zeta+t7*\zeta+t10*\zeta+t5*t12+t14*\zeta+t7*t12-2*t17*t12-2$
$t14 = t19*\zeta+t21*\zeta+2*C14-C3-C6+2*C18+2*C19-C2-2*C18*t12-t24*\zeta-t$
$t15 = t26*\zeta-C7$

214
\begin{align*}
t_{30} &= C_{19} t_9 \\
t_{35} &= C_7 t_9 \\
t_{39} &= C_{18} \eta \\
t_{44} &= -2 t_{30} + C_2 t_{12} + t_{10} + C_3 t_{12} + t_{14} + C_6 t_{12} + t_{21} + C_7 t_{12} + t_{35} + 2 C_{10} \zeta \\
\#a &= -2 t_{19} - 2 C_{19} \zeta + 2 t_{17} - 2 C_{14} t_{12} - 2 t_{39} + 2 C_{10} + 2 t_{39} t_{12} + t_{24} t_{12} \\
\#35 &= 2 t_{30} \zeta \\
t_{46} &= (t_{29} + t_{44})^2 \\
t_{47} &= t_{46}^2 \\
t_{50} &= -1 + \zeta \\
t_{55} &= -1 + \eta \\
t_{64} &= -1 + t_9 \\
t_{69} &= -1 + t_{12}
\end{align*}
SECOND PART OF ELEMENTAL RADIATION VECTOR (FR22) FOR $\zeta = -1$

<table>
<thead>
<tr>
<th>$n$</th>
<th>$\Phi_{22}(n,1)$</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>0</td>
</tr>
<tr>
<td>2</td>
<td>$t_1 t_2^{<em>} (-1 + \eta + \zeta)^{</em>} t_{47}/1024$</td>
</tr>
<tr>
<td>3</td>
<td>$t_1 t_{50}^{<em>} (1 - \eta + \zeta)^{</em>} t_{47}/1024$</td>
</tr>
<tr>
<td>4</td>
<td>0</td>
</tr>
<tr>
<td>5</td>
<td>0</td>
</tr>
<tr>
<td>6</td>
<td>$-t_5 t_{12}^{<em>} (-1 - \eta - \zeta)^{</em>} t_{47}/1024$</td>
</tr>
<tr>
<td>7</td>
<td>$-t_5 t_{50}^{<em>} (1 + \eta - \zeta)^{</em>} t_{47}/1024$</td>
</tr>
<tr>
<td>8</td>
<td>0</td>
</tr>
<tr>
<td>9</td>
<td>0</td>
</tr>
<tr>
<td>10</td>
<td>$-t_6 t_{12}^{*} t_{47}/512$</td>
</tr>
<tr>
<td>11</td>
<td>0</td>
</tr>
<tr>
<td>12</td>
<td>0</td>
</tr>
<tr>
<td>13</td>
<td>0</td>
</tr>
<tr>
<td>14</td>
<td>$-t_6 t_{1}^{*} t_{47}/512$</td>
</tr>
<tr>
<td>15</td>
<td>0</td>
</tr>
<tr>
<td>16</td>
<td>0</td>
</tr>
<tr>
<td>17</td>
<td>0</td>
</tr>
<tr>
<td>18</td>
<td>$t_6 t_{55}^{*} t_{47}/512$</td>
</tr>
<tr>
<td>19</td>
<td>$t_6 t_{50}^{*} t_{47}/512$</td>
</tr>
<tr>
<td>20</td>
<td>0</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL RADIATION

VECTOR \([F_i^\eta]\) FOR \(\eta = 1\) (C1, C2, C3, C4, C9, C10, C11, C12 are the nodal temperatures)

\[
\int_{-1}^{1} \int_{-1}^{1} \left( \begin{bmatrix} N_i (\xi, \zeta) \end{bmatrix}^T \left( \begin{bmatrix} N_i (\xi, \zeta) \end{bmatrix} \right) \right) \, d\xi \, d\zeta
\]

\(t_1 = 1 + \xi\)
\(t_2 = 1 + \zeta\)
\(t_5 = \xi^{**2}\)
\(t_6 = C4 \cdot t_5\)
\(t_8 = C1 \cdot \xi\)
\(t_9 = \zeta^{**2}\)
\(t_{11} = C4 \cdot \xi\)
\(t_{14} = C9 \cdot t_5\)
\(t_{16} = C12 \cdot \xi\)
\(t_{18} = C2 \cdot t_5\)
\(t_{20} = C2 \cdot \zeta\)
\(t_{22} = C11 \cdot t_5\)
\(t_{24} = C1 \cdot t_5\)
\(t_{27} = C10 \cdot \xi\)
\(t_{30} = -16 \cdot \xi + 18 \cdot t_9 - 111 \cdot \xi + 18 \cdot \zeta - 2 \cdot t_{14} \cdot \zeta - 2 \cdot t_{16} \cdot t_9 + t_{18} \cdot \zeta\)

#t_{20} \cdot t_9 + 2 \cdot t_{22} \cdot \zeta + t_{24} \cdot \zeta - t_{20} \cdot \zeta + 2 \cdot C10 - C2 - C3 - C4 + 2 \cdot t_{27} \cdot t_9 - C1 + 2\]

217
#C11+2*C12+t11*t9

t35 = C3*t5

t40 = C3*ξ

t44 = -2*C12*t9+C1*t9+t24+C2*t9+t18+C3*t9+t35+C4*t9+t6+2*C9*ζ-

#2*t14-2*t27-2*C10*t9-2*C11*ζ-2*t22+2*t16+2*C9+t40*ζ-t35*ζt

#a-t40*t9

t46 = (t30+t44)**2

t47 = t46**2

t50 = -1+ξ

t55 = -1+ξ

t64 = -1+t5

t67 = -1+t9
SECOND PART OF ELEMENTAL RADIATION VECTOR \{FR33\} FOR $\eta = 1$

<table>
<thead>
<tr>
<th>$\text{FR33}(1, 1)$</th>
<th>$t1^<em>t2^</em>(\xi - 1 + \zeta)^*t47/1024$</th>
</tr>
</thead>
<tbody>
<tr>
<td>$\text{FR33}(2, 1)$</td>
<td>$-t50^<em>t2^</em>(-\xi - 1 + \zeta)^*t47/1024$</td>
</tr>
<tr>
<td>$\text{FR33}(3, 1)$</td>
<td>$-t50^<em>t55^</em>(\xi + 1 + \zeta)^*t47/1024$</td>
</tr>
<tr>
<td>$\text{FR33}(4, 1)$</td>
<td>$t1^<em>t55^</em>(-\xi + 1 + \zeta)^*t47/1024$</td>
</tr>
<tr>
<td>$\text{FR33}(5, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(6, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(7, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(8, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(9, 1)$</td>
<td>$-164^*t2^*t47/512$</td>
</tr>
<tr>
<td>$\text{FR33}(10, 1)$</td>
<td>$t67^*t50^*t47/512$</td>
</tr>
<tr>
<td>$\text{FR33}(11, 1)$</td>
<td>$t64^*i55^*t47/512$</td>
</tr>
<tr>
<td>$\text{FR33}(12, 1)$</td>
<td>$-167^*t1^*t47/512$</td>
</tr>
<tr>
<td>$\text{FR33}(13, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(14, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(15, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(16, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(17, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(18, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(19, 1)$</td>
<td>$0$</td>
</tr>
<tr>
<td>$\text{FR33}(20, 1)$</td>
<td>$0$</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL RADIATION

VECTOR \([F_r^o]\) FOR \(\eta = -1\) (C5, C6, C7, C8, C13, C14, C15, C16 are the nodal temperatures)

\[
\int \int [N^o(\eta, \zeta)]^T \left( [N^o(\eta, \zeta)] \{T^o(\eta, \zeta)\} \right)^{1/2} \, d\eta \, d\zeta
\]

\[t_{11} = 1 + \xi\]
\[t_{12} = 1 + \zeta\]
\[t_{15} = C7 * \xi\]
\[t_{16} = \xi^{**2}\]
\[t_{18} = C8 * \xi\]
\[t_{10} = C16 * \xi\]
\[t_{12} = C6 * \xi\]
\[t_{14} = \xi^{**2}\]
\[t_{15} = C5 * t_{14}\]
\[t_{18} = C6 * t_{14}\]
\[t_{20} = C7 * t_{14}\]
\[t_{22} = C8 * t_{14}\]
\[t_{24} = C13 * t_{14}\]
\[t_{26} = C15 * t_{14}\]
\[t_{27} = -t_{15} * t_{6} - t_{18} * \xi - C7 - 2 * t_{10} * t_{6} - t_{12} * t_{6} + 2 * C16 - C6 + t_{15} + C5 * t_{6} + C6 * t_{6} + t\]

#18 + C7 * t_{6} + t_{20} + C8 * t_{6} + t_{22} + 2 * C13 * \xi - 2 * t_{24} - 2 * C14 * t_{6} - 2 * t_{26} + 2 * t_{10}

220
t31 = C5*ξ

t38 = C14*ξ

t44 = -2*C16*t6+t5*ξ-t12*ξ+t31*ξ+t15*ξ+t18*ξ-t20*ξ

#t22*ξ-2*t24*ξ+2*t38*t6+2*t26*ξ+t8*t6-2*C15*ξ+2*C14+2*C15-2*t38-C8+2*C13+t31*t6-C5

#t46 = (t27+t44)**2

t47 = t46**2

#t50 = -1+ξ

#t55 = -1+ξ

#t64 = -1+t14

#t67 = -1+t6
| \( \text{FR44}(1,1) \) | 0 |
| \( \text{FR44}(2,1) \) | 0 |
| \( \text{FR44}(3,1) \) | 0 |
| \( \text{FR44}(4,1) \) | 0 |
| \( \text{FR44}(5,1) \) | \( t_1 t_2 (\xi - 1 + \zeta) * t^{47/1024} \) |
| \( \text{FR44}(6,1) \) | \( -t_50 t_2 (\xi - 1 + \zeta) * t^{47/1024} \) |
| \( \text{FR44}(7,1) \) | \( -t_50 t_55 (\xi + 1 + \zeta) * t^{47/1024} \) |
| \( \text{FR44}(8,1) \) | \( t_1 t_55 (\xi + 1 + \zeta) * t^{47/1024} \) |
| \( \text{FR44}(9,1) \) | 0 |
| \( \text{FR44}(10,1) \) | 0 |
| \( \text{FR44}(11,1) \) | 0 |
| \( \text{FR44}(12,1) \) | 0 |
| \( \text{FR44}(13,1) \) | \( -t_64 t_2 t^{47/512} \) |
| \( \text{FR44}(14,1) \) | \( t_67 t_50 t^{47/512} \) |
| \( \text{FR44}(15,1) \) | \( t_64 t_55 t^{47/512} \) |
| \( \text{FR44}(16,1) \) | \( -t_67 t_1 t^{47/512} \) |
| \( \text{FR44}(17,1) \) | 0 |
| \( \text{FR44}(18,1) \) | 0 |
| \( \text{FR44}(19,1) \) | 0 |
| \( \text{FR44}(20,1) \) | 0 |
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL RADIATION

VECTOR $[F_p]$ FOR $\zeta = 1$ (C1, C2, C5, C6, C9, C13, C17, C18 are the nodal temperatures)

$$
\int \int \left[ N^e (\xi, \eta) \right]^T \left[ \left[ N^e (\xi, \eta) \right] \left[ T^e (\xi, \eta) \right] \right] d\xi \ d\eta
$$

$t1 = 1 + \xi$
$t2 = 1 + \eta$
$t5 = \xi^{**2}$
$t6 = C9*t5$
$t8 = C17*\xi$
$t9 = \eta^{**2}$
$t11 = C2*\xi$
$t14 = C5*t5$
$t16 = C1*\xi$
$t19 = C5*\xi$
$t21 = C6*t5$
$t25 = C1*t5$
$t27 = C2*t5$
$t30 = -2*t6*\eta-2*t8*t9-t11*t9-t11*\eta-t14*\eta+t16*\eta+t16*t9-t19*\eta-t21*\eta-2*C17*t9+2*C17+C1*t9+t25+C2*t9+t27+C5*t9+t14+C6*t9+t21*2$

#*18

223
\[ t_{34} = C_{13} \xi \]
\[ t_{36} = C_{6} \xi \]
\[ t_{40} = C_{18} \xi \]
\[ t_{44} = -2C_{18}t_{19} + 2C_{9} \eta - 2t_{6} - 2C_{13} \eta - 2t_{34} + t_{25} \eta + t_{36} \eta + 2t \]
\[ +34 \eta \cdot C_{1} + t_{36} t_{19} - 2t_{40} C_{6} + 2C_{18} + 2C_{13} + 2C_{13} - C_{2} - C_{5} + 2t_{40} t_{19} + t_{27} \eta \]
\[ + t_{19} \eta \]
\[ t_{46} = (t_{30} + t_{44})^2 \]
\[ t_{47} = t_{46}^2 \]
\[ t_{50} = -1 + \xi \]
\[ t_{55} = -1 + \eta \]
\[ t_{64} = -1 + t_{9} \]
\[ t_{69} = -1 + t_{5} \]
SECOND PART OF ELEMENTAL RADIATION VECTOR \{FR55\} FOR \( \zeta = 1 \)

<table>
<thead>
<tr>
<th></th>
<th>Formula</th>
</tr>
</thead>
<tbody>
<tr>
<td>FR55(1,1)</td>
<td>( t_1 \cdot t_2 \cdot (\xi + \eta - 1) \cdot \frac{47}{1024} )</td>
</tr>
<tr>
<td>FR55(2,1)</td>
<td>( t_50 \cdot t_2 \cdot (\xi - \eta + 1) \cdot \frac{47}{1024} )</td>
</tr>
<tr>
<td>FR55(3,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(4,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(5,1)</td>
<td>( -t_1 \cdot t_55 \cdot (\xi - \eta - 1) \cdot \frac{47}{1024} )</td>
</tr>
<tr>
<td>FR55(6,1)</td>
<td>( -t_50 \cdot t_55 \cdot (\xi + \eta + 1) \cdot \frac{47}{1024} )</td>
</tr>
<tr>
<td>FR55(7,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(8,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(9,1)</td>
<td>( -t_69 \cdot t_2 \cdot \frac{47}{512} )</td>
</tr>
<tr>
<td>FR55(10,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(11,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(12,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(13,1)</td>
<td>( 169 \cdot t_55 \cdot \frac{47}{512} )</td>
</tr>
<tr>
<td>FR55(14,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(15,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(16,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(17,1)</td>
<td>( -t_64 \cdot t_1 \cdot \frac{47}{512} )</td>
</tr>
<tr>
<td>FR55(18,1)</td>
<td>( t_64 \cdot t_50 \cdot \frac{47}{512} )</td>
</tr>
<tr>
<td>FR55(19,1)</td>
<td>0</td>
</tr>
<tr>
<td>FR55(20,1)</td>
<td>0</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN SECOND PART OF ELEMENTAL RADIATION

VECTOR $[F_\nu]$ FOR $\xi = -1 (C_3, C_4, C_7, C_8, C_{11}, C_{15}, C_{19}, C_{20}$ are the nodal

temperatures)

$$
\int \int \left[ N^\nu (\xi, \eta) \right]^\nu \left[ (N^\nu (\xi, \eta)) \{ T^\nu (\xi, \eta) \} \right] d\xi d\eta
$$

$t_1 = -1 + \xi$
$t_2 = 1 + \eta$
$t_5 = C_8 \xi$
$t_7 = \xi^2$
$t_8 = C_3 \xi$
$t_{10} = C_4 \xi$
$t_{12} = \eta^2$
$t_{15} = C_{11} \xi$
$t_{18} = C_4 \xi$
$t_{19} = C_7 \xi$
$t_{21} = 5 \xi \eta - 8 \eta - t_{10} \eta - C_7 \xi t_{13} - C_4 \xi t_{12} + C_8 - 2 \xi C_{19} + 2 \xi t_{15} - 2 \xi C_{20} - 2 \xi$

#1 - 2 \xi C_{15} + C_3 + C_7 + C_4 + 2 \xi t_{15} \eta - C_3 \xi t_{12} - t_{10} - t_{18} - t_{19} - C_8 \xi t_{12}$

$t_{22} = C_8 \xi$
$t_{23} = C_{19} \xi$
$t_{25} = C_{20} \xi$
$t_{29} = C_{15} \xi$
\[ t_{30} = C_7 \xi \]
\[ t_{34} = C_3 \xi \]
\[ t_{44} = -t_{22} + 2 \cdot t_{23} + 2 \cdot C_{19} \cdot t_{12} - 2 \cdot t_{25} + 2 \cdot C_{20} \cdot t_{12} - 2 \cdot C_{11} \cdot \eta + 2 \cdot C_{15} \cdot \eta + 2 \cdot t \]
\[ #29-t_{30} \cdot \eta - t_{18} \cdot \eta - t_{10} \cdot t_{12} + t_{34} \cdot \eta - 2 \cdot t_{23} \cdot t_{12} + 2 \cdot t_{25} \cdot t_{12} - 2 \cdot t_{29} \cdot \eta + t \]
\[ #19 \cdot \eta + t_{30} \cdot t_{12} + t_{22} \cdot \eta - t_5 \cdot t_{12} + t_{34} \cdot t_{12} \]
\[ t_{46} = (t_{21} + t_{44})^2 \]
\[ t_{47} = t_{46}^2 \]
\[ t_{50} = 1 + \xi \]
\[ t_{55} = -1 + \eta \]
\[ t_{64} = -1 + t_{12} \]
\[ t_{69} = -1 + t_7 \]
SECOND PART OF ELEMENTAL RADIATION VECTOR \( \{ \text{FR66} \} \) FOR \( \zeta = -1 \)

| \( \text{FR66}(1,1) \) | 0 |
| \( \text{FR66}(2,1) \) | 0 |
| \( \text{FR66}(3,1) \) | \( t1^* t2^* (\zeta - \eta + 1)^* t47/1024 \) |
| \( \text{FR66}(4,1) \) | \( t50^* t2^* (\zeta + \eta - 1)^* t47/1024 \) |
| \( \text{FR66}(5,1) \) | 0 |
| \( \text{FR66}(6,1) \) | 0 |
| \( \text{FR66}(7,1) \) | \( -t1^* t55^* (\zeta + \eta + 1)^* t47/1024 \) |
| \( \text{FR66}(8,1) \) | \( -t50^* t55^* (\zeta - \eta - 1)^* t47/1024 \) |
| \( \text{FR66}(9,1) \) | 0 |
| \( \text{FR66}(10,1) \) | 0 |
| \( \text{FR66}(11,1) \) | \( -t69^* t2^* t47/512 \) |
| \( \text{FR66}(12,1) \) | 0 |
| \( \text{FR66}(13,1) \) | 0 |
| \( \text{FR66}(14,1) \) | 0 |
| \( \text{FR66}(15,1) \) | \( t69^* t55^* t47/512 \) |
| \( \text{FR66}(16,1) \) | 0 |
| \( \text{FR66}(17,1) \) | 0 |
| \( \text{FR66}(18,1) \) | 0 |
| \( \text{FR66}(19,1) \) | \( t64^* t1^* t47/512 \) |
| \( \text{FR66}(20,1) \) | \( -t64^* t50^* t47/512 \) |
DESCRIPTION OF TERMS IN MATRIX $[JJ1] \ [3 \times 20]$ WHICH WHEN MULTIPLIED

BY COORDINATE MATRIX $[20 \times 3]$ GIVES JACOBIAN MATRIX $[J]$

$$
\begin{align*}
  t_1 &= 1 + \eta \\
  t_2 &= 1 + \zeta \\
  t_3 &= t_1 t_2 \\
  t_4 &= \xi + \eta + \zeta - 2 \\
  t_6 &= 1 + \xi \\
  t_7 &= t_6 t_1 \\
  t_8 &= t_7 t_2 \\
  t_{10} &= -\xi + \eta + \zeta - 2 \\
  t_{12} &= 1 - \xi \\
  t_{13} &= t_{12} t_1 \\
  t_{14} &= t_{13} t_2 \\
  t_{16} &= 1 - \zeta \\
  t_{17} &= t_1 t_{16} \\
  t_{18} &= -\xi + \eta - \zeta - 2 \\
  t_{20} &= t_{13} t_{16} \\
  t_{22} &= \xi + \eta - \zeta - 2 \\
  t_{24} &= t_7 t_{16} \\
  t_{26} &= 1 - \eta \\
  t_{27} &= t_{26} t_2
\end{align*}
$$

\[ t_{28} = \xi \eta + \zeta - 2 \]
\[ t_{30} = t_6 \cdot t_{26} \]
\[ t_{31} = t_{30} \cdot t_{12} \]
\[ t_{33} = -\xi \eta + \zeta - 2 \]
\[ t_{35} = t_{12} \cdot t_{26} \]
\[ t_{36} = t_{35} \cdot t_{12} \]
\[ t_{38} = t_{26} \cdot t_{16} \]
\[ t_{39} = -\xi \eta - \zeta - 2 \]
\[ t_{41} = t_{35} \cdot t_{16} \]
\[ t_{43} = \xi \eta - \zeta - 2 \]
\[ t_{45} = t_{30} \cdot t_{16} \]
\[ t_{47} = \eta ^{**} 2 \]
\[ t_{48} = 1 - t_{47} \]
\[ t_{49} = t_{48} \cdot t_{12} \]
\[ t_{50} = t_{48} \cdot t_{16} \]
\[ t_{51} = \xi \cdot t_{11} \]
\[ t_{53} = \zeta ^{**} 2 \]
\[ t_{54} = 1 - t_{53} \]
\[ t_{55} = t_{54} \cdot t_{11} \]
\[ t_{57} = \xi \cdot t_{26} \]
\[ t_{59} = t_{54} \cdot t_{26} \]
\[ t_{61} = t_6 \cdot t_{12} \]
164 = t12 * t2
167 = t12 * t16
170 = t16 * t16
181 = η * t6
183 = η * t12
187 = ζ * 2
188 = 1 - t87
189 = t88 * t2
190 = t54 * t12
191 = t88 * t16
192 = t54 * t16
199 = t48 * t16
110 = t48 * t12
111 = t88 * t11
112 = ζ * t12
114 = ζ * t6
116 = t88 * t26
**MATRIX [J J1] [3 x 20] WHICH WHEN MULTIPLIED BY COORDINATE**

**MATRIX [20 x 3] GIVES JACOBIAN MATRIX [J]**

\[
\begin{array}{cccccccccccccccccccc}
134.56 & 234.56 & 345.67 & 456.78 & 567.89 & 678.90 & 789.01 & 890.12 & 901.23 & 123.45 & 456.78 & 789.01 & 012.34 & 345.67 & 678.90 & 901.23 & 234.56
\end{array}
\]
DESCRIPTION OF TERMS IN MATRIX [JJ2] [2 x 20] WHICH WHEN MULTIPLIED BY COORDINATE MATRIX [20 x 2] GIVES JACOBIAN MATRIX [J] FOR $\xi = 1$

\[ t1 = 1 + \zeta \]
\[ t2 = -1 + \eta + \zeta \]
\[ t4 = 1 + \eta \]
\[ t5 = t4 \cdot t1 \]
\[ t7 = 1 - \zeta \]
\[ t8 = -1 + \eta - \zeta \]
\[ t10 = t4 \cdot t7 \]
\[ t12 = -1 - \eta + \zeta \]
\[ t14 = 1 - \eta \]
\[ t15 = t14 \cdot t1 \]
\[ t17 = -1 + \eta - \zeta \]
\[ t19 = t14 \cdot t7 \]
\[ t23 = \xi^2 \]
\[ t34 = \eta^2 \]
MATRIX [JJ2] [2 x 20] WHICH WHEN MULTIPLIED BY COORDINATE

MATRIX [20 x 2] GIVES JACOBIAN MATRIX [J] FOR $\xi = 1$

\[
\begin{bmatrix}
11^{12}/4+15/4 & 0 & 0 & 17^{12}/8+110/4 & 0 & 0 & 0 & 0 & 1.0/2.0+123/2 & 0 & 0 & -1.0/2.0+123/2 & -\eta^4/16 & 0 & 0 & -\eta^4/16
\end{bmatrix}
\]
DESCRIPTION OF TERMS IN MATRIX \([JJ3] \ [2 \times 2]\) WHICH WHEN MULTIPLIED BY COORDINATE MATRIX \([20 \times 2]\) GIVES JACOBIAN MATRIX \([J]\) FOR \(\zeta = -1\)

\[
\begin{align*}
t1 &= 1 + \zeta \\
t2 &= -1 + \eta + \zeta \\
t4 &= 1 + \eta \\
t5 &= t4 \cdot t1 \\
t7 &= 1 - \zeta \\
t8 &= -1 + \eta - \zeta \\
t10 &= t4 \cdot t7 \\
t12 &= -1 - \eta + \zeta \\
t14 &= 1 - \eta \\
t15 &= t14 \cdot t1 \\
t17 &= -1 - \eta - \zeta \\
t19 &= t14 \cdot t7 \\
t23 &= \zeta \cdot 2 \\
t34 &= \eta \cdot 2
\end{align*}
\]
MATRIX [JJ3] [2 x 20] WHICH WHEN MULTIPLIED BY COORDINATE

MATRIX [20 x 2] GIVES JACOBIAN MATRIX [J] FOR $\xi = -1$

<table>
<thead>
<tr>
<th>0</th>
<th>$t_{12}/4+t_{15}/4$</th>
<th>$t_{12}/4+t_{15}/4$</th>
<th>0</th>
<th>0</th>
<th>$-1^2t_{12}/4+t_{15}/4$</th>
<th>$-1^2t_{17}/4+t_{19}/4$</th>
<th>0</th>
<th>0</th>
<th>1.E0/2.E0-t23/2</th>
<th>0</th>
<th>0</th>
<th>0</th>
<th>0</th>
<th>0</th>
<th>$-\eta_1^*\xi$</th>
<th>$-\eta_7^*\xi$</th>
<th>0</th>
</tr>
</thead>
<tbody>
<tr>
<td>0</td>
<td>$t_{12}/4+t_{15}/4$</td>
<td>$-t_{12}/4-t_{15}/4$</td>
<td>0</td>
<td>0</td>
<td>$-t_{12}/4-t_{15}/4$</td>
<td>$-t_{17}/4-t_{19}/4$</td>
<td>0</td>
<td>0</td>
<td>1.E0/2.E0+t23/2</td>
<td>0</td>
<td>0</td>
<td>0</td>
<td>0</td>
<td>0</td>
<td>$-\eta_1^*\xi$</td>
<td>$-\eta_7^*\xi$</td>
<td>0</td>
</tr>
</tbody>
</table>
DESCRIPTION OF TERMS IN MATRIX [JJ4] [2 × 20] WHICH WHEN MULTIPLIED
BY COORDINATE MATRIX [20 × 2] GIVES JACOBIAN MATRIX [J] FOR η = 1

\begin{align*}
    t_1 &= 1 + \zeta \\
    t_2 &= \xi - 1 + \zeta \\
    t_4 &= 1 + \xi \\
    t_5 &= t_4^* t_1 \\
    t_7 &= -\xi - 1 + \zeta \\
    t_9 &= 1 - \xi \\
    t_{10} &= t_9^* t_1 \\
    t_{12} &= 1 - \zeta \\
    t_{13} &= -\xi - 1 - \zeta \\
    t_{15} &= t_9^* t_{12} \\
    t_{17} &= \xi - 1 - \zeta \\
    t_{19} &= t_4^* t_{12} \\
    t_{22} &= \zeta^{**2} \\
    t_{34} &= \xi^{**2}
\end{align*}
MATRIX \([JJ4] [2 \times 20]\) WHICH WHEN MULTIPLIED BY COORDINATE MATRICE \([20 \times 2]\) GIVES JACOBIAN MATRIX \([J]\) FOR \(\eta = 1\)

\[
\begin{bmatrix}
1.12 \times 10^{-4} & -1.7 \times 10^{-4} & 1.2 \times 10^{-4} & -1.0 \times 10^{-4} & 0 & 0 & 0 & -x'11 & -1.E0/2.E0+122/2 & -x'112 & 1.E0/2.E0+422/2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-1.12 \times 10^{-4} & 1.7 \times 10^{-4} & -1.2 \times 10^{-4} & 1.0 \times 10^{-4} & 0 & 0 & 0 & -zeta'19 & -1.E0/2.E0+134/2 & -zeta'14 & 1.E0/2.E0+434/2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{bmatrix}
\]
DESCRIPTION OF TERMS IN MATRIX [JJ5] [2 x 20] WHICH WHEN MULTIPLIED
BY COORDINATE MATRIX [20 x 2] GIVES JACOBIAN MATRIX [J] FOR \( \eta = -1 \)

\[
\begin{align*}
t_1 &= 1 + \zeta \\
t_2 &= \xi - 1 + \zeta \\
t_4 &= 1 + \xi \\
t_5 &= t_4 * t_1 \\
t_7 &= -\xi - 1 + \zeta \\
t_9 &= 1 - \xi \\
t_{10} &= t_9 * t_1 \\
t_{12} &= 1 - \xi \\
t_{13} &= -\xi - 1 - \xi \\
t_{15} &= t_9 * t_{12} \\
t_{17} &= \xi - 1 - \xi \\
t_{19} &= t_4 * t_{12} \\
t_{22} &= \xi**2 \\
t_{34} &= \xi**2
\end{align*}
\]
MATRIX $[JJ_5] [2 \times 20]$ WHICH WHEN MULTIPLIED BY COORDINATE

MATRIX $[20 \times 2]$ GIVES JACOBIAN MATRIX $[J]$ FOR $\eta = -1$

\[
\begin{array}{cccccccccccccccc}
0 & 0 & 0 & 0 & 0 & 1 & -1/2 & 1/4 & 1/4 & -1/2 & 1/4 & 1/4 & 0 & 0 & 0 & -x'_{11} & -1.E0/2.E0+t2/2 & -x'_{12} & 1.E0/2.E0+t2/2 & 0 & 0 & 0 \\
0 & 0 & 0 & 0 & 1 & -1/2 & 1/4 & 1/4 & -1/2 & 1/4 & 1/4 & 0 & 0 & 0 & 1.E0/2.E0+t3/2 & -zeta't9 & -1.E0/2.E0+t3/2 & -zeta't4 & 0 & 0 & 0 \\
\end{array}
\]
DESCRIPTION OF TERMS IN MATRIX [J J6] [2 × 20] WHICH WHEN MULTIPLIED
BY COORDINATE MATRIX [20 × 2] GIVES JACOBIAN MATRIX [J] FOR ζ = 1

t1 = 1 + η

t2 = ζ + η - 1

t4 = 1 + ξ

t5 = t4 * t1

t7 = -ξ + η - 1

t9 = 1 - ξ
t10 = t9 * t1
t12 = 1 - η
t13 = ξ - η - 1
t15 = t4 * t12
t17 = -ξ - η - 1
t19 = t9 * t12
t21 = η ** 2
t36 = ξ ** 2
MATRIX $[JJ6] \ [2 \times 20]$ WHICH WHEN MULTIPLIED BY COORDINATE

MATRIX $[20 \times 2]$ GIVES JACOBIAN MATRIX $[J]$ FOR $\zeta = 1$

<table>
<thead>
<tr>
<th>$t^*t_{12}/4 + 15/4$</th>
<th>$-t^*t_{17}/4 + 10/4$</th>
<th>0</th>
<th>0</th>
<th>$t^*t_{13}/4 + 15/4$</th>
<th>$-t^*t_{17}/4 + 19/4$</th>
<th>0</th>
<th>0</th>
<th>-$x^*t_{11}$</th>
<th>0</th>
<th>0</th>
<th>-$x^*t_{12}$</th>
<th>0</th>
<th>0</th>
<th>1.0</th>
<th>0</th>
<th>0</th>
<th>0</th>
<th>$-1.0/2.0 + 21/2$</th>
<th>0</th>
<th>0</th>
</tr>
</thead>
<tbody>
<tr>
<td>$t^*t_{12}/4 + 15/4$</td>
<td>$t^*t_{17}/4 + 10/4$</td>
<td>0</td>
<td>0</td>
<td>$-t^*t_{13}/4 + 15/4$</td>
<td>$-t^*t_{17}/4 + 19/4$</td>
<td>0</td>
<td>0</td>
<td>1.0</td>
<td>0</td>
<td>0</td>
<td>1.0</td>
<td>2.0</td>
<td>0</td>
<td>0</td>
<td>$-1.0/2.0 + 13/2$</td>
<td>0</td>
<td>0</td>
<td>0</td>
<td>$-0.0$</td>
<td>0</td>
</tr>
</tbody>
</table>

\[ \begin{align*}
&1.0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 1.0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0
\end{align*} \]
DESCRIPTION OF TERMS IN MATRIX $[J] J_{7}$ $[2 \times 20]$ WHICH WHEN MULTIPLIED
BY COORDINATE MATRIX $[20 \times 2]$ GIVES JACOBIAN MATRIX $[J]$ FOR $\zeta = -1$

$t1 = 1+\eta$
$t2 = -\zeta+\eta-1$
$t4 = 1-\zeta$
$t5 = t4*t1$
$t7 = \zeta+\eta-1$
$t9 = 1+\zeta$
$t10 = t9*t1$
$t12 = 1-\eta$
$t13 = -\zeta-\eta-1$
$t15 = t4*t12$
$t17 = \zeta-\eta-1$
$t19 = t9*t12$
$t21 = \eta**2$
$t36 = \zeta**2$
MATRIX [JJ7] [2 x 20] WHICH WHEN MULTIPLIED BY COORDINATE

MATRIX [20 x 2] GIVES JACOBIAN MATRIX [J] FOR $\zeta = -1$

\[
\begin{bmatrix}
0 & 0 & -11/4 & 11/4 & 0 & 0 & -112/4 & 119/4 & 0 & 0 & -x^11 & 0 & 0 & -x^12 & 0 & 0 & -1.E0/2.E0+121/2 & 1.E0/2.E0+121/2 \\
0 & 0 & 14/4 & 15/4 & 0 & 0 & -4/4 & 113/4 & 119/4 & 0 & 0 & 1.E0/2.E0+136/2 & 1.E0/2.E0+136/2 & -eta^14 & -eta^19
\end{bmatrix}
\]
APPENDIX D

VARIATION OF MATERIAL PROPERTIES OF THE TURBINE BLADE WITH TEMPERATURE

<table>
<thead>
<tr>
<th>TEMP (°C)</th>
<th>E (GPa)</th>
<th>σ_y (MPa)</th>
<th>α (μm/m°k)</th>
<th>c (J/kg)</th>
<th>K (W/m°k)</th>
</tr>
</thead>
<tbody>
<tr>
<td>21</td>
<td>220</td>
<td>840</td>
<td>11.9</td>
<td>400</td>
<td>12.7</td>
</tr>
<tr>
<td>93</td>
<td>215</td>
<td>842</td>
<td>12.07</td>
<td>400</td>
<td>13.0</td>
</tr>
<tr>
<td>205</td>
<td>215</td>
<td>844</td>
<td>12.07</td>
<td>395</td>
<td>13.5</td>
</tr>
<tr>
<td>315</td>
<td>195</td>
<td>846.3</td>
<td>12.4</td>
<td>420</td>
<td>13.8</td>
</tr>
<tr>
<td>425</td>
<td>190</td>
<td>848.4</td>
<td>12.8</td>
<td>440</td>
<td>15.1</td>
</tr>
<tr>
<td>540</td>
<td>185</td>
<td>850</td>
<td>13.1</td>
<td>420</td>
<td>15.2</td>
</tr>
<tr>
<td>650</td>
<td>175</td>
<td>855</td>
<td>13.5</td>
<td>460</td>
<td>17.3</td>
</tr>
<tr>
<td>760</td>
<td>170</td>
<td>840</td>
<td>14.0</td>
<td>480</td>
<td>14.0</td>
</tr>
<tr>
<td>870</td>
<td>160</td>
<td>760</td>
<td>14.8</td>
<td>500</td>
<td>21.6</td>
</tr>
<tr>
<td>980</td>
<td>145</td>
<td>470</td>
<td>15.8</td>
<td>525</td>
<td>24.9</td>
</tr>
</tbody>
</table>
APPENDIX E

PHYSICAL INTERPRETATION OF STRESS STIFFNESS MATRIX \([K_s]\):

If we consider a bar as shown in Fig (a) and assume that it can deform axially but is infinitely stiff in bending, so that it is straight in any displaced configuration, then the axial strain for small displacements is given by

\[ \varepsilon_x = \varepsilon_u + \varepsilon_v \]  

(E1)

where

\[ \varepsilon_u = \frac{u_2 - u_1}{L} \]  

(E2)

and

\[ \varepsilon_v = \frac{1}{2} \left( \frac{v_2 - v_1}{L} \right)^2 \]  

(E3)

Strain \( \varepsilon_v = \frac{\Delta L}{L} \)  

(E4)

where \( \Delta L \) is the lengthening associated with rotation of the bar through a small angle \( \theta \) without any motion in x-direction and is given by

\[ \Delta L = \frac{L}{\cos \theta} - L = L(\sec \theta - 1) = L \frac{\theta^2}{2} = L \frac{1}{2} \left( \frac{v_2 - v_1}{L} \right)^2 \]  

(E5)

246
Therefore strain energy in the bar is given by:

\[ U = A E L \left( \frac{\varepsilon_u^2}{2} + \varepsilon_u \varepsilon_v + \frac{\varepsilon_v^2}{2} \right) = \frac{A E L}{2} (\varepsilon_u^2 + \varepsilon_v^2) + P L \varepsilon_v \]  \hspace{1cm} (E6)

where \( A E \varepsilon_u \) is the axial force \( F \), positive in tension and

\[ \varepsilon_u = \frac{1}{L} \begin{bmatrix} -1 & 1 \end{bmatrix} \begin{bmatrix} u_1 \\ u_2 \end{bmatrix} \]  \hspace{1cm} (E7)

and

\[ \varepsilon_v = \frac{1}{2L^2} \begin{bmatrix} v_1 \\ v_2 \end{bmatrix}^T \begin{bmatrix} -1 & 1 \\ -1 & 1 \end{bmatrix} \begin{bmatrix} v_1 \\ v_2 \end{bmatrix} \]  \hspace{1cm} (E8)

The terms \( \varepsilon_u^2 \) and \( \varepsilon_v \) are quadratic in nature in nodal d.o.f. but \( \varepsilon_v^2 \) is quartic, so is negligible as compared to \( \varepsilon_u^2 \). Thus with \( \{d\} = \{u_1, v_1, u_2, v_2\} \),

\[ U = \frac{1}{2} \{d\}^T \begin{bmatrix} \frac{A E}{L} & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & \frac{F}{L} & 0 \\ 0 & 0 & 0 & 0 \end{bmatrix} \begin{bmatrix} 1 & 0 & -1 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 1 & 0 \\ 0 & 0 & 0 & -1 \end{bmatrix} \{d\} \]  \hspace{1cm} (E9)

It is clear from the equation above that the first 4 x 4 matrix with coefficient \( A E/L \) is the conventional stiffness matrix \([K]\) and the second 4 x 4 matrix with the coefficient \( F/L \) (in which \( F \) is the axial force) and non-linear strain \( \varepsilon_v \) is the stress stiffness matrix \([K_{st}]\).

The element stress stiffness matrix \([K_{st}]\) is a symmetric matrix and \([G]\) must be arranged to yield displacement derivatives in the same order as shown below:
\{\delta\} = [G] \{d\} \tag{E10}

where

\{\delta\} = \{u_x, u_y, u_z, v_x, v_y, v_z, w_x, w_y, w_z\} \tag{E11}

\{d\} is obtained by the relation:

\{F\} = [K^o] \{d\} \tag{E12}

where

[K^o] is the stress stiffness of the matrix

\{F\} is the force vector for the system obtained after assembling \{f\}

\{f\} is the force vector for an element given by \(\int_v N^T f' \, dv\)

f' is the centrifugal force acting on the element

The matrix [K^o] is given as:

\[
[K^o] = \int_v [G]^T \begin{bmatrix} s & 0 & 0 \\ 0 & s & 0 \\ 0 & 0 & s \end{bmatrix} [G] \, dv \tag{E13}
\]
Fig. (a)  TRANSVERSE FORCE ON AN AXIALLY LOADED TRUSS MEMBER  
(Warlikoo, 1989)